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# The Role of Fretting Fatigue on Aircraft Rivet Hole Cracking

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Final Report

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## EXECUTIVE SUMMARY

This is the final report on the FAA Grant No. 93-G-068 program conducted by the Quality and Integrity Design Engineering Center (QIDEC), Department of Mechanical Engineering, University of Utah. The program is entitled "The Role of Fretting Corrosion and Fretting Fatigue on Aircraft Rivet Hole Cracking."

A lap joint panel removed from an aircraft used in service was investigated for evidence of fretting induced cracking in and adjacent to the rivet holes. Cracks were found in all of the rivet holes that were inspected. Generally, they had nucleated in regions where there was evidence of fretting. This is significant because it indicates a potential for multiple-site damage occurring more rapidly than might be anticipated from a fatigue analysis or testing that did not consider fretting. In addition, fretting also could produce cracks at holes not viewed as "fatigue critical."

A sensitivity study was conducted to determine the effects of fretting on the fatigue lives of 2024-T3 clad sheet aluminum alloy riveted joint specimens prepared with either FV or CE rivets using either C-squeeze riveting or a rivet gun with bucking bar riveting procedures. It was found that fretting damage led to crack nucleation in all failed specimens. It also was concluded that with respect to fretting fatigue lives, based only on the results of this research program, the better overall rivet is the 7050 FV rivet. In spite of the previous conclusion, seven of eight specimens riveted by FV/C-squeeze procedures had rivet heads crack. This was concluded to have caused a reduction in specimen lives.

A method was developed to predict coefficient of friction characteristics within a fretted contact during the nucleation of a crack. This required development of a system capable of determining the coefficient of friction at controllable slip amplitudes as small as 80 micrometers and a verification test system. The method received limited verification and would be worth pursuing in further research because the results of such research might provide meaningful insights concerning the mechanisms of failure in riveted joints and other connections which involve fretting fatigue.

A finite element method was used to calculate the state of stress at CE rivet locations where fretting-nucleated cracks were observed during the sensitivity study portion of this grant program. This model predicted that after joint loading and unloading a residual compressive stress remains in the plate with the countersink, near the location where the body of the rivet makes contact with the plate. Then when the far field stress is reapplied the material near where the body of the rivet contacts the plate attains a tensile stress of less than 25,000 psi while higher tensile stresses occur near the junction of the body and countersink portions of the rivet.

The primary conclusion of this grant program is that fretting in riveted joints is a potentially major cause of crack nucleation in aircraft skin structure. It is a hazard at each rivet. This leads to the conclusion that design of riveted joints which ignores or inadequately considers fretting fatigue may result in multiple-site damage, which is recognized by the aviation community as a significant, yet unresolved, issue.

## 1 INTRODUCTION.

This report is submitted in accordance with the requirements of Department of Transportation, Federal Aviation Administration Regulation 9550.7, Research Grants Program. It covers the two year grant no. 93-G-068 program entitled "The Role of Fretting Corrosion and Fretting Fatigue on Aircraft Rivet Hole Cracking." The program was conducted by the Quality and Integrity Design Engineering Center (QIDEC), Department of Mechanical Engineering, University of Utah.

The American Society for Testing and Materials defines fretting as "a wear phenomenon occurring between two surfaces having oscillatory relative motion of small amplitude" and fretting corrosion as "a form of fretting wear in which corrosion plays a significant role" [1]. Fretting and fretting corrosion can be present in any area of an aircraft structure (e.g., engines, aircraft primary and secondary structure, and landing gear components) in which small amplitude cyclic slip between adjacent contacting materials is possible. When at least one of the fretted components also experiences fatigue loading, the process is called fretting fatigue. The effects are synergistic with the component life possibly being reduced by an order of magnitude or more as a result of fretting fatigue when compared to fatigue without fretting. The main effect of the fretting is an accelerated nucleation of cracks which then may propagate due primarily to the fatigue loading [2]. Rivets and mechanically fastened joints in general are particularly susceptible to fretting fatigue which can result in multiple-site damage because the damage can occur at many rivet holes. The damage can link up in aircraft structures and if not detected during inspection could create catastrophic results. Thus, it is imperative to understand the role of fretting fatigue in producing multiple-site damage in riveted aircraft joints.

The work performed under this grant was organized into four subprograms as explained in the procedures section which follows. The work was interrelated, with the performance of each subprogram enhanced by the results and experience obtained during the performance of the others.

## 2 PROCEDURES.

### 2.1 INSPECTION OF A LAP JOINT PANEL FROM A SERVICE AIRCRAFT.

A lap joint panel from Saudi Air, Boeing 707-320C, serial number 19810 (39,834 hours and 26,017 flight cycles) was investigated for evidence of fretting induced cracking in and adjacent to the rivet holes. A picture of this panel, which was provided by Boeing Defense and Space Group, is shown in figure 1. Of interest is the lap joint region that runs horizontally across the panel, slightly above the center of the picture.

The panel was visually inspected and inspected with an optical microscope. Selected riveted joints then were sectioned in the vertical direction in figure 1 and the rivets removed. The rivet holes were inspected visually and with an optical microscope. Then the rivet holes and regions adjacent to the holes were inspected in a scanning electron microscope (SEM). To aid in crack detection, tensile loading was used to expand the rivet holes. Procedures were similar to those used by Piascik et al. [3]. Following these inspections, those rivet hole sections which had cracks of interest, based on subjective evaluation, were fractured so that the crack surfaces could be inspected to determine the crack nucleation sites.

Figure 2 shows the rivet pattern and numbering scheme assigned to the rivets in the panel. The critical row of rivets is that containing rivets 1, 4, ..., 46, 49. Rivet 46 was randomly selected from this row and then its hole and those of proximate rivets 43, 47, 48, and 49 were inspected in a pilot investigation. Based on the results of this investigation, it was decided to inspect all rivet holes in the critical row and five additional randomly selected rivet holes in the other two rows. The rivet holes that were inspected are darkened in figure 2.

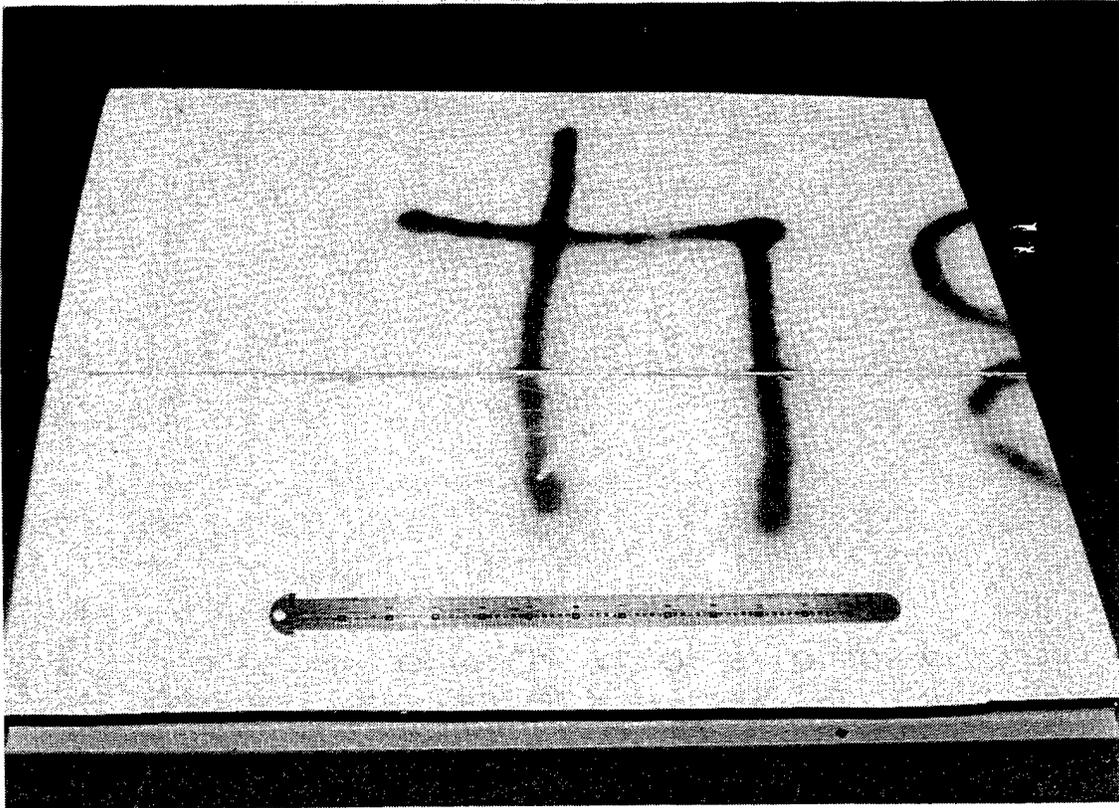


Figure 1. Picture of the lap joint panel that was investigated.

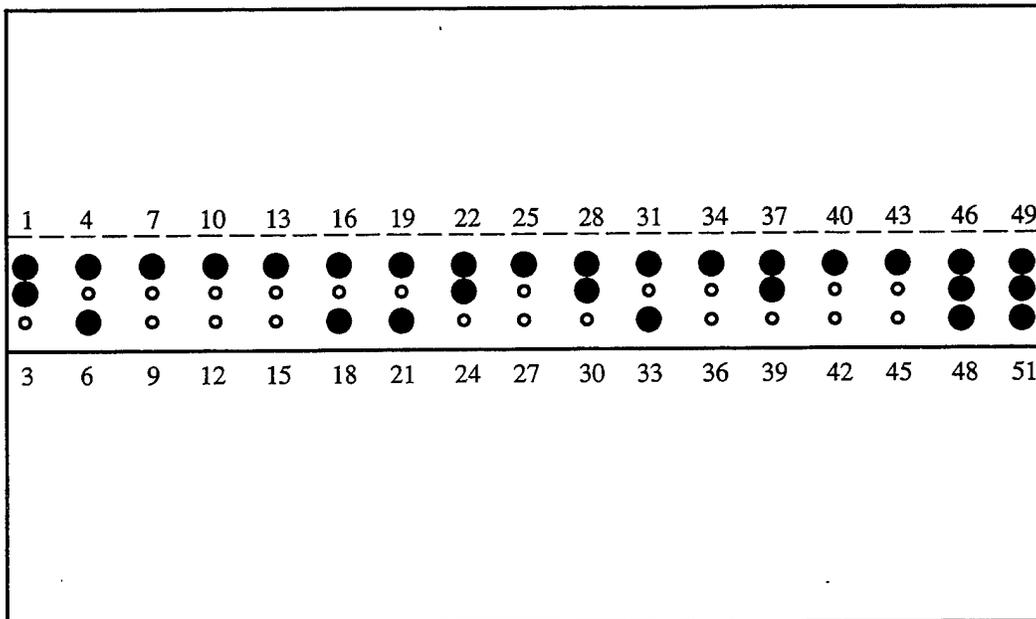


Figure 2. Rivet pattern and numbering scheme assigned to the rivets in the panel. The rivet holes that were inspected are darkened.

## 2.2 SENSITIVITY STUDY.

A sensitivity study was conducted to determine the effects of fretting on the fatigue lives of riveted joint specimens made from 2024-T3 clad sheet aluminum prepared with either 7050 aluminum alloy FV rivets or 2017 aluminum alloy CE rivets. Figure 3 depicts profile views of an FV and a CE rivet. Specimens were riveted using either C-squeeze riveting or a rivet gun with bucking bar.

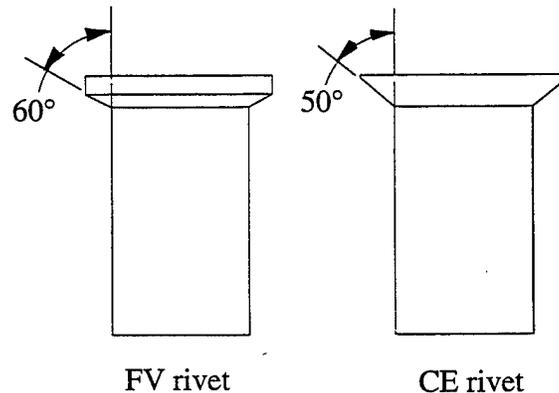


Figure 3. Profile views of an FV and a CE rivet.

Initially, eighteen specimens were prepared, tested, and examined in accordance with the plan in appendix A. The 0.063 gage 2024-T3 clad sheet aluminum alloy coupons for the specimens were inventoried, sized, allodined, and primed. Then, all specimens were fabricated in accordance with the drawings in appendix B. The eighteen specimens were fabricated with riveting procedures as follow:

- four C-squeeze joints with 7050 FV rivets,
- four C-squeeze joints with 2017 CE rivets,
- five rivet gun with bucking bar joints with 7050 FV rivets, and
- five rivet gun with bucking bar joints with 2017 CE rivets.

Testing was conducted using a single MTS servo hydraulic load system with a 10 Hz sine wave driving signal, a maximum nominal tensile stress of 17 ksi, and a stress ratio  $R = 0.1$ . The drawings in appendix C are for the system that was used to counter the tendency of the riveted joints to bend in the riveted area when under out-of-plane tensile loading. Figure 4 shows a specimen attached in the MTS load frame with the antibending system installed.

Analysis of the test results indicated that the lives of CE riveted specimens were shorter than expected. Investigation showed that the rivet hole countersink depths were too great for these specimens and this was hypothesized to be the cause of the decreased lives. Consequently, twelve additional specimens were prepared, tested, and examined using similar procedures as before, in accordance with the plan in appendix D. The twelve specimens were fabricated from 0.063 gage 2024-T3 clad sheet aluminum alloy with riveting procedures as follow:

- four C-squeeze joints with 7050 FV rivets,
- four C-squeeze joints with 2017 CE rivets, and
- four rivet gun with bucking bar joints with 2017 CE rivets.

The CE riveted specimens were included for the data they generated. The FV riveted specimens were included to allow correlation with the initial testing program. To the extent possible, all

conditions for the second test were the same as the conditions for the initial testing program except for the depth of the CE rivet hole countersinks.

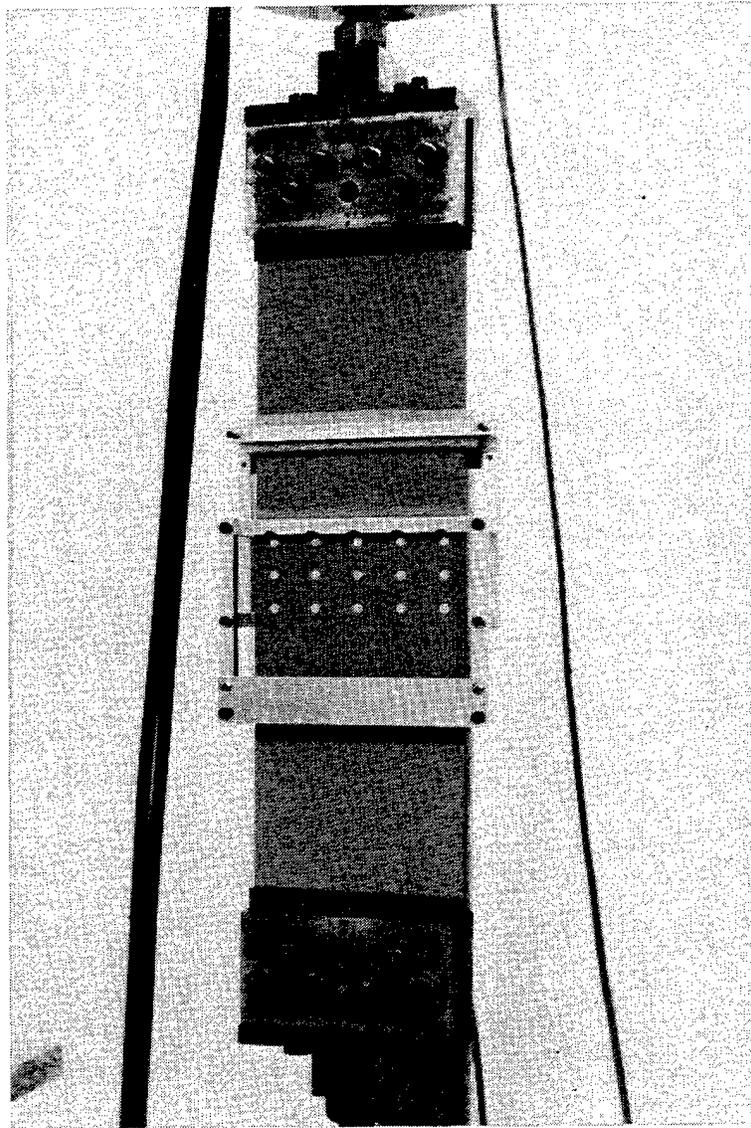


Figure 4. Specimen attached to the MTS load frame with the antibending system installed.

All actions for the sensitivity study portion of this program were performed in such a manner as to ensure the statistical integrity of the program. Full use was made of random number generation, blocking, and other statistical sampling techniques in deciding the riveting procedures to use and the orders in which specimens were to be machined, riveted, or tested. The operator, equipment, and procedures were the same for all specimen preparation and testing operations.

During data reduction, statistical comparisons of the fatigue lives of specimens within a sample\* and between samples were made by standard cumulative normal distribution function hypothesis testing methods [4]. In these procedures, a hypothesis is stated concerning whether the values to be compared are from the same population. The level of confidence desired in the results of the analysis is also assumed. A standard table is entered with a "Z" value that is a measure of the number of standard deviations between the values to be compared. This provides a measure of the confidence that the values are not from the same population which is compared to the desired level of confidence. Alternately, given a desired confidence level, the corresponding "Z" value can be determined. Then the hypothesis is accepted or rejected based on the confidence or "Z" value determined. Acceptance of the hypothesis means that you can be confident (at the level of confidence chosen) that the values are not from the same population. This provides statistical justification for considering one population superior to the other based on the conditions of the experimental program and the confidence level chosen. For example, for this program the sample with the "statistically" longer mean life is considered to be from a population with better fretting fatigue life characteristics than the sample with the shorter mean life. Rejection of the hypothesis means that there is no statistical basis for considering one population superior to the other based on the conditions of the experimental program and the confidence level chosen.

For this program 99 percent confidence ("Z" value of 2.326) was used for comparison in all cases except one where 95 percent confidence ("Z" value of 1.645) was used. In comparing specimens within a sample, the specimen being considered was assumed not to be from the same population. Therefore, the difference between the sample mean (without the value of the specimen being considered) and the value for that specimen was determined. This difference was divided by the sample standard deviation (computed without the value of that specimen) to determine the "Z" value. For comparisons between samples, the two samples were assumed to be from the same population. A combined standard deviation (standard error of the difference) for the two samples was computed by the equation:

$$SD = [\sigma_1 / (n_1)^{1/2}] + [\sigma_2 / (n_2)^{1/2}]$$

where: SD is the standard error of the difference  
 $\sigma_1$  is the standard deviation for the first sample  
 $n_1$  is the number of values in the first sample  
 $\sigma_2$  is the standard deviation for the second sample  
 $n_2$  is the number of values in the second sample

Then the difference in the sample means was divided by this "SD" value to determine the "Z" value which was compared to the "Z" value of 2.326 or 1.645.

Following testing, the fracture surfaces of the failed specimens were investigated in order to gain insights into the factors that resulted in failure, especially the influence of fretting on the fatigue process.

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\* Except as justified in the discussion, a sample is considered to be all specimens from the same test program produced with the same type rivets and riveting procedures.

### 2.3 PREDICTION OF COEFFICIENT OF FRICTION DURING FRETTING.

Traditionally for many materials the metal dry friction between contacting surfaces has been characterized by a ratio of the friction force which impedes relative motion between the surfaces to the normal force which holds the surfaces together. There are two such ratios: the coefficient of static friction ( $\mu_s$ ) as motion is impending and the lower-value coefficient of kinetic friction ( $\mu_k$ ) while there is relative motion or slip between the surfaces. These values are global in that they and slip amplitude are considered constant within the contact region. Also, they are not considered to be functions of wear (cycles), contact area, geometry, and surface roughness.

In fretting fatigue portions of the fretted interface may experience relative slip while other portions may stick (not slip) and the boundary of the stick-slip region experiences high-stress concentrations that can result in crack nucleation. Additionally, the friction characteristics within a fretted interface change with cycling due to wear. Therefore, to better understand the fretting fatigue process it is necessary to understand friction at the local level and how it changes with cycling. In this program where a finite element model was used local is defined as the element size of 0.00048 in<sup>2</sup>.

A method was developed to predict coefficient of friction characteristics within a fretted contact during the nucleation of a crack. The method consists of an iterative procedure that compares computed local frictional values from a three-dimensional (3D) linear ADINA® finite element model with predicted frictional values and updates the predictions as necessary until agreement is reached at each model node. Starting with the first cycle, the method steps forward in cycles at a rate that ensures convergence until the required number of cycles has been considered. A flow chart of the protocol that ensures convergence is shown in figure 5.

The method uses coefficient of friction data determined using the system shown mounted in an MTS load frame in figure 6. A drawing of this system is shown in figure 7, and a drawing of the contact area is shown in figure 8. This system is capable of determining coefficient of friction values at slip amplitudes as low as 80 microinches (2 micrometers) with uncertainty as shown in table 1.

Table 1. Total uncertainty in the measurement of coefficient of friction data (worst on worst error analysis)

Variable	Sources of uncertainty	Total uncertainty
Ratio of tangential to normal forces	Thermal drift of tangential load sensor	±7.9 % maximum
	Signal noise of tangential load sensor	
	Angle of load application (wire) relative to bridge	
	Thermal drift of displacement sensor	
	Signal noise of displacement sensor	
	Angle of sensor relative to bridge	
Relative displacement	Thermal drift of displacement sensor	±9.7 % maximum
	Signal noise of displacement sensor	
	Angle of sensor relative to bridge	

Eighteen tests were conducted to determine frictional values of 2024-T3 on 2024-T3 at slip amplitudes of 80, 440, or 800 microinches and average normal tractions of 2,000, 6,750, or 13,000 psi to develop the coefficient of friction data needed to support the method verification.

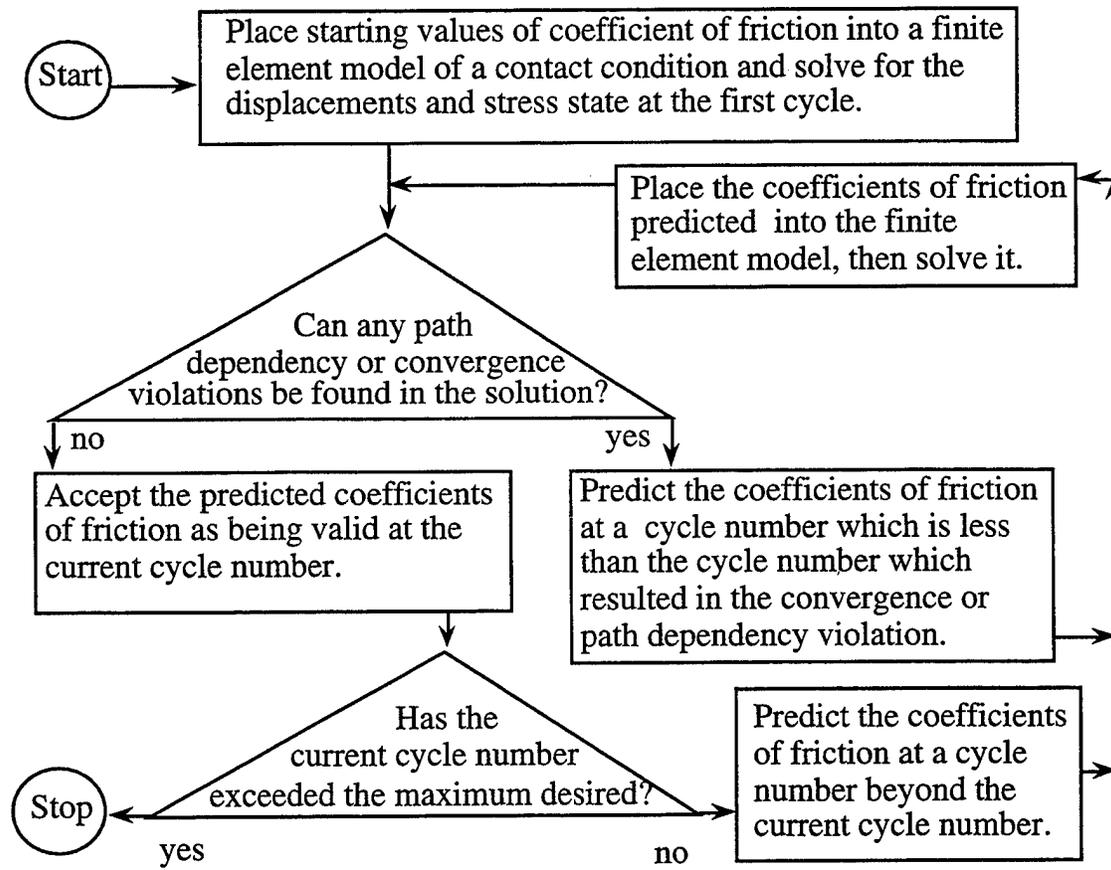


Figure 5. Flow chart of the method to predict how the coefficient of friction will change for a fretting contact.

A system was developed to provide verification of the coefficient of friction prediction method discussed above. The system is shown mounted in an MTS load frame in figure 9 and a drawing of the system is shown in figure 10. This system uses indirect verification by measuring strain values adjacent to the fretted contact which can be compared with strain values predicted by the coefficient of friction prediction method. The verification system uses a flat rectangular contact surface 1.22 in. along by 0.1 in. across the direction of fatigue loading.

The coefficient of friction prediction method is general purpose and can be used with an ADINA® model of whatever contact geometry is being considered. The verification system geometry was modeled during this program. Due to symmetry along and across the fatigue loading direction, one quarter of the 1.22 by 0.1 in. surface was modeled by the ADINA® program. This was modeled with a 64 element array (16 elements along by 4 elements across the direction of fatigue loading).

The systems mentioned above were developed as part of the FAA program. More detailed information concerning these systems is contained in reference [5] or can be obtained by contacting the authors.

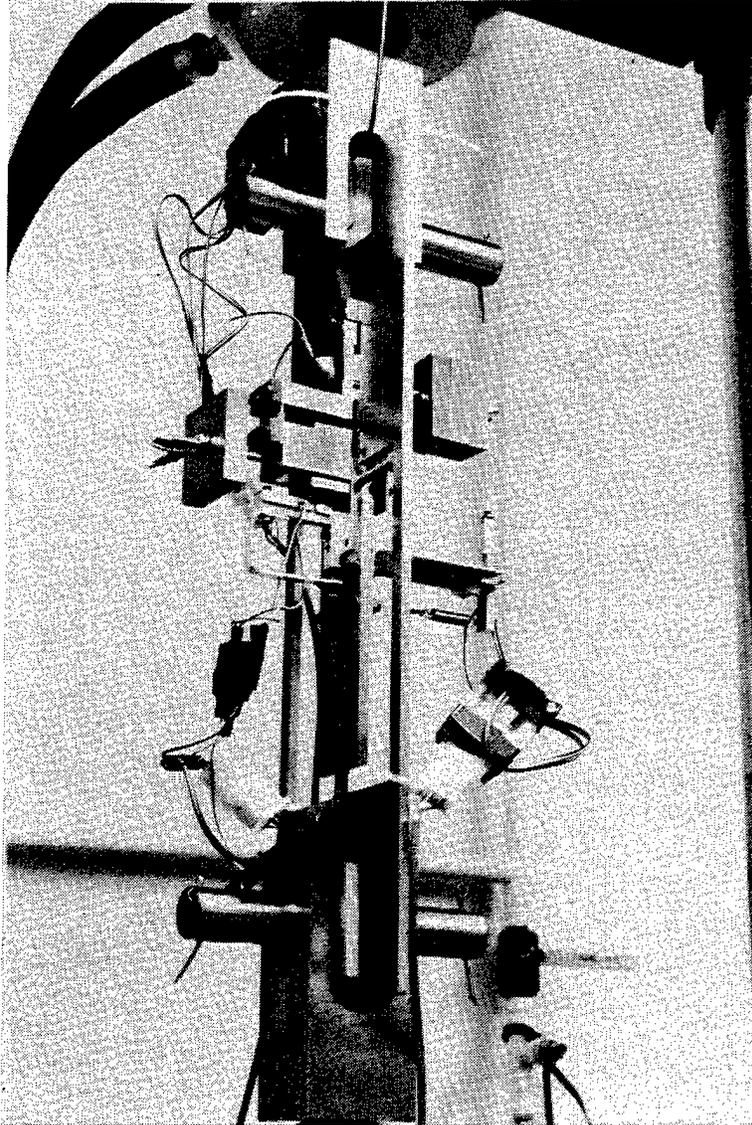


Figure 6. Close-up view of the fretting test apparatus.

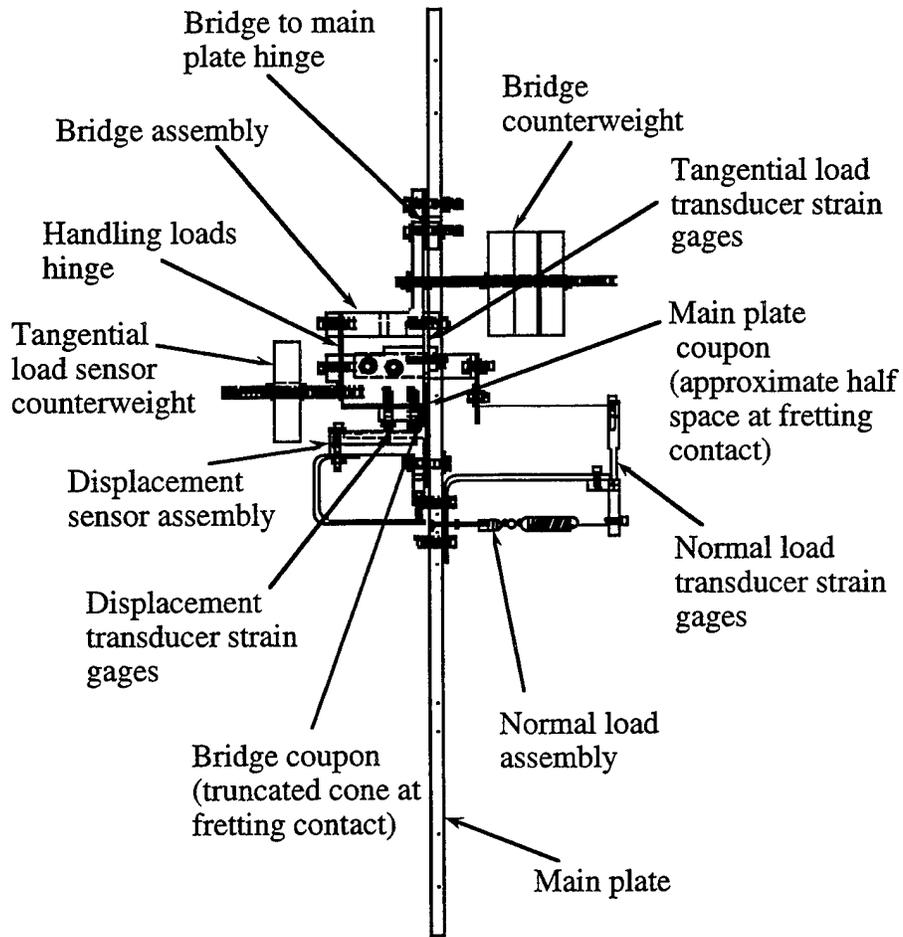


Figure 7. Major components of the coefficient of friction during fretting test apparatus.

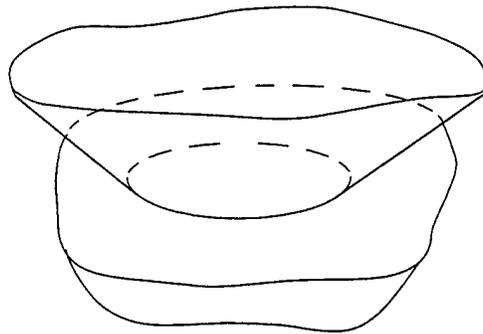


Figure 8. Truncated cone on half space fretting contact geometry.

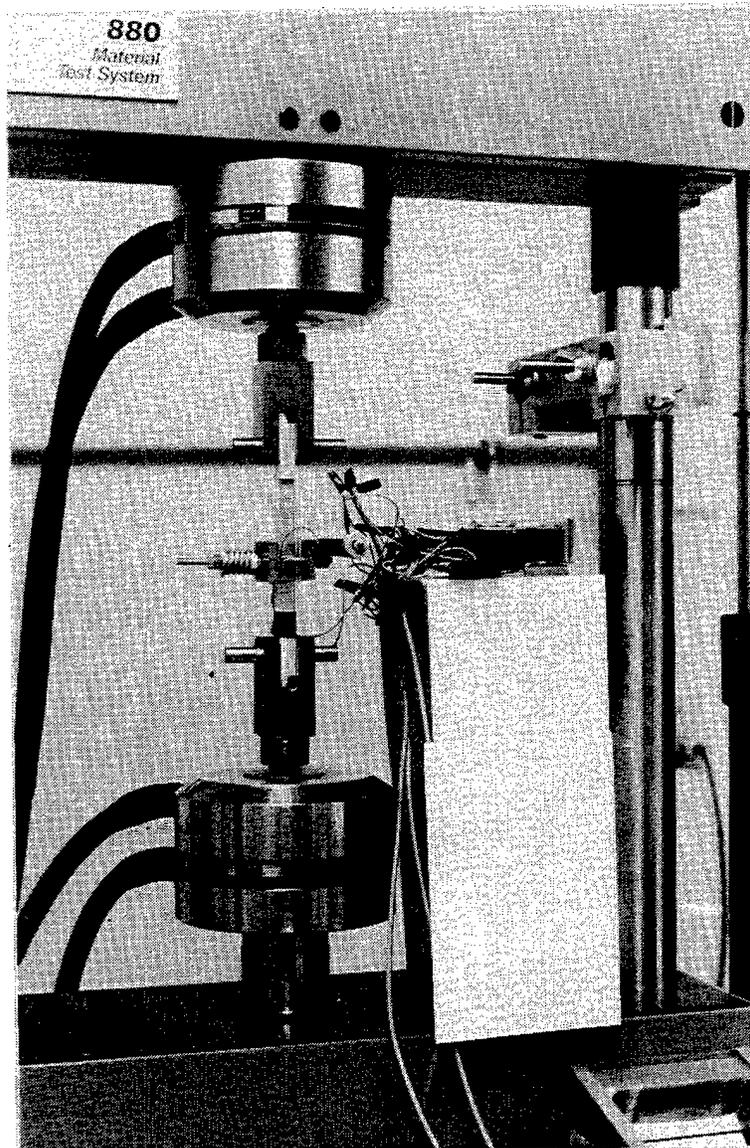


Figure 9. The verification test apparatus installed in a load frame. (The items to the right of the apparatus are bridge completion resistors, thermal compensation gages, and shunt calibration jumpers).

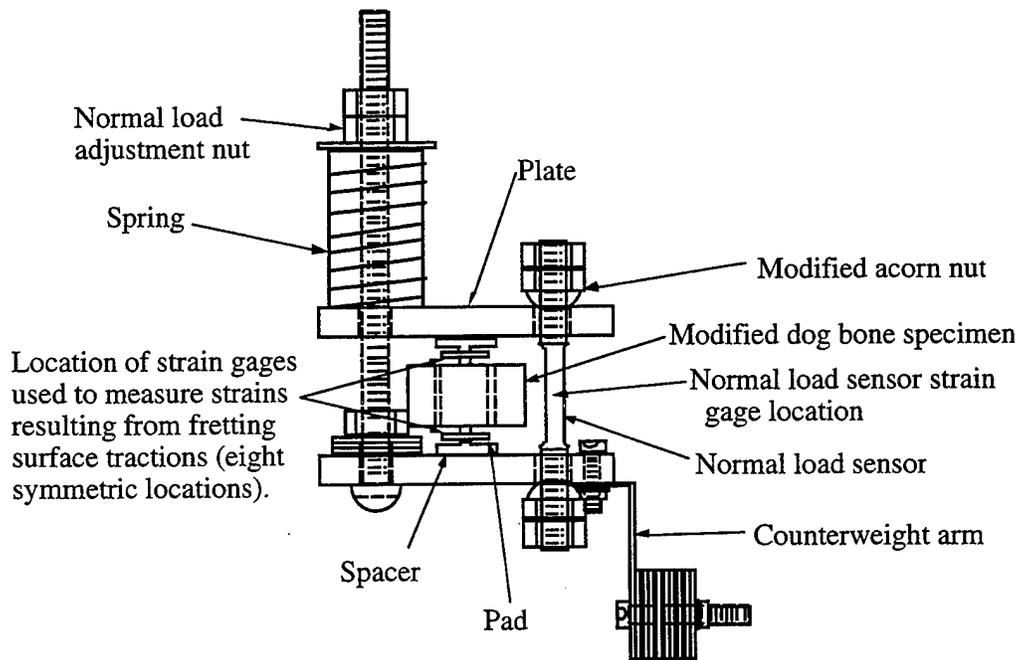


Figure 10. The verification test apparatus with the major components labeled.

#### 2.4 RIVETED JOINT COMPUTER SIMULATION.

A finite element model of a one rivet section of the CE joint tested during the sensitivity study portion of this grant was constructed. This was considered to be a valid baseline condition to use for applying the method for prediction of coefficient of friction during fretting (section 2.3) to a riveted joint. This model was developed and executed for a simulated 1 1/2 cycles as shown in figure 11.

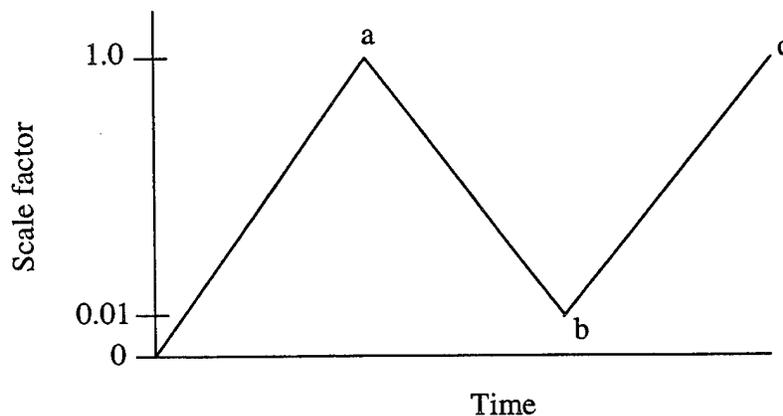


Figure 11. Cycles simulated during the analysis. The scale factor was based upon a far field maximum stress of 17,000 psi as used in the sensitivity study.

The model simulated one of the fasteners on the critical row as shown in figure 12. The boundary conditions of the model are shown in figure 13. The portion of far field stress which was considered to be transferred by the critical row of fasteners was based on other research in QIDEC [6]. The model included the influence of plasticity of the upper sheet, lower sheet, and rivet. Kinematic hardening was assumed. The interfacial sealant and the friction between the rivet and the sheets also were included in the model. The direction of the axis of the rivet was not constrained. The ADINA-IN<sup>®</sup> and ADINA-PLOT<sup>®</sup> codes used to obtain the plots found in this report will be made available upon request to the authors.

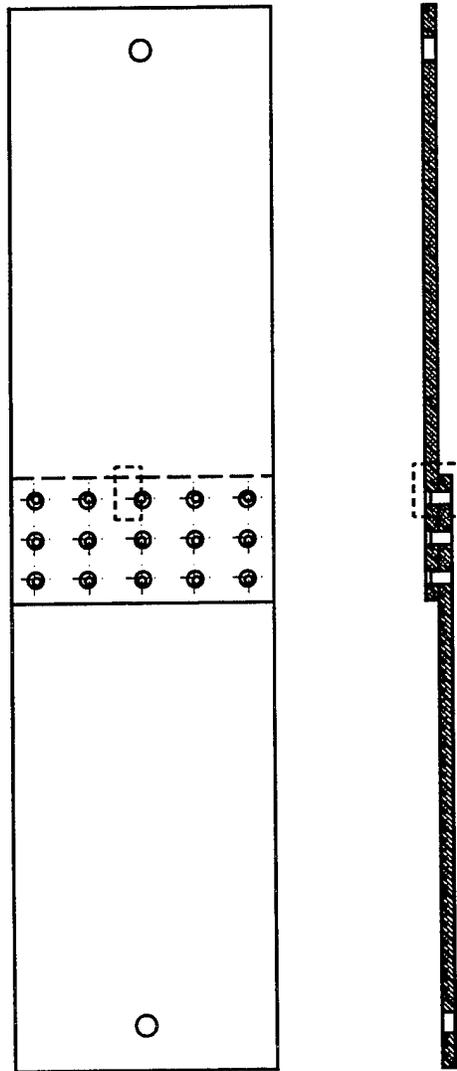


Figure 12. Diagram of a riveted joint used during the sensitivity study with the boundaries of the computer model shown in dashed lines.

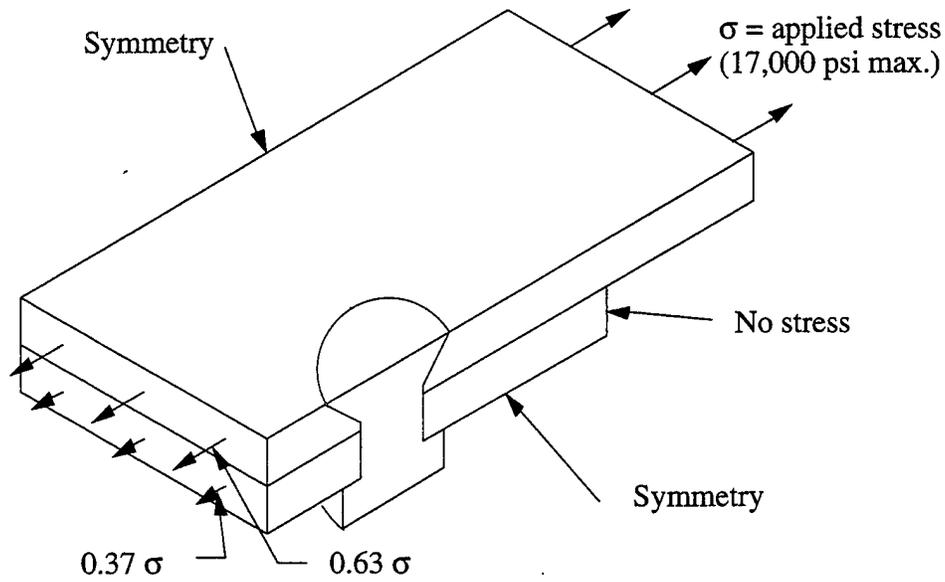


Figure 13. Boundary conditions of the model.

A plot of the model is shown in figure 14 and a plot of the riveted joint mesh is shown in figure 15. A plot of half of the upper plate mesh is shown in figure 16. This figure includes an arrow indicating the high stress region in which the stress plots described in section 3.4 are shown. Figure 17 shows a magnified view of this high stress region mesh.

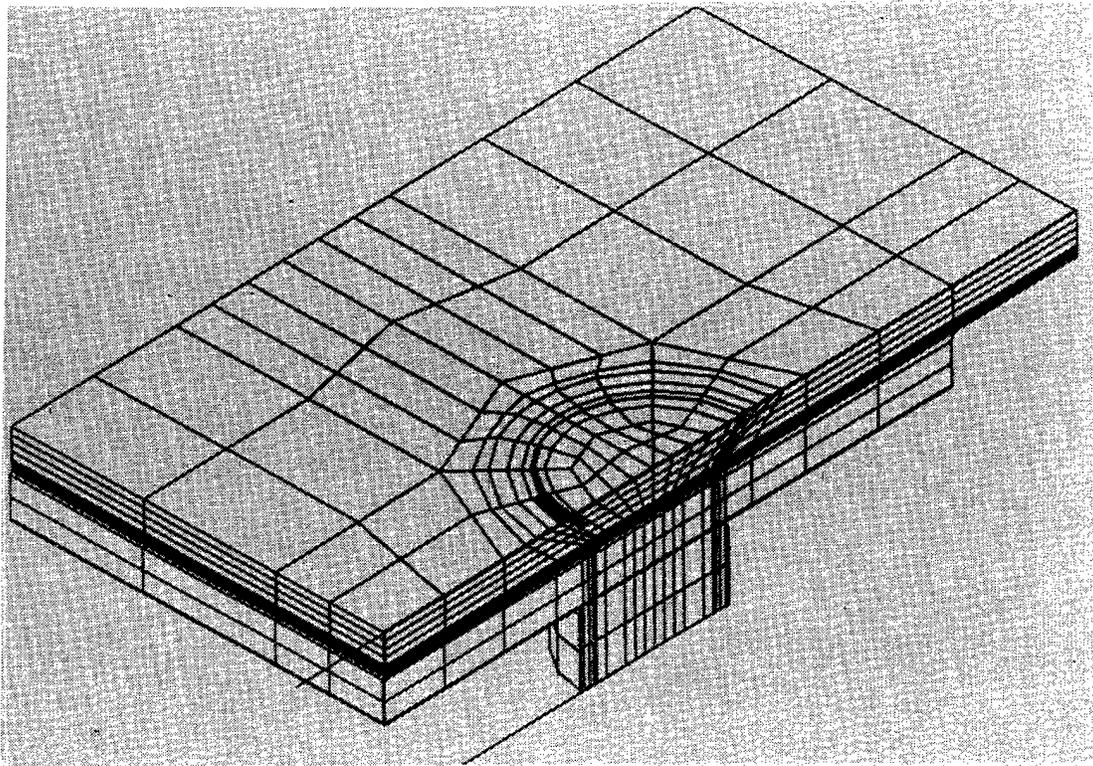


Figure 14. Mesh of the entire model. The dark areas are due to the resolution of the plotter and have no practical significance.

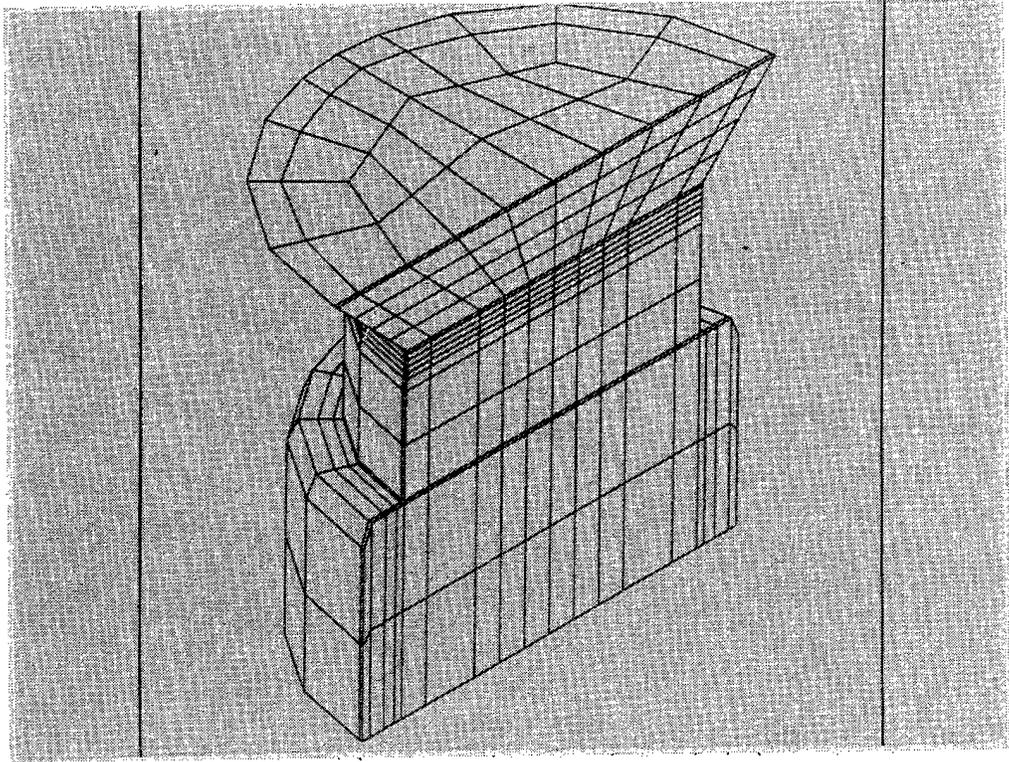


Figure 15. Mesh of the rivet.

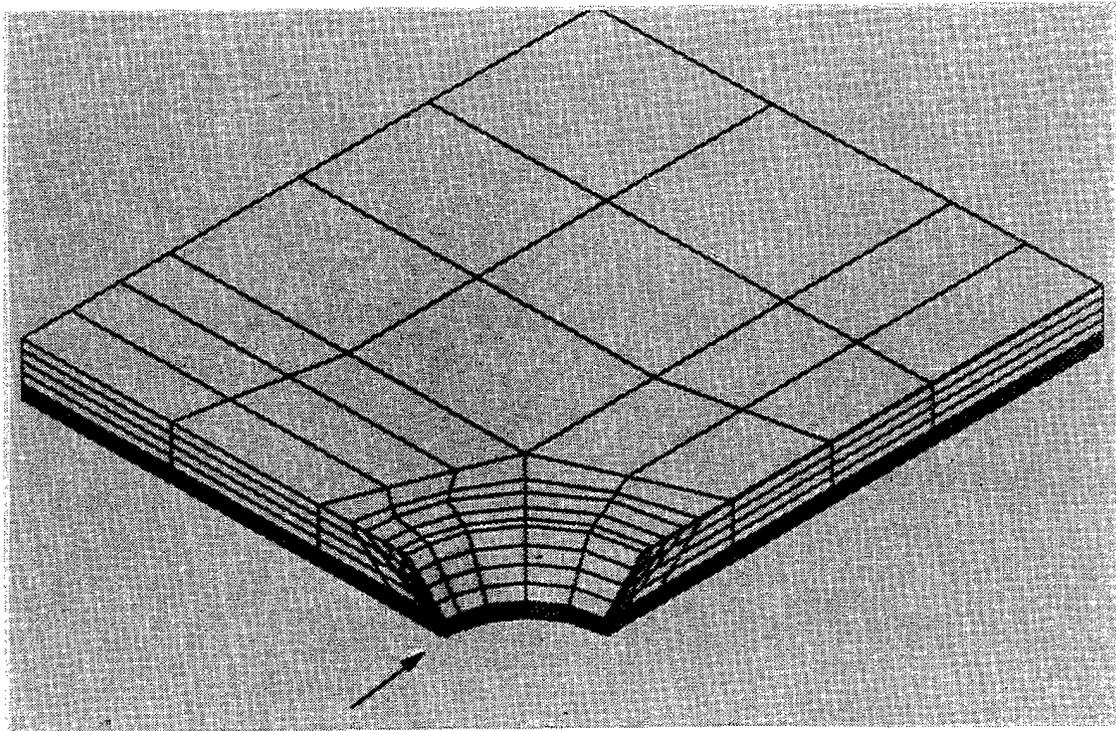


Figure 16. Mesh of one quarter of the upper plate. An arrow marks the high stress region for which the stress plots in section 3.4 are shown. The stress direction for the plots is the direction of fatigue loading which is from upper right to lower left in this figure.

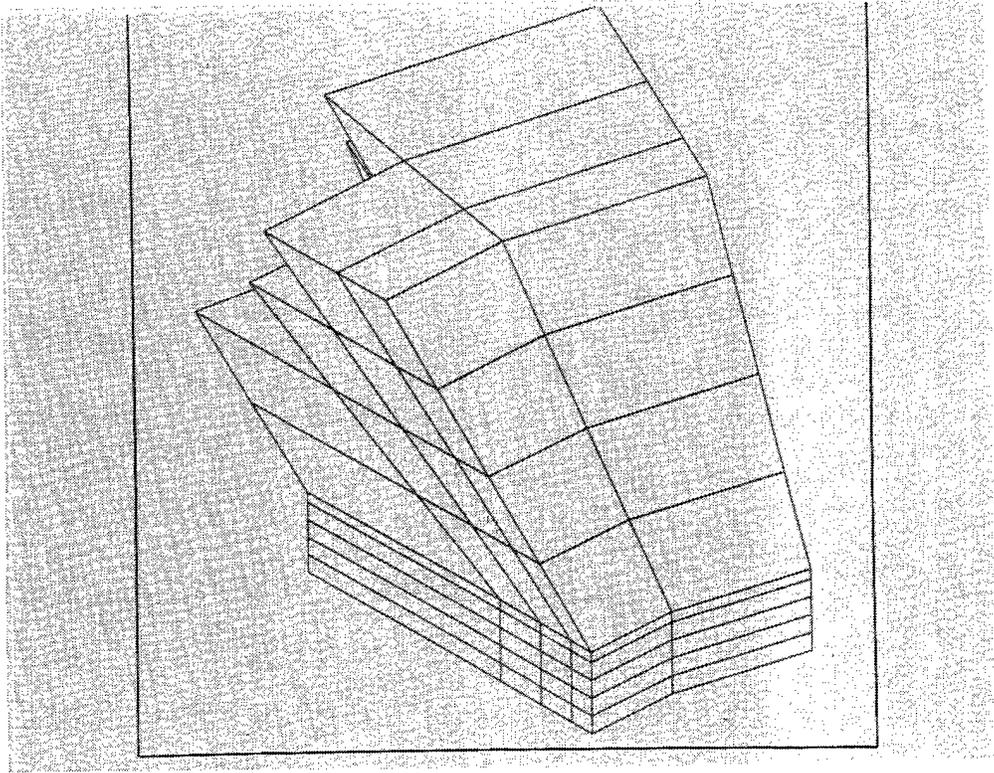


Figure 17. Mesh of the region just underneath the countersink in the upper plate in the region of highest stress concentration.

### 3 RESULTS.

#### 3.1 INSPECTION OF A LAP JOINT PANEL FROM A SERVICE AIRCRAFT.

Visually, without magnification, the panel appeared to be serviceable and corrosion free. There is no reason to believe that this panel would have been inspected further or replaced in a periodic inspection. However, close optical inspection showed the presence of a black substance indicative of fretting at all but two of the rivet holes in the panel.

By optical microscope and/or SEM inspection fretting debris was found in all of the rivet holes that were inspected. Additionally, cracks were found in the fretting debris in each hole. Figures 18 and 19 show examples of cracks in fretting debris in rivet holes as previously defined in figure 2.

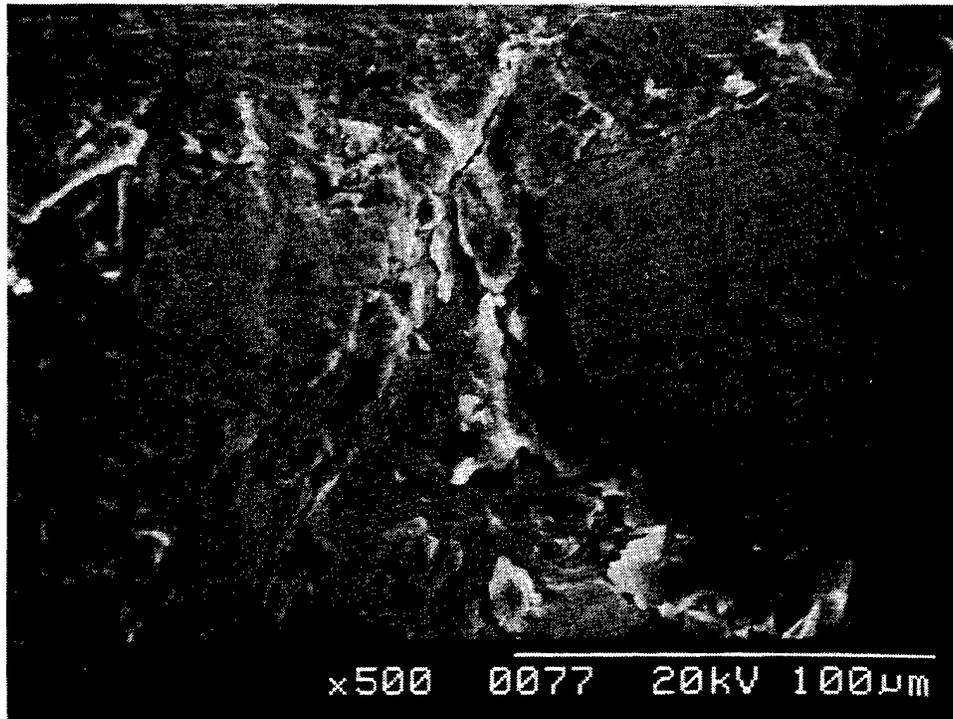


Figure 18. Example of cracks in fretting debris in rivet hole 1. The top of the picture is in the countersink region and the bottom is in the rivet body region.

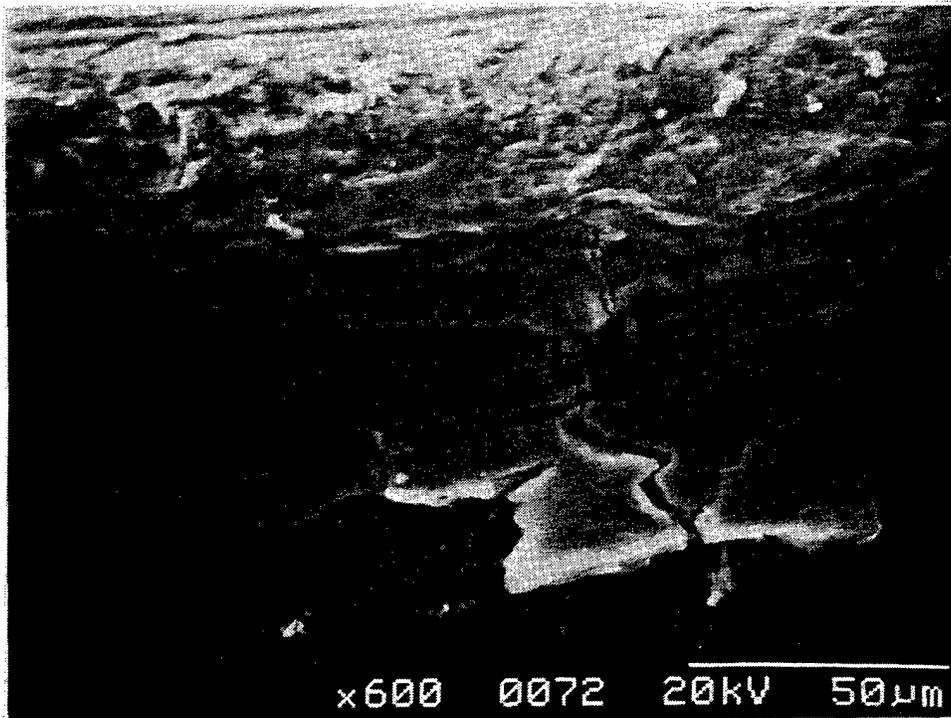


Figure 19. Example of cracks in fretting debris in rivet hole 46. The top of the picture is in the countersink region and the bottom is in the rivet body region.

Several cracks were found in regions that appeared to have had relatively little fretting damage. However, these cracks were generally attributable to other identifiable causes such as the crack emanating from a void or inclusion as shown in figure 20.

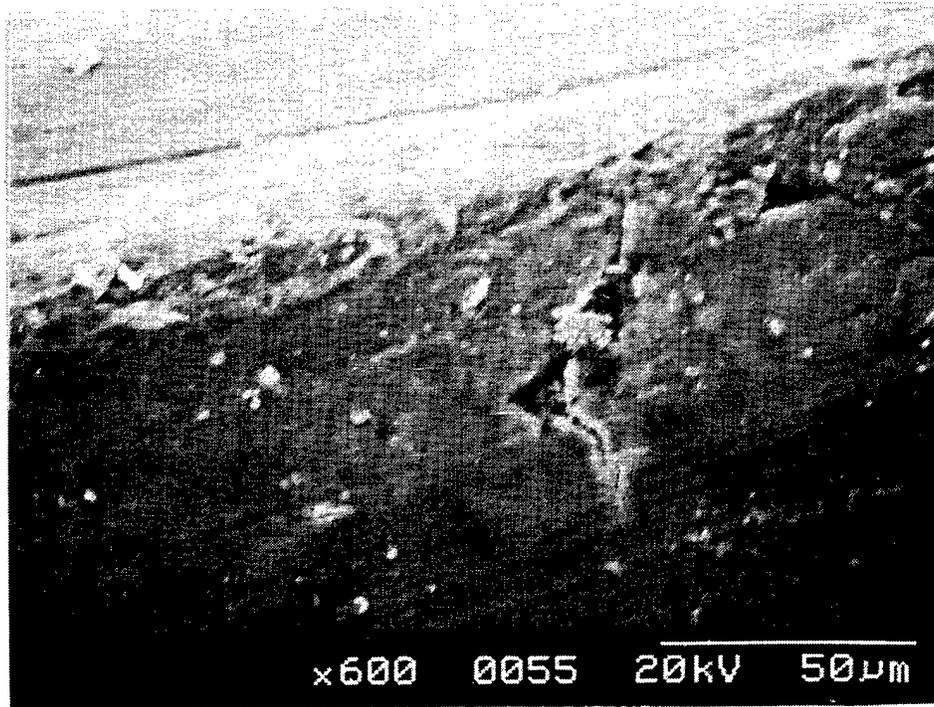


Figure 20. Crack emanating from a void or inclusion in a relatively unfretted region in rivet hole 49.

Based on examination of the surfaces of the rivet holes and examination of the crack surfaces for those cracks in fretting debris, it was determined that crack nucleation generally occurred in the rivet body region of the rivet holes near the countersink region of the holes. This can be seen in figures 18 and 19. Crack surface views of two other fatigue cracks within the rivet holes are shown in figures 21 and 22.

Observation of figures 21 and 22 can lead to a misconception that only single cracks occurred. This was not necessarily the case, but rather a function of how the region near the rivet hole was fractured for inspection. As can be seen in figure 23, numerous cracks were possible within the rivet holes.

Subjectively, it seemed that fretting damage was more severe in holes in the region near rivet hole 47. For this rivet, the axis of the machined countersink was not concentric to the axis of the hole as drilled. It is believed that this misalignment may be responsible for the increase in fretting damage in holes near rivet hole 47.

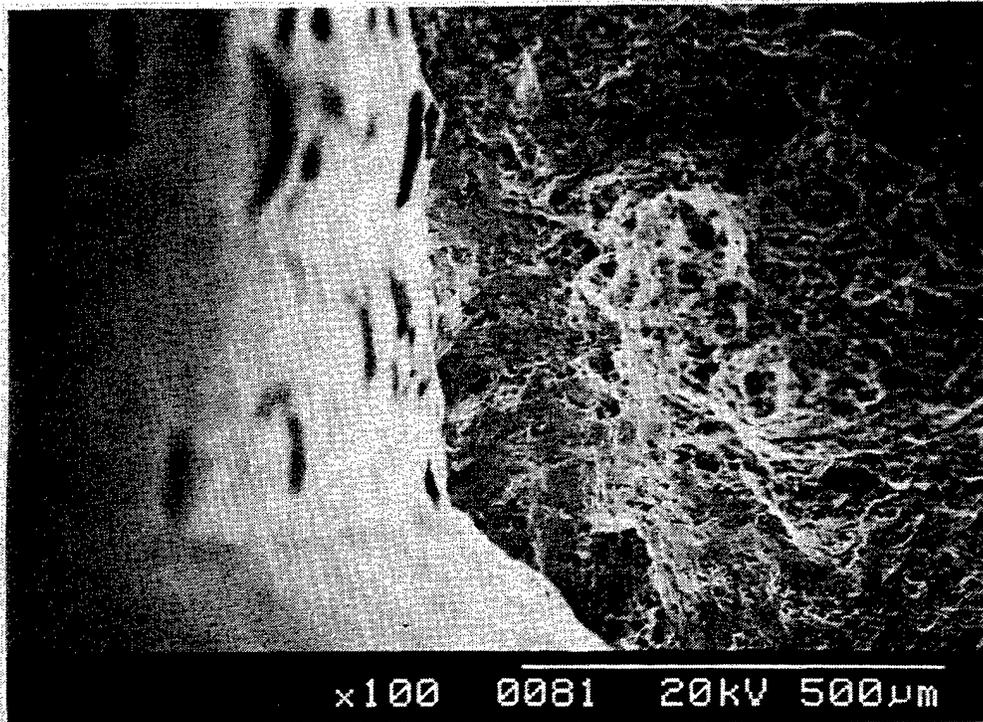


Figure 21. Crack surface at rivet hole 49. The top of the picture is in the rivet body region and the bottom is in the countersink region.

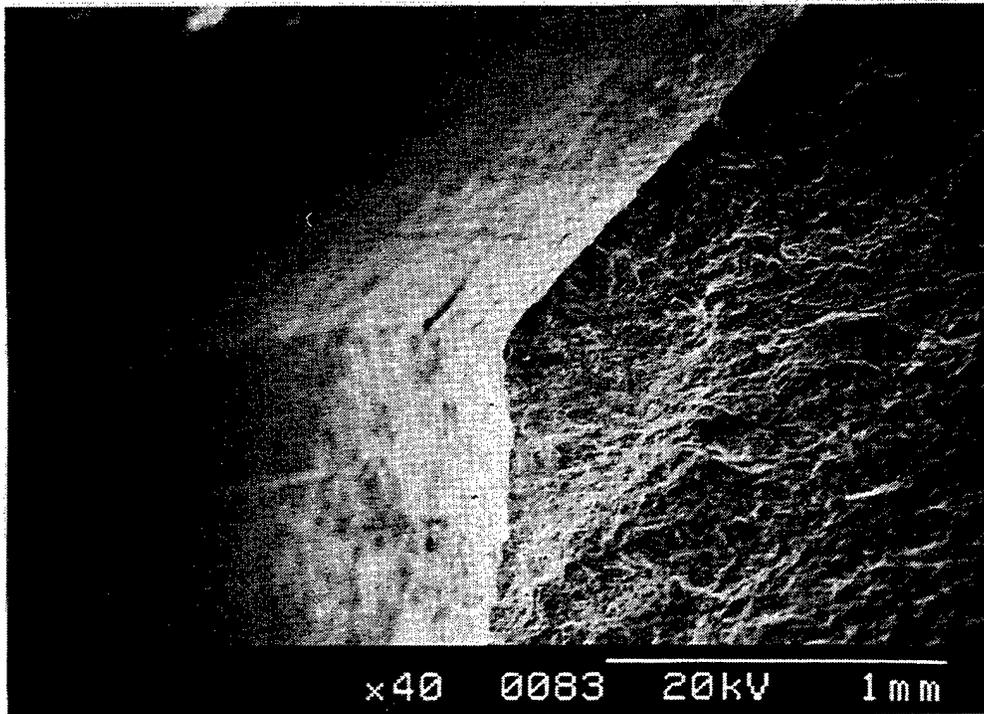


Figure 22. Crack surface at rivet hole 50. The top of the picture is in the countersink region and the bottom is in the rivet body region.

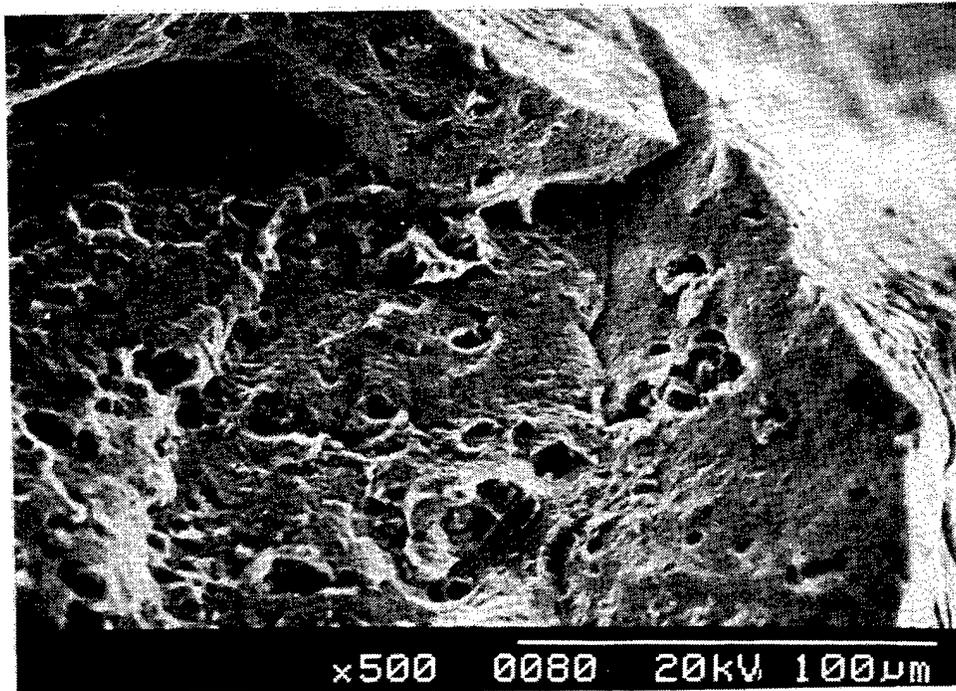


Figure 23. Crack surface at rivet hole 1. The fracture surface shows several cracks, and additional cracking can be seen on the rivet hole surface to the right of the figure.

### 3.2 SENSITIVITY STUDY.

Data pertaining to the fabrication, testing, and analysis of individual specimens are contained in appendix E. Specimens 1 through 18 are the initial test specimens. Specimens 1R through 12R are for the second test program.

Desired rivet head clearance above the joint surface was 0.003 in. (See drawings FAASPC94044 and FAASPC94045 in appendix B.). For the initial testing, the modal value of CE rivet head clearance was found to be 0.000 in. (flush) with 77 percent of the values from -0.001 to 0.001 in. For the follow-on testing, the modal value of CE rivet head clearance was 0.003 in. with 88 percent of the values from 0.002 to 0.004 in. For all testing, the modal value of FV rivet head clearance was 0.002 in with 81 percent of the values from 0.001 to 0.003 in.

For all specimens, failure resulted from instability due to fatigue crack propagation in the line of rivets 11 through 15 as shown in figure 24. Cycles-to-failure data are shown in table 2. With the exception of some relatively short cracks, all fatigue cracks that were identified had nucleated in fretted regions within or adjacent to the rivet holes. These locations of fretting were identified by observed fretting debris. Tables 3 and 4 contain a listing of the locations of fretting debris in which cracks nucleated for the CE and the FV rivets respectively. Figures 25 and 26 contain the keys for the location codes in tables 3 and 4 respectively. Figures 27 and 28 show typical cracks which nucleated in fretted regions. Figure 29 shows the fracture surface of a plate for which crack nucleation was in the region between the 2024 alloy sheets.

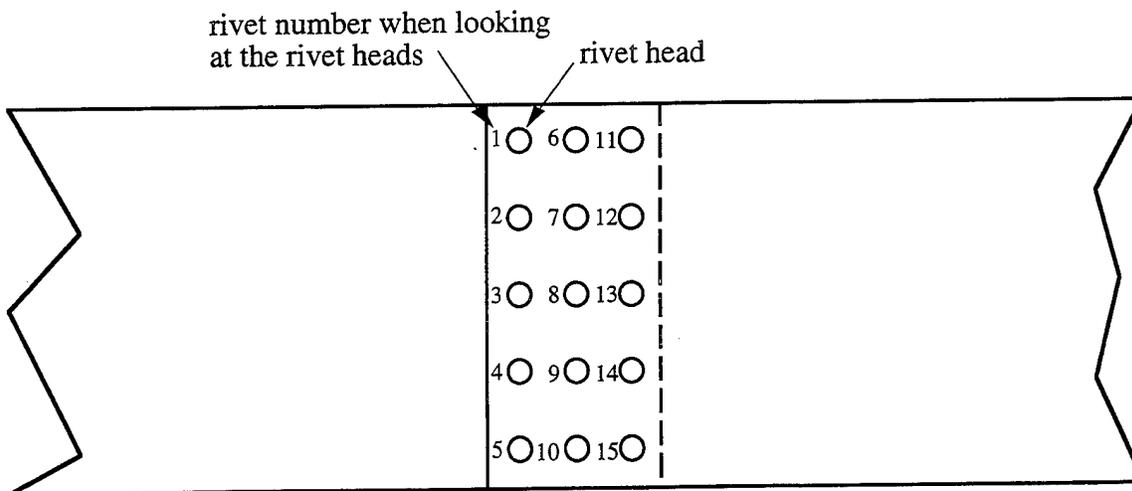


Figure 24. Riveted region in a sensitivity study specimen showing the rivet numbering scheme.

Specimens 3, 13, and 10R did not fracture (break into two pieces) because the error detectors sensed failure and activated the interlocks which stopped specimen loading before fracture could occur. Because of the extent of cracking in these specimens (for example see figure 30), it is felt that additional cycling would have been minimal if the test had not been stopped because the interlocks were activated. Therefore, the cycle count at the time of interlock for each specimen was used in the data analysis.

Seven of the eight specimens riveted with FV/C-Squeeze procedures were found through optical microscope inspection to have cracks along a chord of the circular top of the rivet head. Four of these rivet heads (one each in specimens 8, 11, 3R, and 10R) had fractured. Figure 31 shows fractured specimen 11 in which rivet 13 has fractured and a crack is observed in rivet 11. In all, 12 rivets were observed to be cracked or fractured. Of these, one was in rivet 11, three were in rivet 12, five were in rivet 13, and three were in rivet 14. No cracks were found in any CE or FV/rivet gun with bucking bar specimens. The fracture surface of rivet 13 specimen 11 is shown in figure 32. Ratchet marks near the arrow in this figure indicate that fatigue cracks nucleated at the countersink surface of the rivet. This is confirmed in figure 33 which shows fatigue striations located at the point of the arrow in figure 32.

Table 2. Sensitivity study cycles to failure

	2017 CE specimens		7050 FV specimens	
	<u>Specimen</u>	<u>Cycles</u>	<u>Specimen</u>	<u>Cycles</u>
<u>Initial Program</u>				
Rivets installed by rivet gun with bucking bar	9	32,200	5	178,020
	18	33,580	1	216,450
	7	36,880	17	248,090
	15	41,400	16	248,860
	4	<u>43,130</u>	12	<u>363,780</u>
Mean	37,438		251,040	
Standard Deviation	4,763		69,376	
Rivets installed by C-squeeze	14	37,380 <sup>1</sup>	8	142,660 <sup>2,3</sup>
	3	40,270	11	146,860 <sup>2,3</sup>
	10	54,080	13	167,720
	6	<u>58,030</u>	2	<u>193,520<sup>3</sup></u>
	Mean	47,440		162,690
Standard Deviation	10,146		23,292	
Mean without 37,380 value	50,793			
Standard Deviation without 37,380	9,325			
<u>Follow-on Program</u>				
Rivets installed by rivet gun with bucking bar	2R	87,680		
	4R	95,690		
	11R	101,890		
	8R	<u>107,710</u>		
Mean	98,243			
Standard Deviation	8,583			
Rivets installed by C-squeeze	12R	112,890	3R	120,050 <sup>2,4</sup>
	9R	127,980	10R	156,640 <sup>2,3</sup>
	5R	139,250	6R	157,340 <sup>3</sup>
	1R	<u>168,980</u>	7R	<u>174,460<sup>3</sup></u>
	Mean	137,275		152,123
Standard Deviation	23,736		22,915	
Mean without 120,050 value			162,813	
Standard Deviation without 120,050			10,092	
<u>Consolidated FV/C-squeeze statistics</u> (without 120,050 value)				
Mean			162,743	
Standard Deviation			17,471	

- 1 For unknown reasons, seating of the rivets in this specimen was found to be incomplete on inspection the day after the riveting. Twelve rivets were removed and the specimen riveted.
- 2 Specimen had fractured rivet heads.
- 3 Cracks were observed in rivet heads that had not fractured as a result of testing.
- 4 Failure of the grips (not the specimen) occurred during this test. The grips were fixed and the test continued.

Table 3. Locations of fretting debris in which cracks nucleated and frequency of nucleation within the debris for CE riveted joints. See figure 25 for the meaning of the location codes.

Specimen	CE1	CE2	CE3	CE4	CE5	CE6
3						
4			6		1	
6	1?		3	1	1	
7		2	2	2		
9		1	5	4		
10		2	4	1		
14		2	6		1	
15		2	6	1		
18			8	1		
1R		1	6	2	2	
2R			3	1		
4R			7	1	1	
5R	1?	2	7	1	1	
8R	1, 1?	1	5	3	3	
9R		1	5	3	2	
11R			7	2	1	
12R		2	5	1	2	

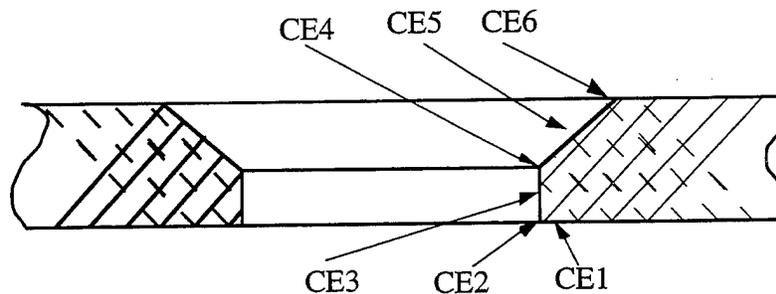


Figure 25. Drawing of a hole in a plate for a CE rivet showing the coded locations of cracks which nucleated from fretted regions. The codes are used in table 3 above and in the text.

Table 4. Locations of fretting debris in which cracks nucleated and frequency of nucleation within the debris for FV riveted joints. See figure 26 for the meaning of the location codes.

Specimen	FV1	FV2	FV3	FV4	FV5	FV6	FV7
1			3	1		2	
2	3?		1	2, 1?			1
5			2	6			3
8	5?	1?		2			
11	4?						
12			7	1			
13							
16			4	1			
17		1	4	5		4	
3R	1?			3		3	2
6R		1		2		1	3
7R			1	1	1		1
10R							

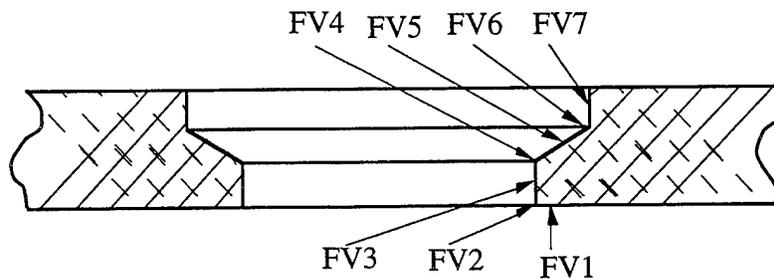


Figure 26. Drawing of a hole in a plate for an FV rivet showing the coded locations of cracks which nucleated from fretted regions. The codes are used in table 4 above and in the text.

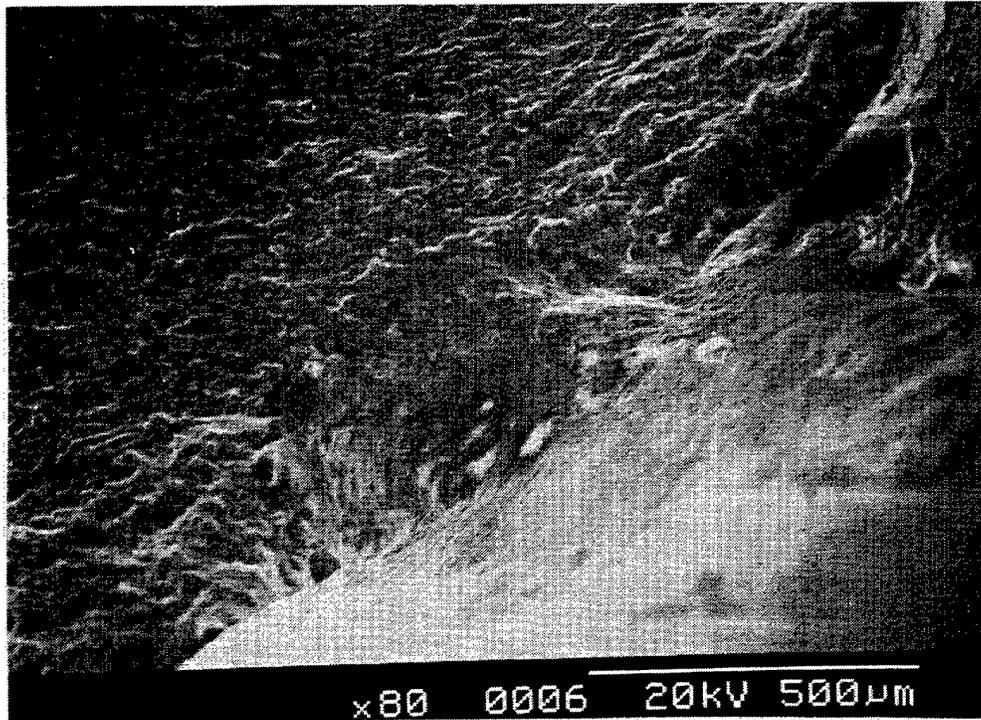


Figure 27. Fracture surface of a crack which nucleated within a region subject to fretting. The fretting is at location CE5 of figure 25.

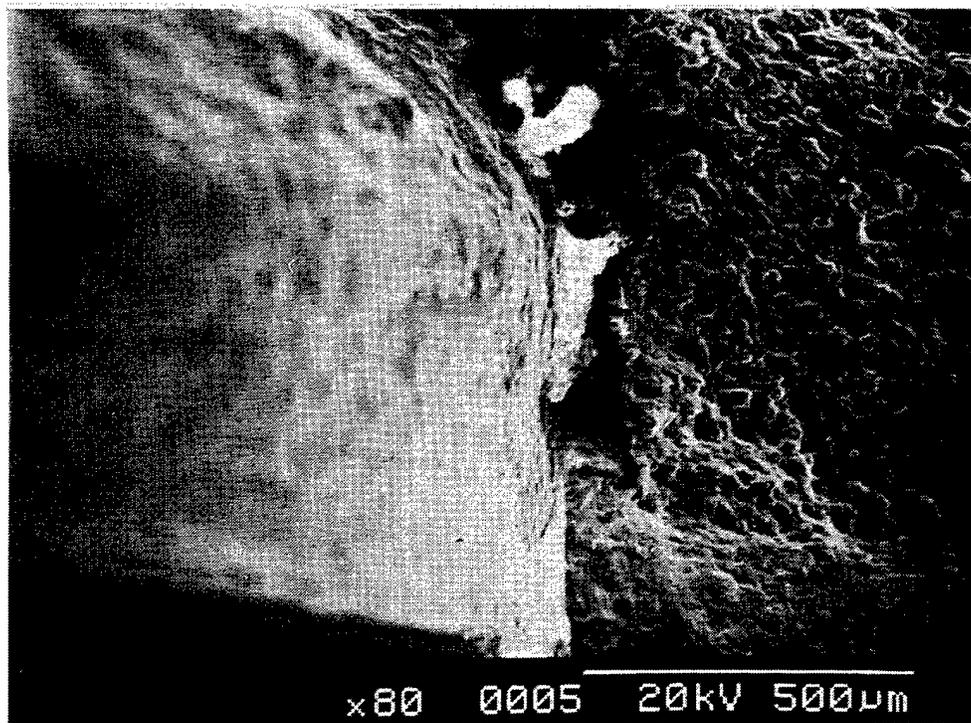


Figure 28. Fracture surface of a crack which nucleated within a region subject to fretting. The fretting is at location FV6 in figure 26.



Figure 29. Fracture surface (bottom of picture) for crack from a fretted area (see arrow) between the plates. The substance at the top of the picture is interfacial sealant.

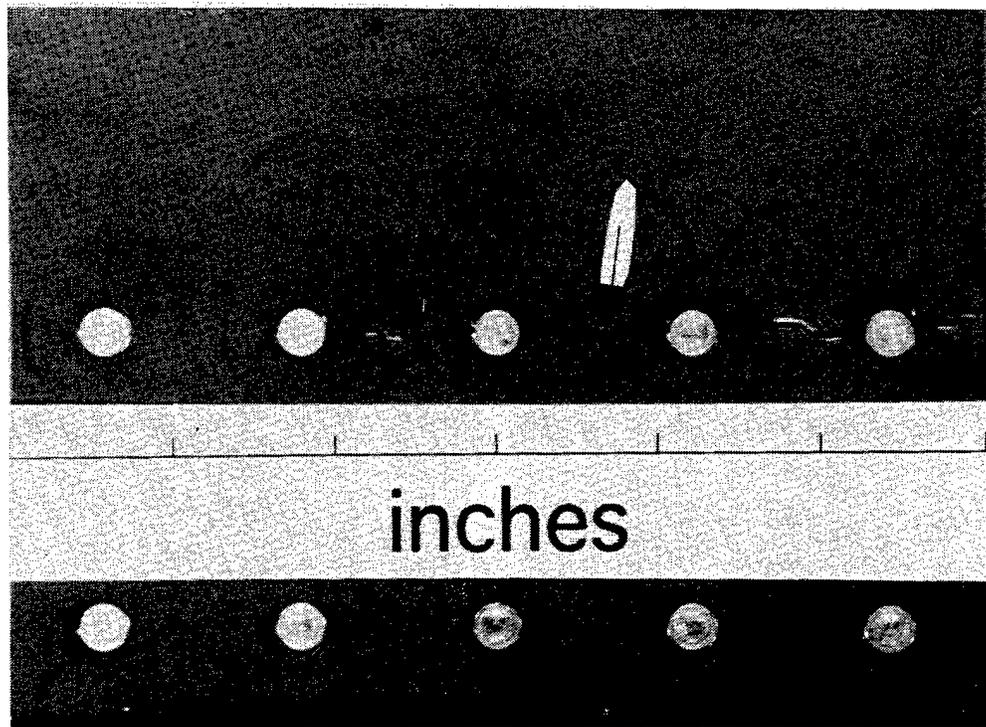


Figure 30. View of test specimen 3 showing extensive cracking but not fracture.

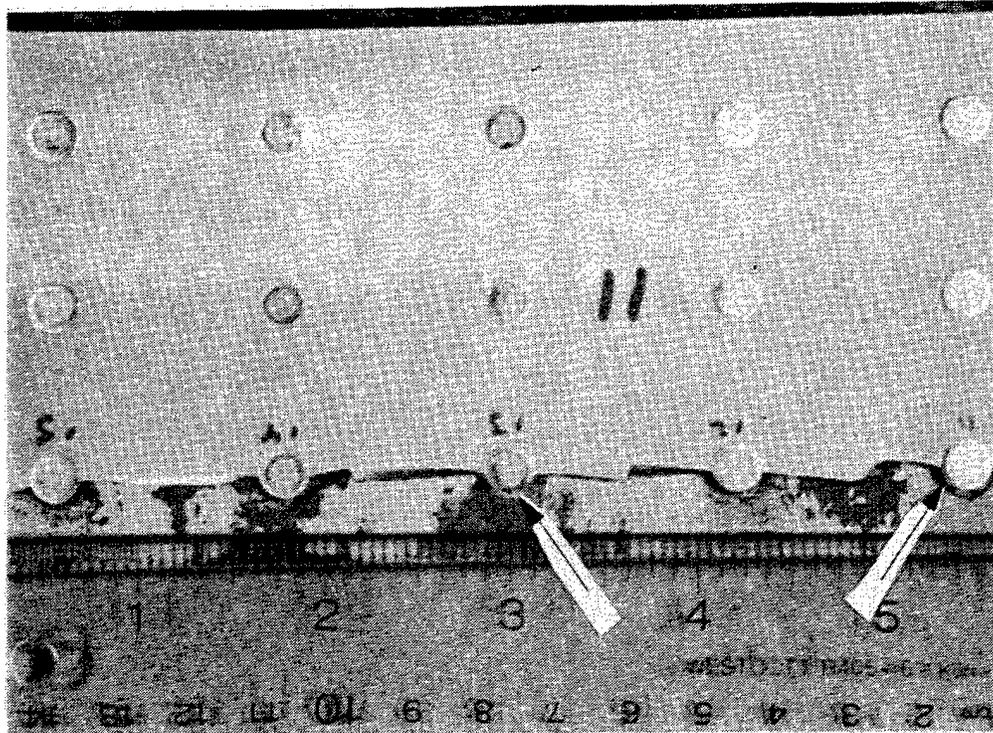


Figure 31. Fractured specimen 11 in which rivet 13 has fractured and a crack is observed in rivet 11.

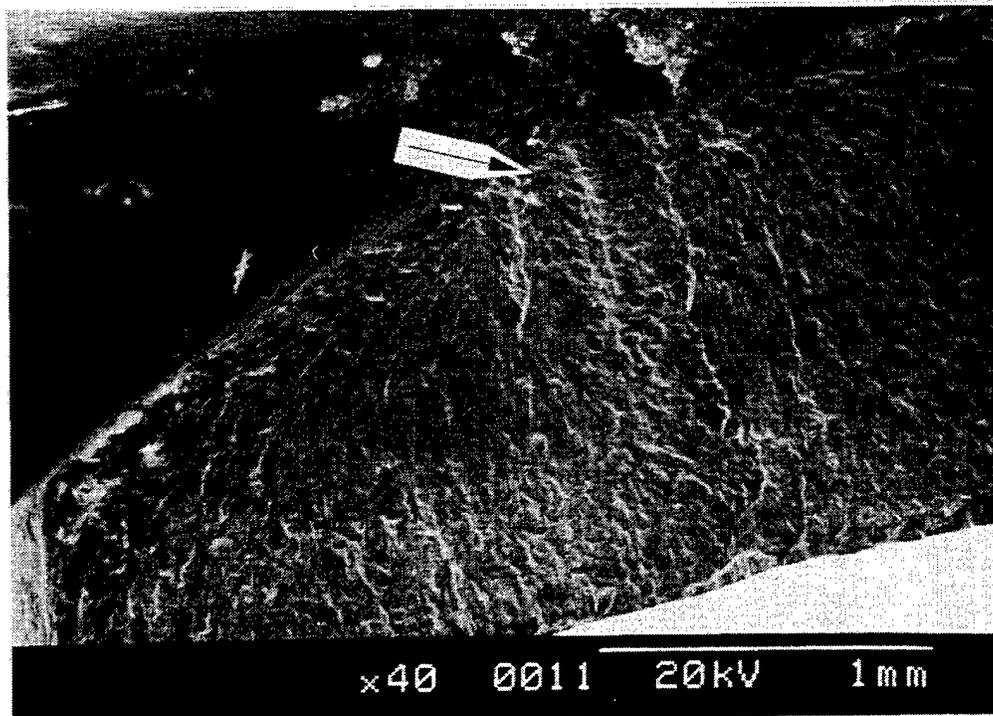


Figure 32. The fracture surface of rivet 13 specimen 11. Ratchet marks near the arrow indicate that fatigue cracks nucleated at the countersink surface. The striations in figure 33 are at the point of the arrow in this figure.

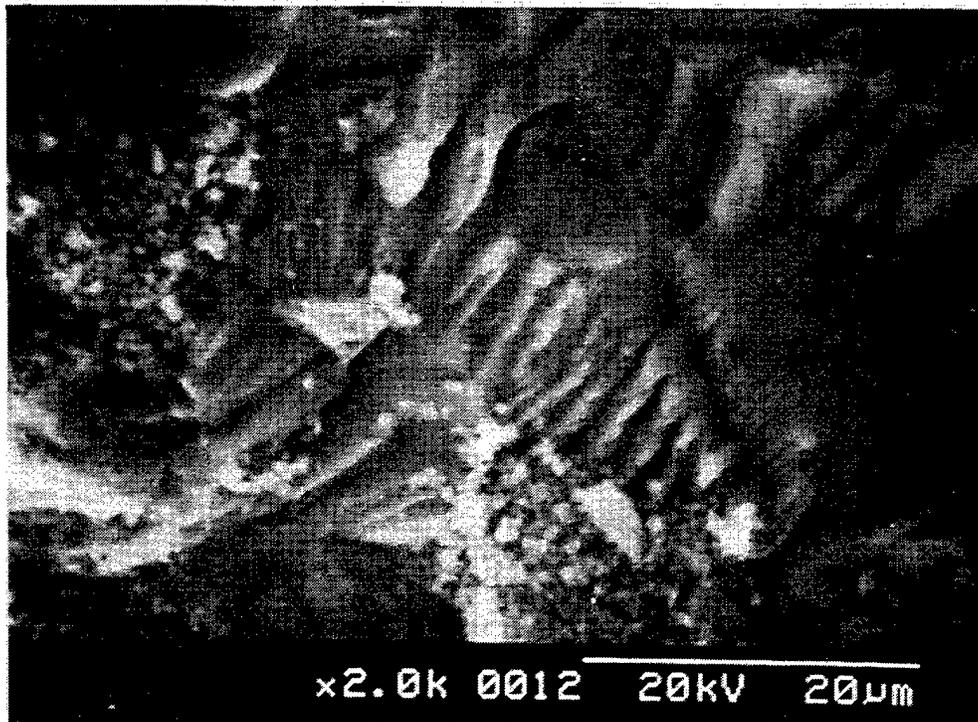


Figure 33. Striations indicative of fatigue crack propagation in rivet 13 specimen 11. This picture is at the point of the arrow in figure 32 and is oriented in the same direction as that figure.

### 3.3 PREDICTION OF COEFFICIENT OF FRICTION DURING FRETTING.

Based on the work of Vincent, et al. [7] results of each of the eighteen coefficient of friction data collection tests were collected in a three-dimensional plot (friction log) of the ratio of the tangential to the normal force versus relative displacement versus cycles as shown in figure 34. The plot in figure 34 is for a test with 440 microinches slip and 6,750 psi normal load. A corresponding plot of the ratio of tangent to normal forces versus relative displacement showing the hysteresis loops is in figure 35. The related plot of the ratio of tangent to normal forces versus log cycles is in figure 36. The mismatch in the scale in figure 36 is because of the 5.0° offset in the view as shown in figure 37.

The plots of the eighteen tests were analyzed and a curve fitting equation was developed which was used to compute frictional values for the coefficient of friction prediction method discussed in section 2.3. An example plot of the curve fitting equation values is shown by the plot in figure 38 which also corresponds to 440 microinches slip and 6,750 psi normal load. The variability of the coefficient of friction can be seen by comparing figures 36 and 38 with corresponding figures 39 and 40 which are for 800 microinches slip and 13,000 psi normal load.

Representative local frictional values resulting from application of the coefficient of friction prediction method to the verification system geometry are shown in table 5. Because for a given cycle count the four values across the array were the same for any distance along the fatigue loading direction, a single value is shown for each of the 16 element sets along the fatigue loading direction for each cycle count. These are shown for several cycle count values in table 5.

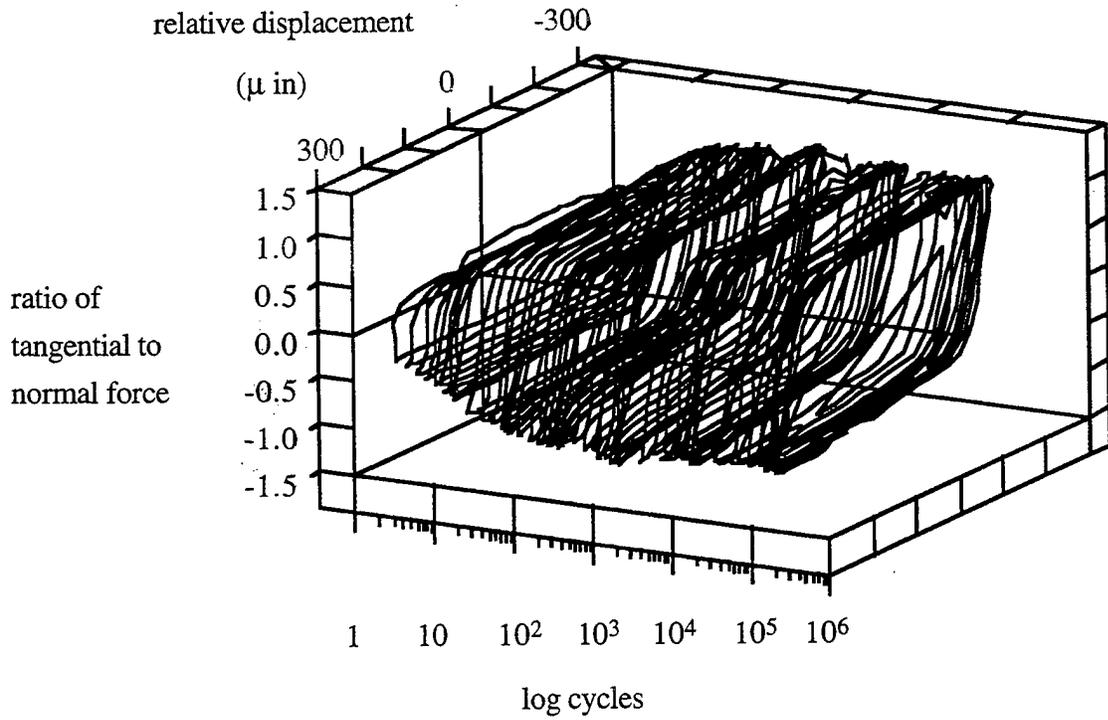


Figure 34. Typical friction log of coefficient of friction during fretting test.

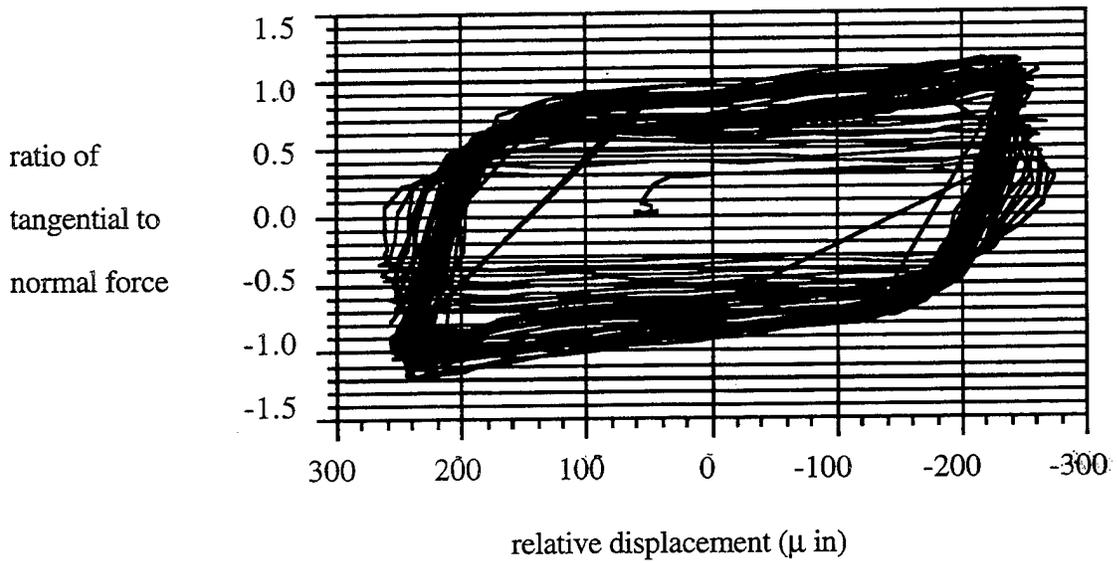


Figure 35. Example of hysteresis loops for coefficient of friction during fretting test (see figure 37 for an explanation of the view).

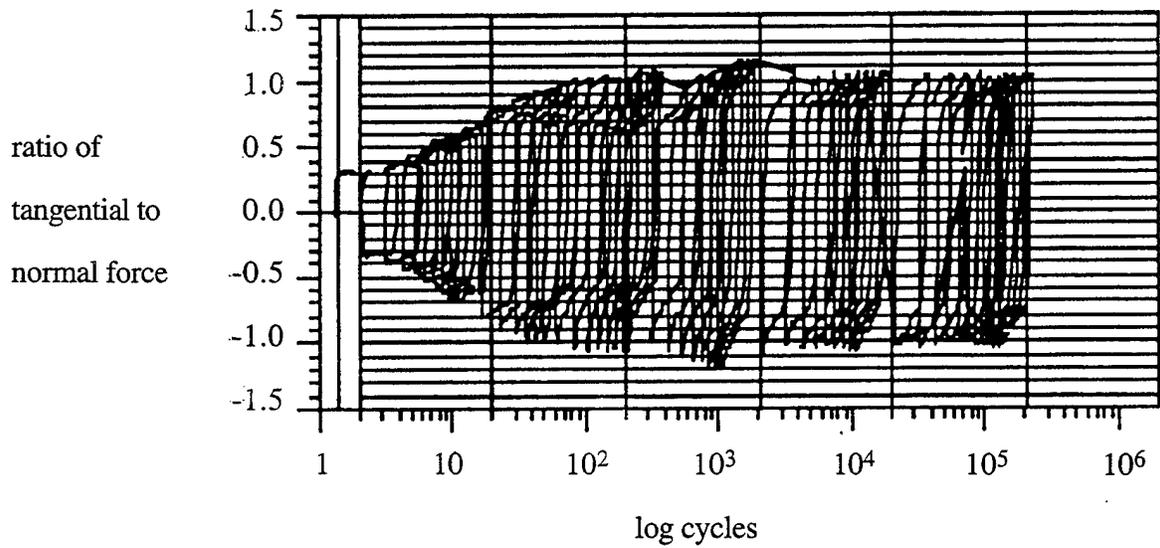


Figure 36. Example of the ratio of tangential to normal force versus cycles for a coefficient of friction during fretting test (440 microinches slip and 6,750 psi normal load). See figure 37 for an explanation of the view.

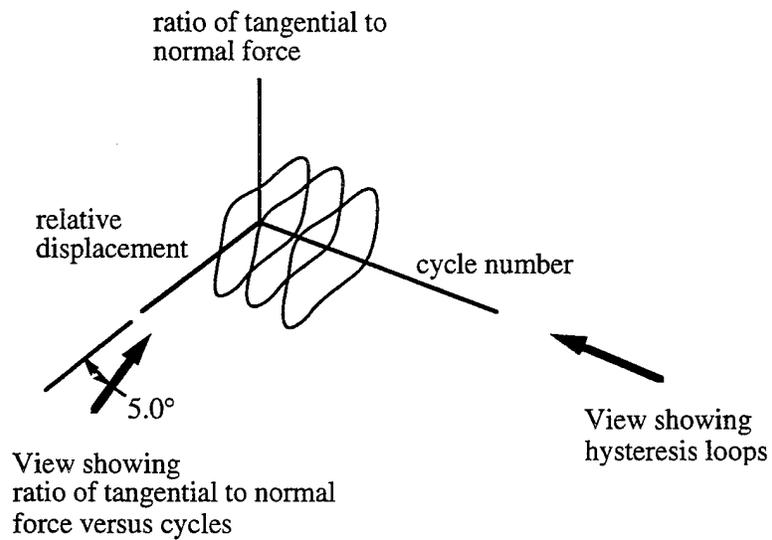


Figure 37. Friction log idealization with illustration of the different views of coefficient of friction during fretting test data plotted.

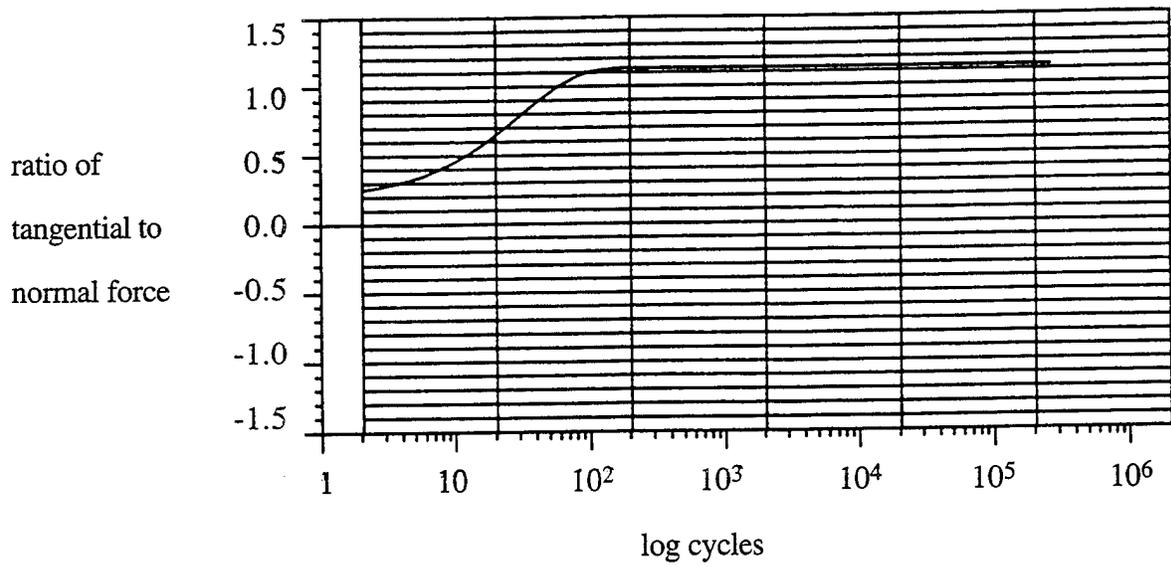


Figure 38. An example plot of the curve fitting equation values corresponding to 440 microinches slip and 6,750 psi normal load.

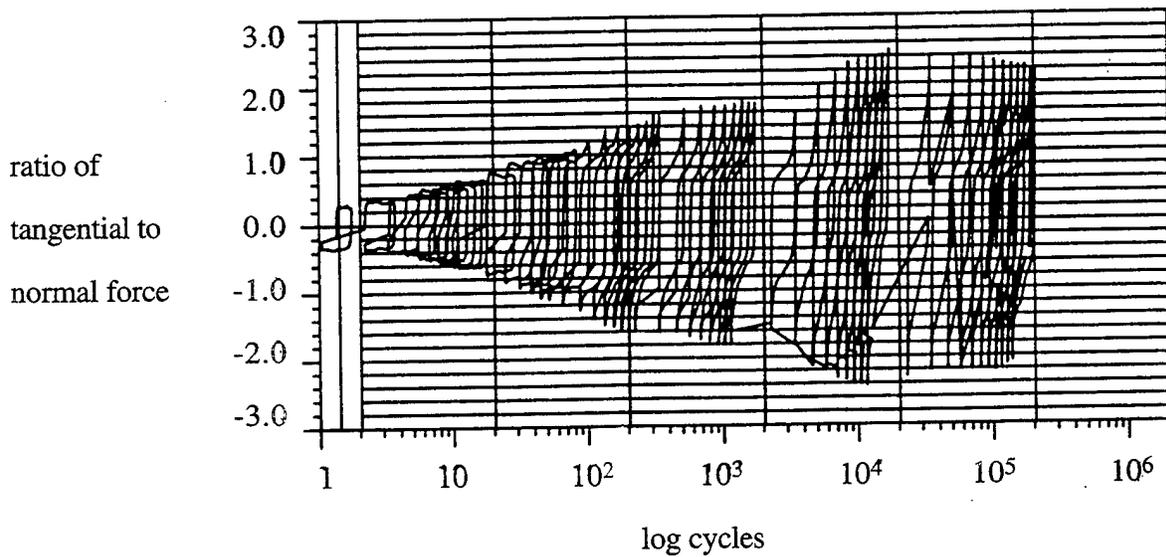


Figure 39. Example of the ratio of tangential to normal force versus cycles for a coefficient of friction during fretting test (800 microinches slip and 13,000 psi normal load).

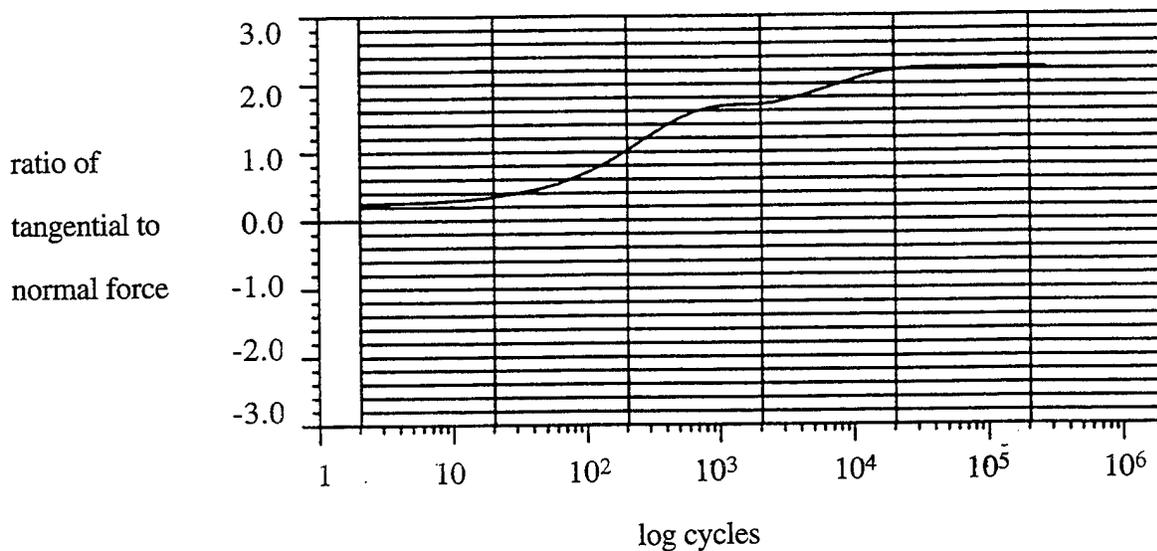


Figure 40. An example plot of the curve fitting equation values corresponding to 800 microinches slip and 13,000 psi normal load.

Table 5. Representative local coefficient of friction values resulting from application of the coefficient of friction prediction method to the verification system geometry. The top of the table is at the edge and the bottom is at the center of the fretting pad as measured along the axis of fatigue loading. The ditto marks mean the value is the same as the value above it.

Cycles --->	0	4	16	64	103	155	259	301	448	895
Pad edge	0.251	0.323	0.521	1.130	0.845	0.972	1.112	1.091	1.127	1.130
"	"	"	"	0.619	0.652	0.779	1.033	0.933	"	"
"	"	"	"	"	"	"	"	"	"	"
"	"	"	0.386	0.503	0.536	0.663	0.917	0.817	1.108	"
"	"	0.251	0.281	0.398	0.431	0.558	0.812	0.712	1.003	"
"	"	"	"	"	"	"	"	"	"	"
"	"	"	"	"	"	"	"	"	"	"
"	"	"	"	"	"	"	"	"	"	"
"	"	"	"	"	"	"	"	"	"	"
"	"	"	"	"	"	"	"	"	"	1.003
"	"	"	"	"	"	"	"	"	0.837	0.837
"	"	"	"	"	"	"	"	"	0.712	0.712
"	"	"	"	"	"	"	0.558	0.558	0.558	0.558
"	"	"	"	"	"	0.431	0.431	0.431	0.431	0.431
Pad center	"	"	"	"	0.368	0.368	0.368	0.368	0.368	0.368

A comparison of the strains predicted by the coefficient of friction prediction method and experimentally determined strains from the verification system test are shown in figure 41. Three experimental curves are shown as three of the eight possible symmetric locations were gaged during the test. Of these three gages, two were located on the same side of one fretting pad and the third gage was located on the other fretting pad.

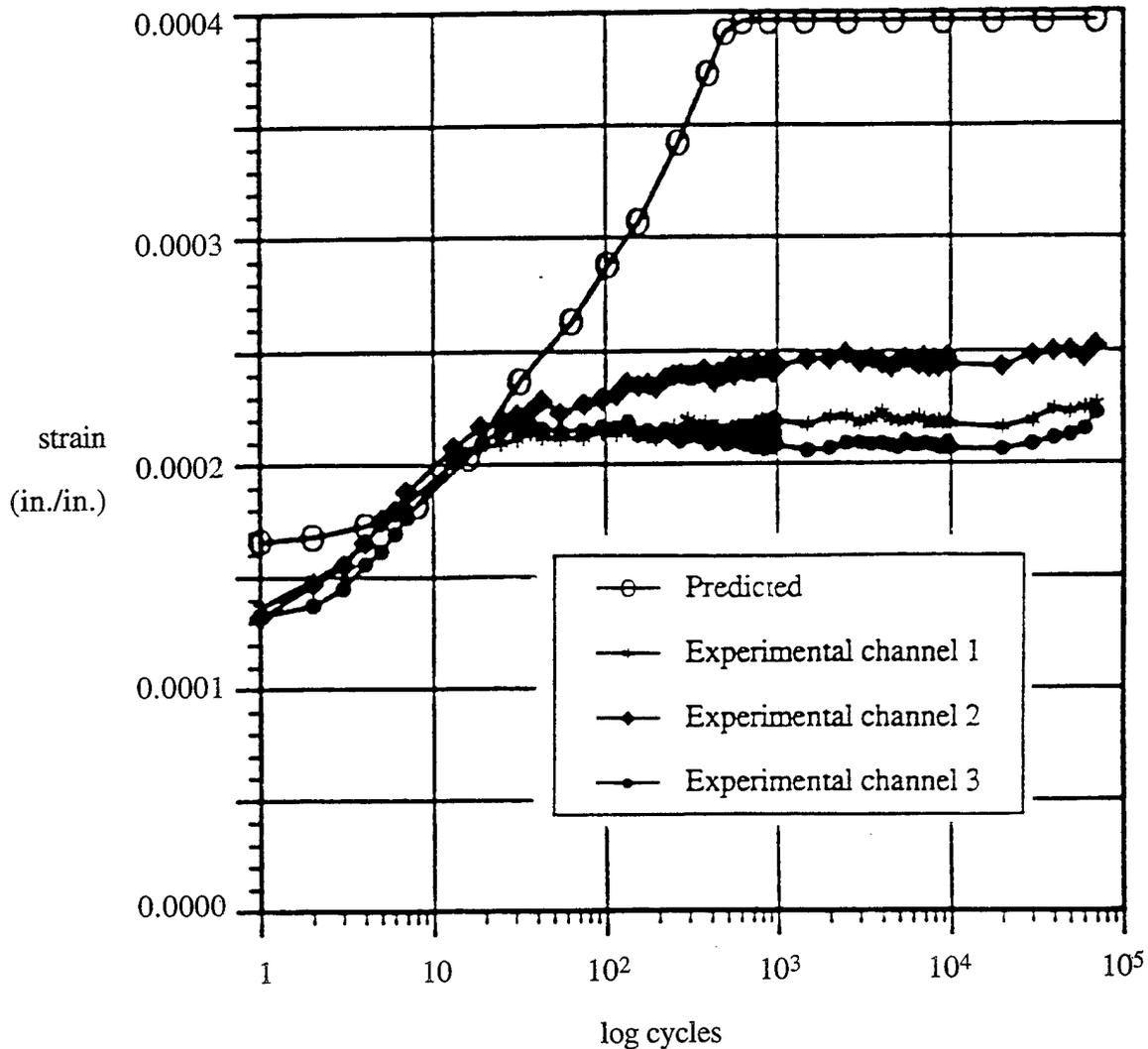


Figure 41. Predicted and experimentally determined strains of the verification apparatus.

### 3.4 RIVETED JOINT COMPUTER SIMULATION.

Figure 42 is a mapping of the fatigue loading direction stresses as defined in figure 16 at the point indicated by the "b" in figure 11. This is when the far field stress is returned to 170 psi after the first loading to 17,000 psi. Figure 43 is a mapping of the fatigue loading direction stresses as defined in figure 16 at the point indicated by the "c" in figure 11. This is when the far field stress is increased back to 17,000 psi.

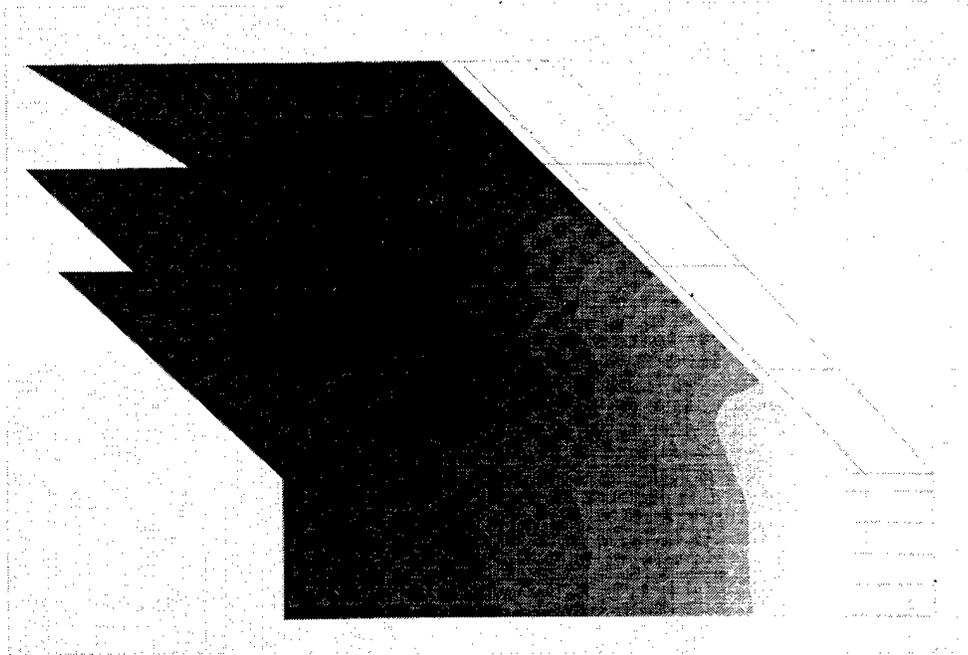


Figure 42. Fatigue loading direction stresses within the plate after the far field stress is increased to 17,000 psi, then decreased to 170 psi. Stresses are as discussed in figure 16. The stresses range from 15,000 psi (darkest) to -55,000 psi (lightest) in fifteen increments of 5,000 psi each.

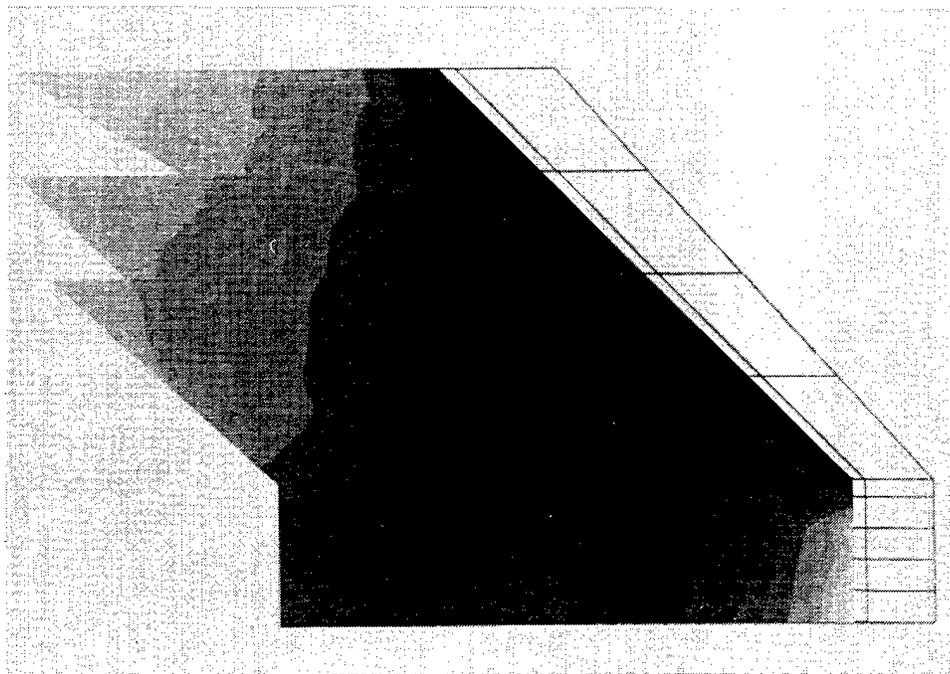


Figure 43. Fatigue loading direction stresses within the plate after the far field stress is increased to 17,000 psi, decreased to 170 psi, and returned to 17,000 psi. Stresses are as discussed in figure 16. The stresses range from 60,000 psi (darkest) to 18,000 psi (lightest) in fifteen increments of 3,000 psi each.

## 4 DISCUSSION.

### 4.1 INSPECTION OF A LAP JOINT PANEL FROM A SERVICE AIRCRAFT.

Some general comments can be made from this investigation. Multiple cracking was observed in the riveted joint panel that was examined. Virtually all rivet holes contained cracks. The longest of those that were measured was 0.025 in. long, as shown in figure 21. These cracks generally nucleated in regions of damage due to fretting. It is not known if these cracks were propagating during most of the life of the joint while in service or if they nucleated and grew to their present size then stopped or if nucleation required a relatively large portion of the life to this investigation and then the cracks propagated steadily. This type of information is necessary to determine the influence these cracks would have in a multiple-site damage assessment.

### 4.2 SENSITIVITY STUDY.

The data from table 2 were analyzed following the statistical procedures described in paragraph 2.2.

The life of specimen 14 (37,380 cycles) was compared with the mean life of the other CE/C-squeeze specimens (50,793 cycles) resulting in a "Z" value of 1.438 which is less than 2.326. Therefore, specimen 14 was retained as part of the sample. The reader may chose to eliminate this specimen for procedural reasons. Such elimination does not change the results of the analyses which follow.

The life of specimen 3R (120,050 cycles) was assumed not to be from the same population as the lives of the other FV/C-squeeze specimens. Therefore, its value was set aside while the mean value of the FV/C-squeeze sample from the initial testing program (162,690 cycles) was compared to that from the second program (162,813 cycles, without specimen 3R). Comparison of these samples resulted in a "Z" value of 0.007 which is less than 2.326. Therefore the two samples, without specimen 3R, were assumed to be from the same population. Comparison of the life for specimen 3R with the combined sample of all seven of the other FV/C-squeeze specimens (mean = 162,743 cycles and standard deviation = 17,471 cycles) resulted in a "Z" value of 2.444 which is greater than 2.326. Therefore the assumption that specimen 3R was not from the same population was justified and the data for specimen 3R were not considered further.

By inspection, results for those specimens which had not fractured (3, 13, and 10R) or had rivet holes where fatigue cracking was not observed (1 and 7R) were considered to be part of the populations of their respective samples. This was based on the fact that none of their fatigue lives were a low extreme value within the sample.

The CE/rivet gun sample from the initial test program (mean = 37,438 cycles) was compared to that from the second program (mean = 98,243 cycles) resulting in a "Z" value of 9.468. Additionally, the CE/C-squeeze sample from the initial test program (mean = 47,440 cycles) was compared to that from the second program (mean = 137,275 cycles) resulting in a "Z" value of 5.303. Because both of these "Z" values were greater than 2.326 and in both cases the greater lives were from the second test program, the hypothesis that the excessive countersink depth was the cause of decreased lives for the initial testing was considered to be validated. Therefore the CE data from the initial program were not considered further.

The CE/rivet gun data from the second test program (mean = 98,243 cycles) were compared to the CE/C-squeeze data from the second program (mean = 137,275 cycles) resulting in a "Z" value of 2.415 which is greater than 2.326. Therefore, for the 2017 CE riveted specimens, C-squeeze fretting fatigue lives were longer than rivet gun with bucking bar lives.

The FV/riquet gun data (mean = 251,040 cycles) were compared to the combined FV/C-squeeze data (mean = 162,743 cycles) resulting in a "Z" value of 2.347 which is greater than 2.326. Therefore, for the 7050 FV riveted specimens, rivet gun with bucking bar fretting fatigue lives were longer than C-squeeze lives. Also, it can be seen from inspection of table 2 (notes 2 and 3) that the cracked and fractured rivets were in the shorter lived FV/C-squeeze specimens. Also, those FV/C-squeeze specimens with fractured rivet heads had the shortest lives.

The CE/riquet gun data from the second test program (mean = 98,243 cycles) were compared to the FV/riquet gun data (mean = 251,040 cycles) resulting in a "Z" value of 4.326 which is greater than 2.326. Therefore, for the rivet gun with bucking bar procedure, 7050 FV riveted specimens had longer fretting fatigue lives than 2017 CE riveted specimens.

The CE/C-squeeze data from the second test program (mean = 137,275 cycles) were compared to the FV/C-squeeze data (mean = 162,743 cycles) resulting in a "Z" value of 1.650 which is less than 2.326. Therefore, at the 99 percent confidence level no conclusion could be drawn. However at the reduced 95 percent confidence level ("Z" = 1.645) the data from this test program indicated that for C-squeeze procedures, 7050 FV riveted specimens had longer fretting fatigue lives than 2017 CE riveted specimens.

#### 4.3 PREDICTION OF COEFFICIENT OF FRICTION DURING FRETTING.

Several observations can be made from the results of the verification test as shown in figure 41. The values of the predicted and experimentally determined strains were within 16% of each other during the first 40 to 50 cycles. Also, the general shapes of the predicted and experimentally determined strain curves were similar. However, after the first 1000 cycles the experimentally obtained values of strain were only about 56% of those predicted. This was considered to be a successful comparison given the complexity of the various input systems. Nonetheless, research to determine the reasons for the variation between the predicted and experimental values might provide insights into the local mechanisms of friction.

#### 4.4 RIVETED JOINT COMPUTER SIMULATION.

Figure 42 shows compressive stresses in the plate near where the shank of the rivet contacts the plate when the far field stress was reduced to 1% of its maximum value after the initial loading to 17,000 psi. This indicates that plasticity would have occurred in this region during the initial loading to 17,000 psi. From this, it can be inferred that residual stresses would be developed within the material(s) that would influence the overall stress state as cycling continued. This might be an advantage for cracks nucleated in the plate near where it is contacted by the shank of the rivet. The propagation of these cracks might be slower during the period in which the crack tip remained within the region influenced by the residual compressive stresses. The plasticity also could result in tipping of the rivet from its initial axis. This would change the stress and displacement conditions within the fretted interface adjacent to the rivet thereby influencing the local fretting fatigue process and possibly resulting in tractions due to friction that increased maximum shear stress and influenced yielding.

In figure 43 after reloading to 17,000 psi, the region of material near where the shank of the rivet contacts the plate attains a tensile stress of less than 25,000 psi while higher tensile stresses occur near the junction of the shank and countersink portions of the rivet.

## 5 CONCLUSIONS.

### 5.1 INSPECTION OF A LAP JOINT PANEL FROM A SERVICE AIRCRAFT.

It is concluded that fretting damage led to crack nucleation within the panel that was investigated. This is significant because it indicates a potential for multiple-site damage occurring more rapidly than might be anticipated from a fatigue analysis or testing that did not consider fretting. In addition, fretting also could produce cracks at holes not viewed as "fatigue critical."

### 5.2 SENSITIVITY STUDY.

Fretting damage led to crack nucleation in all failed specimens.

It is concluded with 99 percent confidence that with respect to fretting fatigue lives and based on the results of this research program:

For 2017 CE rivets, C-squeeze is a better riveting procedure than rivet gun with bucking bar.

For 7050 FV rivets, rivet gun with bucking bar is a better riveting procedure than C-squeeze.

If the rivet gun with bucking bar procedure is to be used it is better to use 7050 FV rivets than 2017 CE rivets.

It is concluded with 95 percent confidence that with respect to fretting fatigue lives and based on the results of this research program:

If the C-squeeze procedure is to be used it is better to use 7050 FV rivets than 2017 CE rivets. This coupled with the preceding conclusion leads to the conclusion that with respect to fretting fatigue lives and based only on the results of this research program, the better rivet is the 7050 FV rivet.

Rivet fracture by fatigue contributed to the decreased lives for applicable specimens from the combined FV/C-squeeze riveted sample.

### 5.3 PREDICTION OF COEFFICIENT OF FRICTION DURING FRETTING.

It is concluded that the method used to predict the coefficient of friction during fretting and the method of obtaining the material data are worth pursuing in further research. The results of such research might provide meaningful insights concerning the mechanisms of failure in riveted joints and other connections which involve fretting fatigue.

### 5.4 RIVETED JOINT COMPUTER SIMULATION.

The state of stress predicted by the model indicates that if cracks nucleated by fretting or other mechanisms at the surface of the plate which contacts the body of the rivet, the propagation of these cracks might be slower during the period in which the crack tip remained within the region influenced by the residual compressive stresses.

## 6 REFERENCES.

1. *Compilation of ASTM Standard Definitions, 6th Edition*, 1986.
2. Hoepfner, D. W. and Goss, G. L., "A Fretting-Fatigue Damage Threshold Concept," *Wear*, Vol. 27, 1974, pp. 61-70.
3. Piascik, R. S., Willard, S. A., and Miller, M., "The Characterization of Widespread Fatigue Damage in Fuselage Structure," NASA Conference Publication 3274, 1994.
4. O'Connor, Patrick D. T., *Practical Reliability Engineering*, John Wiley & Son, 1991.
5. Moesser, M. W., *Toward Prediction of Coefficient of Friction During Fretting*, Ph.D. dissertation, University of Utah, 1996.
6. Kinyon, S. E., *A Finite Element Analysis of a Riveted Joint*, masters thesis, University of Utah, 1994.
7. Vincent, L., Berthier, Y., and Godet, M., "Testing Methods in Fretting Fatigue: A Critical Appraisal," *Standardization of Fretting Fatigue Test Methods and Equipment*, M. H. Attia and R. B. Waterhouse (eds.), ASTM, 1992, pp. 33-48.

## APPENDIX A—PLAN FOR FABRICATION, FATIGUE TESTING, AND EXAMINATION OF RIVETED JOINT SPECIMENS (MARCH 20, 1995)

### 1. Overview

1.1. The Quality and Integrity Design Engineering Center (QIDEC) of the University of Utah is in the process of fabricating 18 riveted joint specimens to be used in fatigue testing. QIDEC is either contracting or performing all fabrication procedures as described in paragraph 2. This is a revision from the previous plan which assumed that QIDEC would contract with Macnab Manufacturing which would fabricate the specimens. QIDEC will perform all testing and examination of failed specimens in accordance with paragraph 3.

1.2. The 18 specimens will be fabricated from 0.063 gage 2024-T3 clad sheet aluminum alloy with riveting procedures as follow:

four C-squeeze joints with 7050 Briles rivets,  
four C-squeeze joints with 2017 CE rivets,  
five rivet gun with bucking bar joints with 7050 Briles rivets, and  
five rivet gun with bucking bar joints with 2017 CE rivets.

1.3. In the paragraphs that follow, reference is continually made to table A1 which shows the specimen preparation and testing protocol to be followed. Two sets of QIDEC drawings are also attached. The first gives the specific procedures to be followed in those cases where the specimen preparation work is not contracted outside of QIDEC. The second shows the system to be used during the fatigue testing to counter the tendency of the riveted joints to bend in the riveted area when under tensile loading.

### 2. Joint Fabrication Procedures

2.1. QIDEC obtained 36 6-in.-wide coupons of 2024-T3 clad sheet aluminum, a bag of CE rivets and a bag of Briles rivets.

2.1.1. The coupons were numbered consecutively as unpacked and the thickness of each end of each coupon was measured twice with a digital micrometer and recorded. Based on these measurements, the coupons were paired for manufacturing into sets such that the coupon thickness in the coupon where the rivet heads are to be located was the same for all joints, and the coupon thicknesses adjacent to the bucktails were conducive to experimental blocking. Otherwise, random number generation was used to match coupons and, within each experimental block, to determine rivet type, riveting procedure, the order of machining, and the order of riveting. The results of this sorting procedure are shown in table A1.

2.1.2. Rivets from the bags of rivets were measured until sufficient rivets of each type with identical measurements were available for all joints to be manufactured. These dimensions were 0.1875 in. diameter by 0.3475 in. length (including head) for the Briles rivets and 0.188 in. diameter by 0.346 in. length (including head) for the CE rivets.

2.2. The 36 coupons were allodined in a single batch in accordance with MIL-C-5541 class 1A.

2.3. The 36 coupons were coated in a single batch with BMS 10-11 type 1 class A adhesive primer.

2.4. Each coupon will be machined to size in accordance with QIDEC drawings FAASPC94042 and FAASPC94043.

2.5. Each set of coupons will be clamped together in a machining fixture and not separated until all of the machining in the rivet area is completed.

2.6. All sets will be machined and marked in accordance with QIDEC drawings FAASPC94044 and FAASPC94045 using one milling machine. The order of machining will be as shown in table A1. Initially, pilot holes will be drilled with a centering drill to ensure alignment. Number 10 double margin drills ground to 0.192 in. diameter will be used to drill all rivet holes. Countersink stops will be used to drill all countersinks in order to ensure consistent countersink depth. These stops were set based on inspection of holes drilled and countersunk during pilot machining work sessions. The settings will not be changed during the specimen machining program. The specific drill and countersink used to machine each hole will be identified and documented.

2.7. Rivet hole diameters will be measured using a Meyer C-10 gage set and recorded for all rivet holes in each coupon where the rivet heads are to be located.

2.8. Joints will be riveted in accordance with QIDEC drawings FAASPC94046, FAASPC94047, FAASPC94048, and FAASPC94049. The order of riveting will be as shown in table A1. All specimens within a block will be riveted during the same riveting session. Procedures for each block and between blocks will be as consistent as possible and documented for each rivet, specimen, and block. Specifically, the same person will perform the same tasks using the same equipment and procedures during riveting of all specimens to the extent feasible and appropriate. The diameter of each bucktail will be determined to be within specifications by use of go/no-go gages. Environmental (temperature and relative humidity) and lighting conditions will be as consistent as possible. Temperature and relative humidity will be recorded at the time that the riveting of each specimen is begun.

2.9. Rivet head clearances will be computed and recorded for all rivets in each specimen based upon measurements using a machinists microscope.

### 3. Joint Fatigue Testing and Examination Procedures

3.1. All testing will be conducted on the same calibrated MTS servo hydraulic tension testing load system. Prior to testing, the load system will be aligned by direct alignment of the grips. Additionally, alignment will be verified prior to and following testing of each block of specimens (see table A1) using an alignment specimen to be produced by QIDEC.

3.2. Testing will be conducted sequentially within four test blocks. The order of testing, based on random number generation within each block, will be as shown in table A1. Fatigue loading will be with a 10 Hz sine wave driving signal, a maximum nominal tensile stress of 17 ksi, and a stress ratio of  $R = 0.1$ . The system shown in the attached drawings will be used during the fatigue testing to counter the tendency of the riveted joints to bend in the riveted area when under tensile loading. Temperature and relative humidity will be recorded at the time that the testing of each specimen is begun. Total cycles to fracture will be determined for each specimen.

3.3. Following testing, failed joints will be stored in a desiccant chamber when they are not being examined.

3.4. The joints will be examined optically with little or no magnification and using an optical microscope to identify probable locations and causes of crack nucleation. Then, metallographic and fractographic investigations will be performed to confirm the causes of crack nucleation and identify the mechanisms involved in the process. Photographs will be taken as appropriate.

Table A1  
Riveted Joint Work Order Plan

Block (Sub-block) No.	Joint		Countersink <sup>1</sup> Coupon		Bucktail <sup>2</sup> Coupon		Rivet Type <sup>3</sup>	Riveting Procedure <sup>4</sup>	Machining Order <sup>5</sup>	Riveting Order <sup>5</sup>	Test Order <sup>5</sup>
	No.	No.	No.	Thickness	No.	Thickness					
1	1	31	0.064 in	7	0.0630 in	Briles	Rivet gun	1	10	9	
	2	5	"	29	"	Briles	C-squeeze	2	8	7	
	3	19	"	32	"	CE	C-squeeze	3	7	8	
	4	35	"	36	"	CE	Rivet gun	4	9	10	
2	5	23	"	2	0.0635 in	Briles	Rivet gun	7	12	14	
	6	14	"	8	"	CE	C-squeeze	6	13	11	
	7	18	"	17	"	CE	Rivet gun	5	14	13	
	8	25	"	27	"	Briles	C-squeeze	8	11	12	
3	9	12	"	9	0.0645 in	CE	Rivet gun	12	17	18	
	10	30	"	16	"	CE	C-squeeze	11	18	16	
	11	28	"	24	"	Briles	C-squeeze	10	16	15	
	12	4	"	26	"	Briles	Rivet gun	9	15	17	
4(C)	13	3	"	11	0.0640 in	Briles	C-squeeze	14	1	3	
	14	33	"	20	"	CE	C-squeeze	16	2	2	
4(RG-)	15	1	"	6	0.0635 in	CE	Rivet gun	13	3	5	
	16	22	"	34	"	Briles	Rivet gun	18	4	6	
4(RG+)	17	15	"	13	0.0650 in	Briles	Rivet gun	17	6	4	
	18	10	"	21	"	CE	Rivet gun	15	5	1	

1. Chosen randomly from coupons with the same thickness.

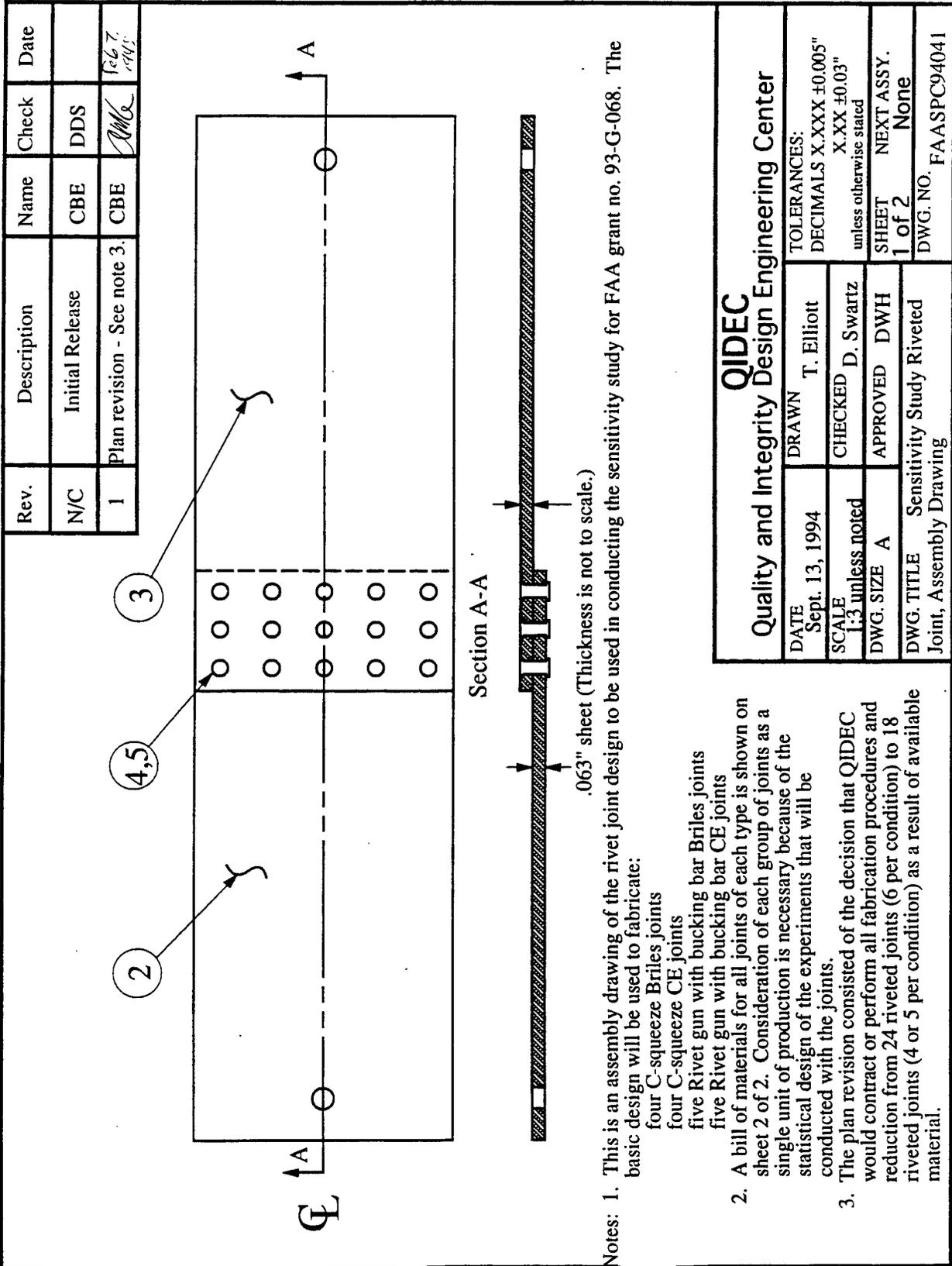
2. Chosen randomly from coupons with the same thickness within each block or sub-block.

3. Two of each type chosen randomly within each block or one of each type chosen randomly within each sub-block.

4. Chosen randomly to cause one of each of the four type/procedure combinations within each block for blocks 1 through 3. Chosen to cause bucktail coupon thicknesses for rivet gun specimens to bracket C-squeeze thicknesses for block 4.

5. Chosen to have all specimens within a block machined/riveted/tested sequentially then randomly within each block.

APPENDIX B—FABRICATION DRAWINGS FOR SENSITIVITY STUDY SPECIMENS



**QIDEC**  
Quality and Integrity Design Engineering Center

DATE	Sept. 13, 1994	DRAWN	T. Elliott	TOLERANCES:	DECIMALS X.XXX ±0.005"
SCALE	1:3 unless noted	CHECKED	D. Swartz		X.XX ±0.03"
DWG. SIZE	A	APPROVED	DWH		unless otherwise stated
DWG. TITLE	Sensitivity Study Riveted Joint, Assembly Drawing			SHEET	NEXT ASSY.
				1 of 2	None
				DWG. NO.	FAASPC94041

Rev.		Description		Name	Check	Date
N/C		Initial Release		CBE	DDS	
1		See note 1.		CBE	<i>[Signature]</i>	6-16-94
60	75	Rivet, BACR15-CE6D6	N/A			5
60	75	Rivet, Briles BACR15-FV6KE6	N/A			4
1/2 qt.	1/2 qt.	BMS 10-11 type 1 class A epoxy primer	N/A			
1 pt.	1 pt.	MILS - 81733 type 4 faying surface sealant	N/A			
4	5	13.43" coupon	FAASPC94043-1			3
4	5	13.39" coupon	FAASPC94042-1			2
	5	Rivet gun CE joint	-4 (Machining: FAASPC94045) (Riveting: FAASPC94049)			
	5	Rivet gun Briles joint	-3 (Machining: FAASPC94044) (Riveting: FAASPC94048)			
	4	C-squeeze CE joint	-2 (Machining: FAASPC94045) (Riveting: FAASPC94047)			
4		C-squeeze Briles joint	-1 (Machining: FAASPC94044) (Riveting: FAASPC94046)			
4/-1	5/-3	Sensitivity Study Riveted Joint-Assy.	FAASPC94041			
Quantity/ Assembly		Description		Drawing No.	Item No.	

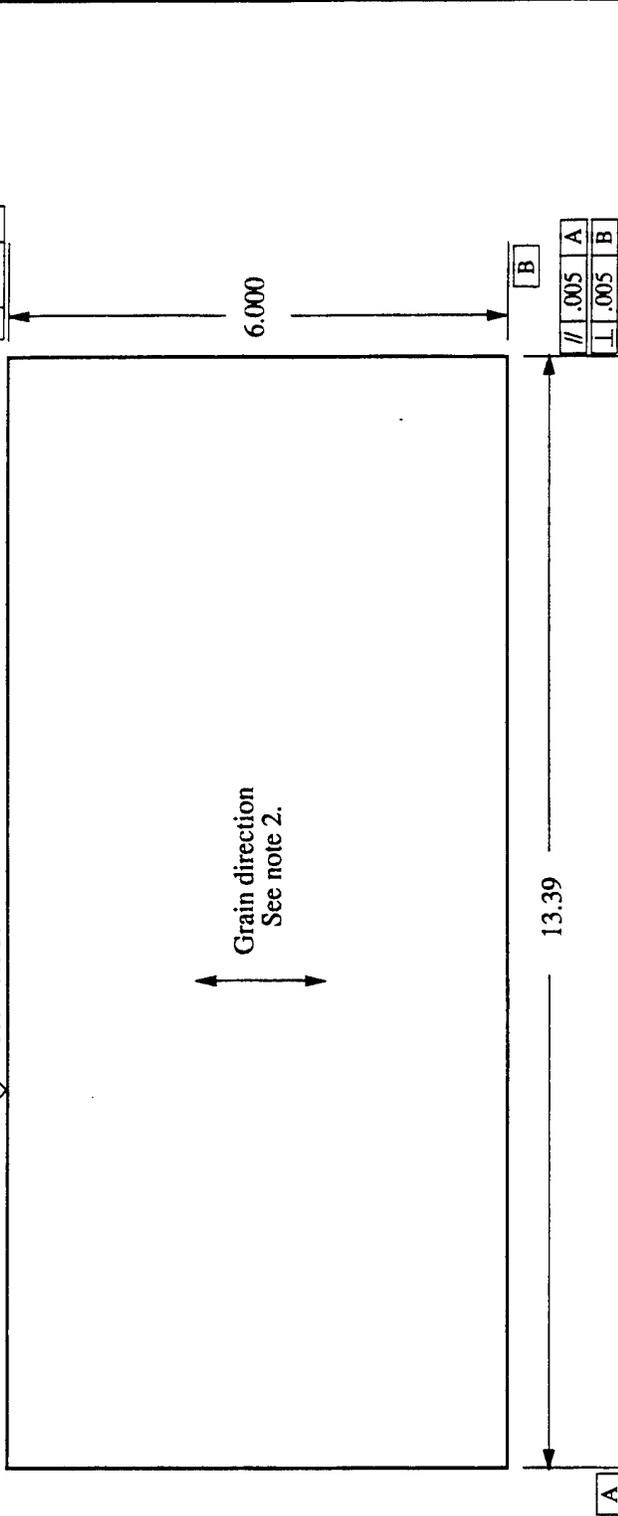
Note 1. Changes are reductions from 6 joints per condition to 4 or 5 with corresponding reductions in rivets from 90 to 60 or 75 and addition of primer and sealant.

**QJDEC**  
Quality and Integrity Design Engineering Center

DATE Sept. 13, 1994	DRAWN T. Elliott	TOLERANCES: DECIMALS X.XXX ±0.005" X.XX ±0.03" unless otherwise stated
SCALE N/A	CHECKED D. Swartz	SHEET NEXT ASSY. 2 of 2 None
DWG. SIZE A	APPROVED DWH	DWG. NO. FAASPC94041
DWG. TITLE Sensitivity Study Riveted Joint, Assembly Drawing		

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	DDS	
1	Plan revision - See note 4.	CBE	UMC	Feb 7, 1995

~~32~~ See note 3.



- Notes: 1. All dimensions are in inches.  
 2. Coupons provided to QIDEC were cut in the correct orientation with respect to the grain direction.  
 3. All edges should be machined to 32 microinches. Sheet surfaces should be left in the as-clad condition. Break all edges .010.  
 4. Changes are reduction of quantity required from 24 to 18 and wording of note 2.

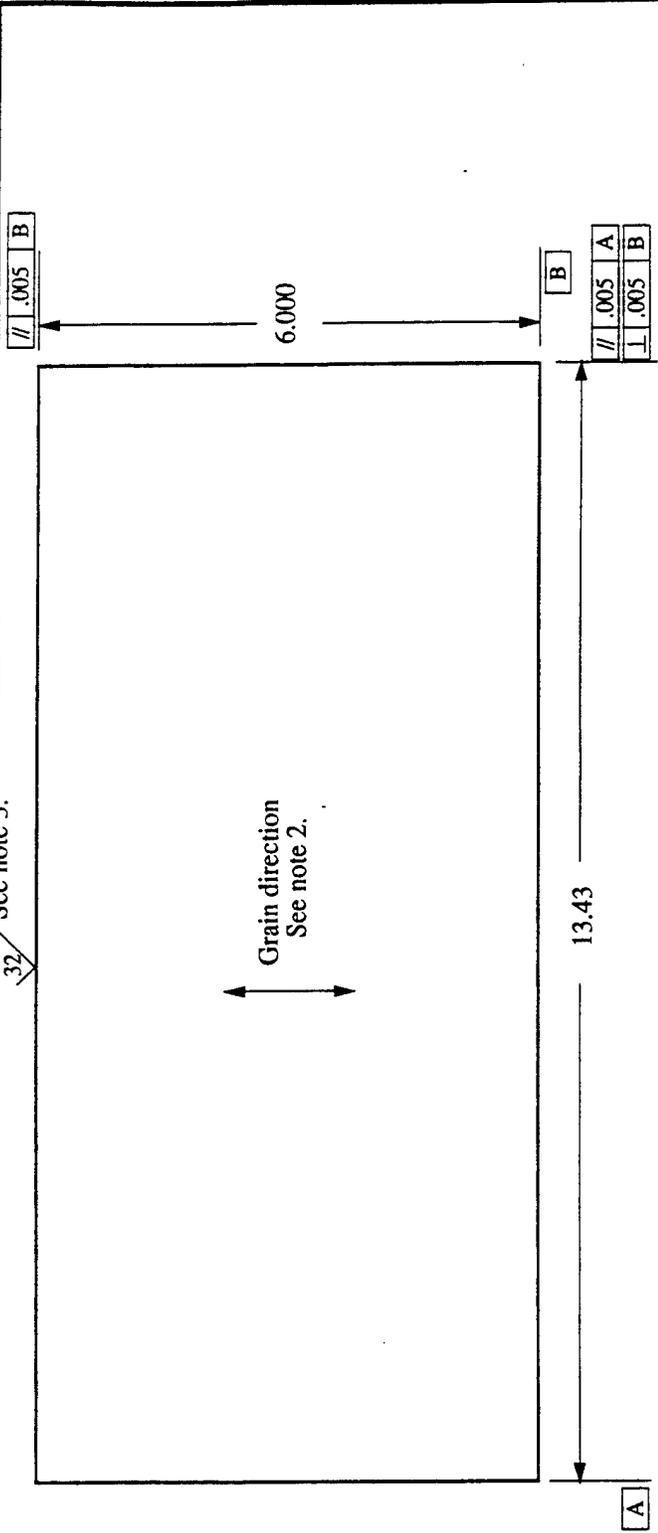
18	-1	13.39" long coupon	2024-T3 per Fed. Spec. QQ-A-250/5	clad/ note 2	.063" sheet	N/A
Quan. Req.	Part No.	Description	Material/ Specification	Finish	Stock Size	Part Mark

## QIDEC Quality and Integrity Design Engineering Center

DATE	Sept 14, 1994	DRAWN	T. Elliott	TOLERANCES:	DECIMALS X.XXX ±0.005"
SCALE	NTS	CHECKED	D. Swartz		X.XX ±0.03"
DWG. SIZE	A	APPROVED	DWH		unless otherwise stated
DWG. TITLE	Sensitivity Study Riveted Joint, 13.39" Coupon			SHEET	NEXT ASSY.
				1 of 1	FAASPC94041
					DWG. NO. FAASPC94042

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	DDS	
1	Plan revision - See note 4	CBE	AMC	Feb 7 1995

32/ See note 3.

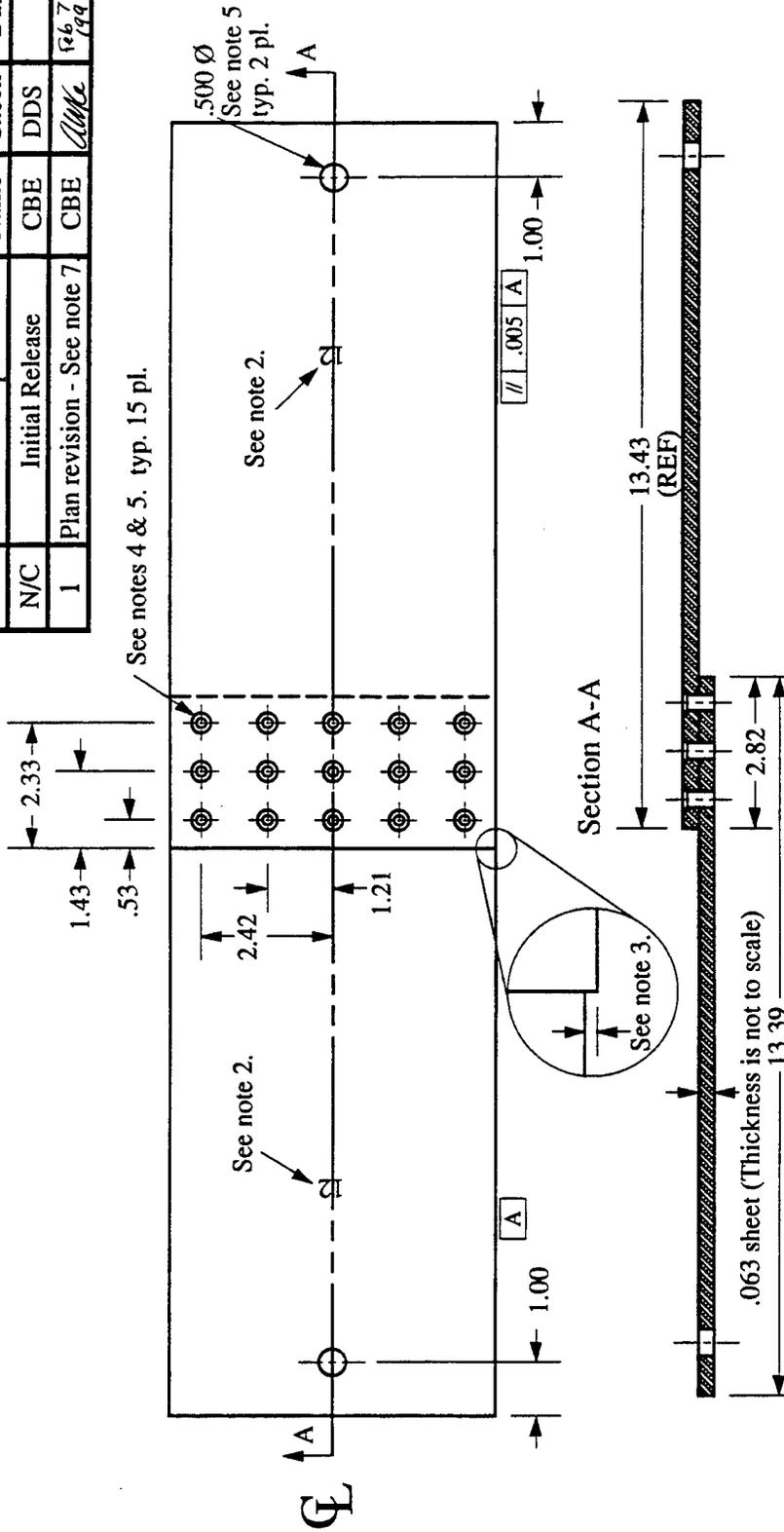


- Notes: 1. All dimensions are in inches.  
 2. Coupons provided to QIDEC were cut in the correct orientation with respect to the grain direction.  
 3. All edges should be machined to 32. microinches. Sheet surfaces should be left in the as-clad condition. Break all edges .010.  
 4. Changes are reduction of quantity required from 24 to 18 and wording of note 2.

<b>QIDEC</b> Quality and Integrity Design Engineering Center		DATE	Sept 14, 1994	DRAWN	T. Elliott	TOLERANCES:	DECIMALS X.XXX ±0.005"
		SCALE	NTS	CHECKED	D. Swartz		X.XX ±0.03"
QTY	18	DESCRIPTION	13.43" long coupon	FINISH	clad/ note 2		unless otherwise stated
MATERIAL	2024-T3 per Fed. Spec. QQ-A-250/5	STOCK SIZE	.063" sheet	APPROVED	DWH	SHEET	NEXT ASSY.
REDUCED QTY	18	REDUCED QTY REASON	reduction of quantity required from 24 to 18 and wording of note 2.			1 of 1	FAASPC94041
							DWG. NO. FAASPC94043

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	DDS	
1	Plan revision - See note 7	CBE	AMC	06.7 1995

See notes 4 & 5. typ. 15 pl.



7. Changes are number of joints and machining instructions in note 2, hole diameter and head clearance in note 4, and elimination of original note 6.

### QIDEC

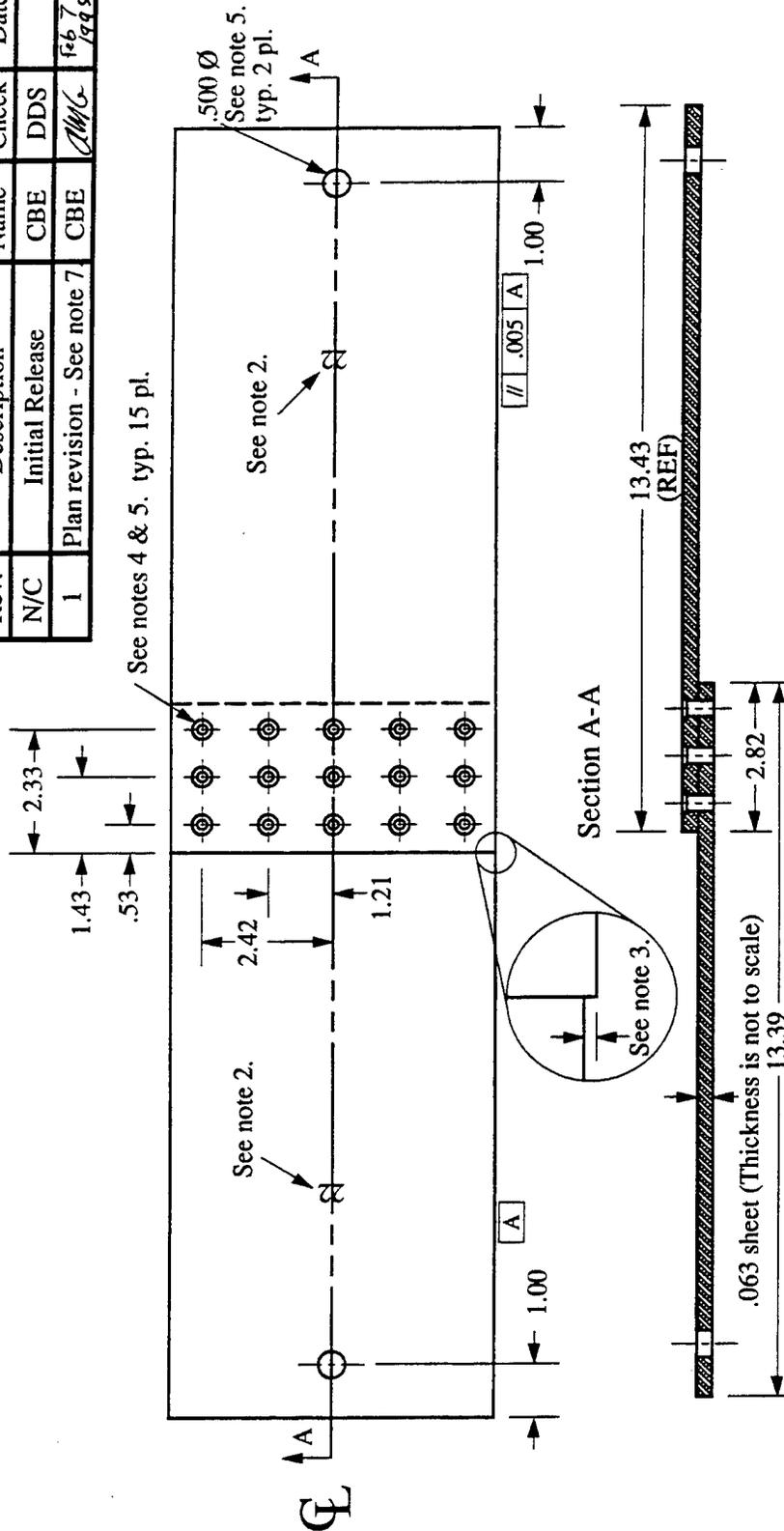
Quality and Integrity Design Engineering Center

DATE	Sept. 15, 1994	DRAWN	T. Elliott	TOLERANCES:	DECIMALS X.XXX ±0.005"
SCALE	1:3 unless noted	CHECKED	D. Swartz		X.XX ±0.03"
DWG. SIZE	A	APPROVED	DWH		unless otherwise stated
DWG. TITLE	Sensitivity Study Riveted Joint, Machining, Briles Riveted Joints			SHEET	NEXT ASSY.
				1 of 1	FAASPC94041
				DWG. NO.	FAASPC94044

1. All dimensions are in inches.
2. Nine joints will be produced from this drawing in accordance with the machining plan. They should be marked 4 inches from each joint end by a vibrator marker.
3. Maximum mismatch .005.
4. Hole diameter .192 ± .001 (no. 10 double margin drill) with 120° Briles countersink to a depth which will allow clearance of .003 ± .002 of rivet head above the joint surface (expected .043 ± .001 depth).
5. Deburr all holes.
6. Separate the coupons, remove chips and clean the surfaces with a dry cloth.

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	DDS	
1	Plan revision - See note 7	CBE	AMG	Feb 7 1995

See notes 4 & 5. typ. 15 pl.



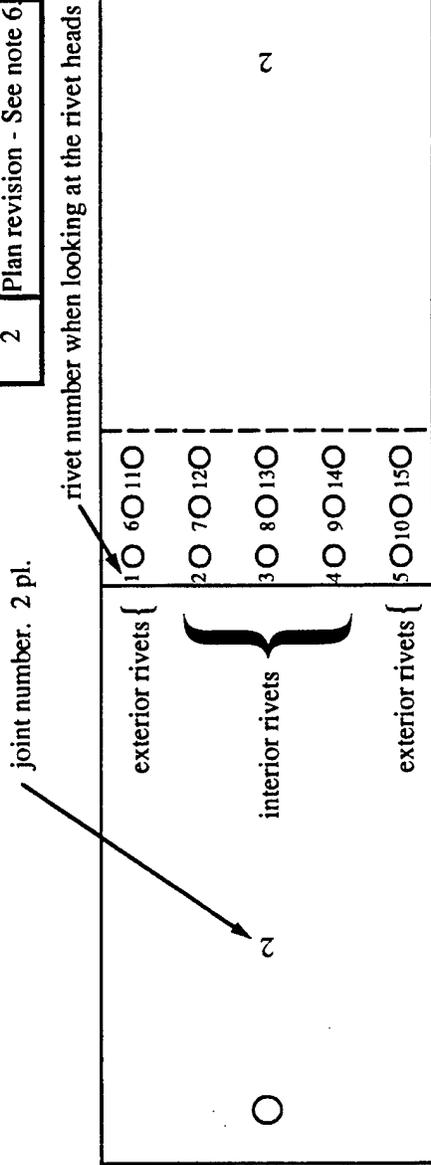
7. Changes are number of joints and machining instructions in note 2, hole diameter and head clearance in note 4, and elimination of original note 6.

**QIDEC**  
**Quality and Integrity Design Engineering Center**

DATE	DRAWN	TOLERANCES:
Sept. 15, 1994	T. Elliott	DECIMALS X.XXX ±0.005"
SCALE	CHECKED	X.XX ±0.03"
1:3 unless noted	D. Swartz	unless otherwise stated
DWG. SIZE	APPROVED	SHEET
A	DWH	1 of 1
		NEXT ASSY.
		FAASPC94041
		DWG. NO. FAASPC94045

- Notes:
1. All dimensions are in inches.
  2. Nine joints will be produced from this drawing in accordance with the machining plan. They should be marked 4 inches from each joint end by a vibrator marker.
  3. Maximum mismatch .005.
  4. Hole diameter .192 ±.001 (no. 10 double margin drill) with 100° countersink to a depth which will allow clearance of .003 ±.002 of rivet head above the joint surface (.047 ±.001 depth).
  5. Deburr all holes
  6. Separate the coupons, remove chips and clean the surfaces with a dry cloth.

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	DDS	
1	Plan revision - See note 5.	CBE	AMG	
2	Plan revision - See note 6.	CBE	<i>AMG</i>	March 20, 1995



- Notes:
1. Roll on MILS - 81733 type 4 (e.g. Pro Seal 870-C12) faying surface sealant to .003" ±.001" thickness.
  2. Reassemble the part with alignment pins.
  3. The order of riveting for each joint should be as follows:  
 exterior rivets: 12, 13, 14, 4, 3, 2, 7, 8, 9  
 interior rivets: 1, 15, 5, 11, 6, 10
  4. Establish procedures to drive rivets to obtain .263" + .010"/-.000" button (bucktail) diameter for interior rivets and .273" + .010"/-.000" button diameter for exterior rivets. However, once the procedure is established and riveting is started retain the same procedure for all specimens of the same type.
  5. Changes are elimination of original note 1, reduction in quantity required from 6 to 4, changes in joint numbers and changes in current note 4 to reflect the change in the number of joints.
  6. Changes are to note 4 to account for riveting in blocks of specimens instead of by type.

QIDEC		Quality and Integrity Design Engineering Center	
DATE	DRAWN	T. Elliott	TOLERANCES: DECIMALS X.XXX ±0.005" X.XX ±0.03" unless otherwise stated
SCALE	NTS	CHECKED D. Swartz	SHEET NEXT ASSY 1 of 1 FAASPC94041
DWG. SIZE	A	APPROVED DWH	DWG. NO. FAASPC94046
DWG. TITLE	Sensitivity Study Riveted Joint, Riveting, C-squeeze Briles Joints		
4	2, 8, 11, 13	C-squeeze Briles joint	
Quan. Req.	Joint No.	Description	Material/ Specification
			Finish
			Stock Size
			Part Mark

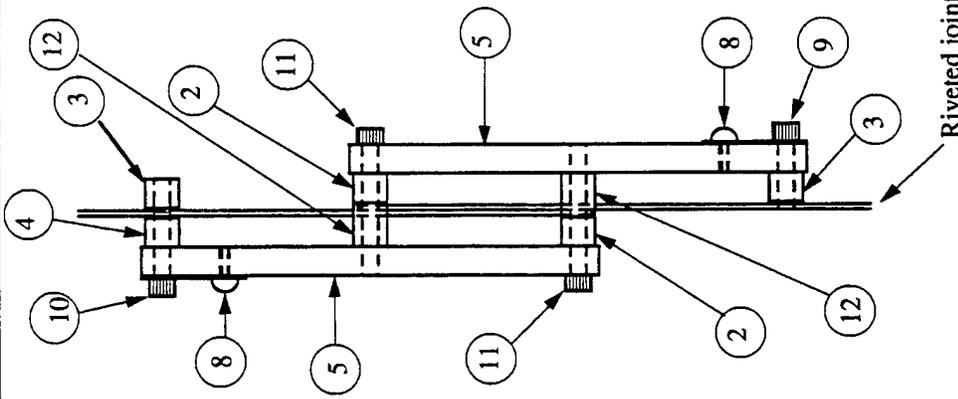
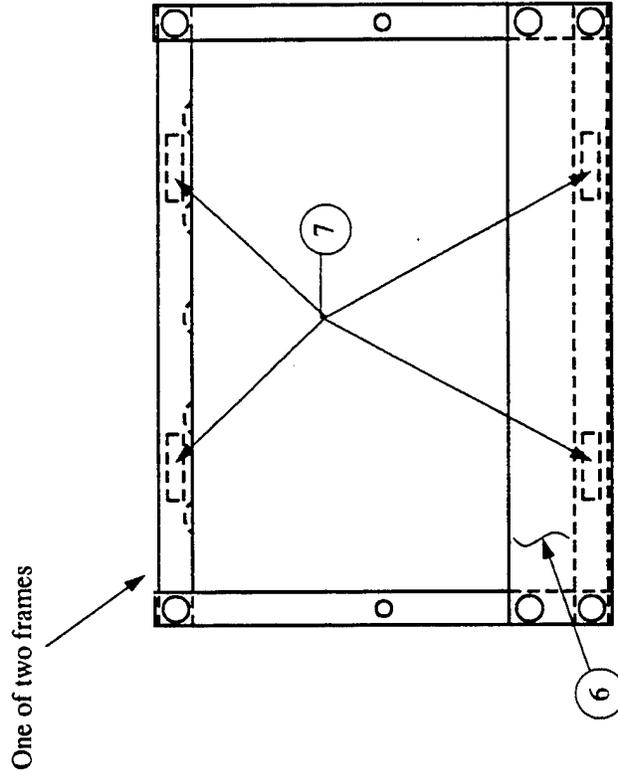






APPENDIX C—DRAWINGS OF ANTIBENDING SYSTEM

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	Addition of spacers - parts 12	CBE	<i>Ula</i>	12/67/1995



- Notes: 1. This is an assembly drawing of the system to be used to counter the tendency of a riveted joint (QIDEC drawing FAA/SPC94041) to bend when under tensile loading conditions.  
 2. The bill of materials is on sheet 2 of 2.

**QIDEC**  
 Quality and Integrity Design Engineering Center

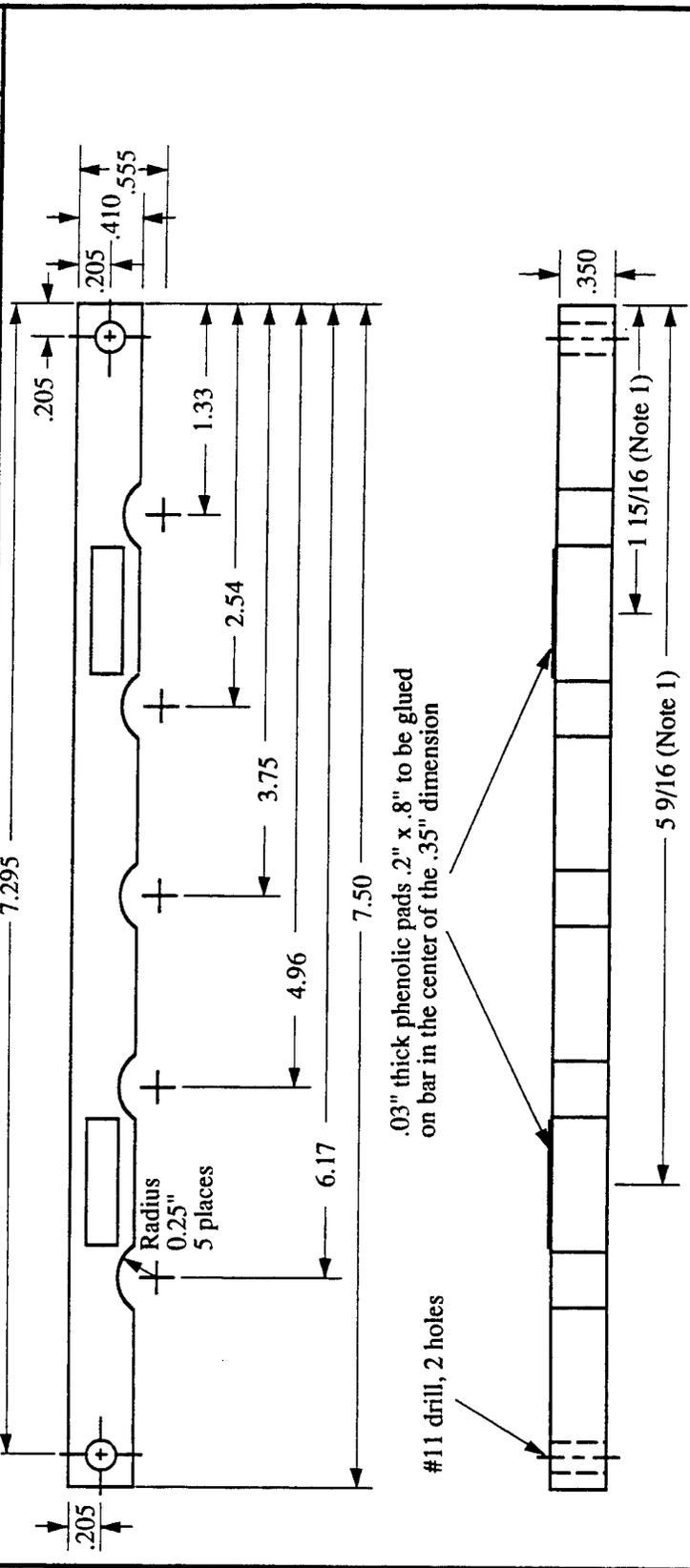
DATE	Jan. 24, 1995	DRAWN	C. B. Elliott	TOLERANCES:	DECIMALS X.XXX ±0.005"
SCALE	1:2	CHECKED	A. Georgeson		X.XX ±0.03"
DWG. SIZE	A	APPROVED	C. B. Elliott		unless otherwise stated
DWG. TITLE	Assy. Drawing - Riveted Joint Stiffner System				
				SHEET	NEXT ASSY. 1 of 2
				DWG. NO.	FAAMSC95036

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	Addition of Item No. 12	CBE	<i>Be</i>	Feb 7, 1995

Quantity	Description	Drawing No.	Item No.
2	Constraint beam at joint (Al alloy)	FAAMSC95037	2
2	Constraint beam away from joint - threaded (Al alloy)	FAAMSC95038	3
1	Constraint beam away from joint - unthreaded (Al alloy)	FAAMSC95039	4
4	Side beam (Al alloy)	FAA MSC95040	5
2	Shape plate (Al alloy)	FAAMSC95041	6
8	.2" x .8" x .032" thick micarta pads		7
4	1/4" #8-32 machine screws		8
2	5/8" #10-32 socket head cap screws		9
2	1" #10-32 socket head cap screws		10
4	1 1/2" #10-32 socket head cap screws		11
4	Spacer (Al alloy)	FAAMSC95042	12

<b>QIDEC</b> Quality and Integrity Design Engineering Center		DATE	Jan. 24, 1995	DRAWN	C. B. Elliott	TOLERANCES:	
		SCALE	N/A	CHECKED	A. Georgeson	DECIMALS X.XXX ±0.005" X.XX ±0.03" unless otherwise stated	
		DWG. SIZE	A	APPROVED	C. B. Elliott	SHEET	NEXT ASSY.
		DWG. TITLE	Assy. Drawing - Riveted Joint Stiffner System			2 of 2	none
						DWG. NO. FAAMSC95036	

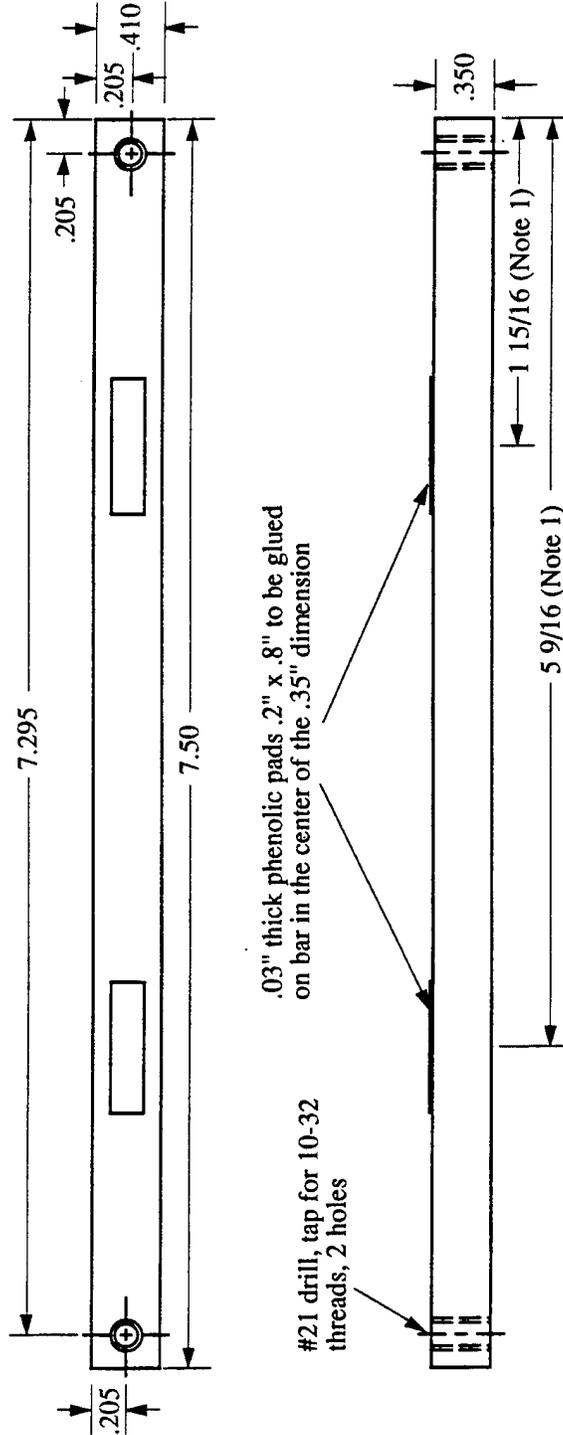
Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	.175 to .205 three times, .350 to .410, .525 to .555, 7.325 to 7.295	CBE	AG	Feb 7 1995



2		- 1	beam	Al alloy	as mach.	N/A	none	QIDEC	
								Quality and Integrity Design Engineering Center	
Quan. Req.	Part No.	Description	Material/ Specification	Finish	Stock Size	Part Mark	DATE Jan. 24, 1995 DRAWN C. B. Elliott CHECKED A. Georgeson APPROVED C. B. Elliott DWG. TITLE Riveted Joint Stiffner System - Constraint Beam at Joint		
							TOLERANCES: DECIMALS X.XXX ±0.005" X.XX ±0.03" unless otherwise stated		
							SHEET NEXT ASSY. 1 of 1 FAAMSC95036 DWG. NO. FAAMSC95037		

Notes: 1. These dimensions are to the centers of the phenolic pads.

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	.175 to .205 three times, .350 to .410, 7.325 to 7.295	CBE	AG	Feb 7, 1995



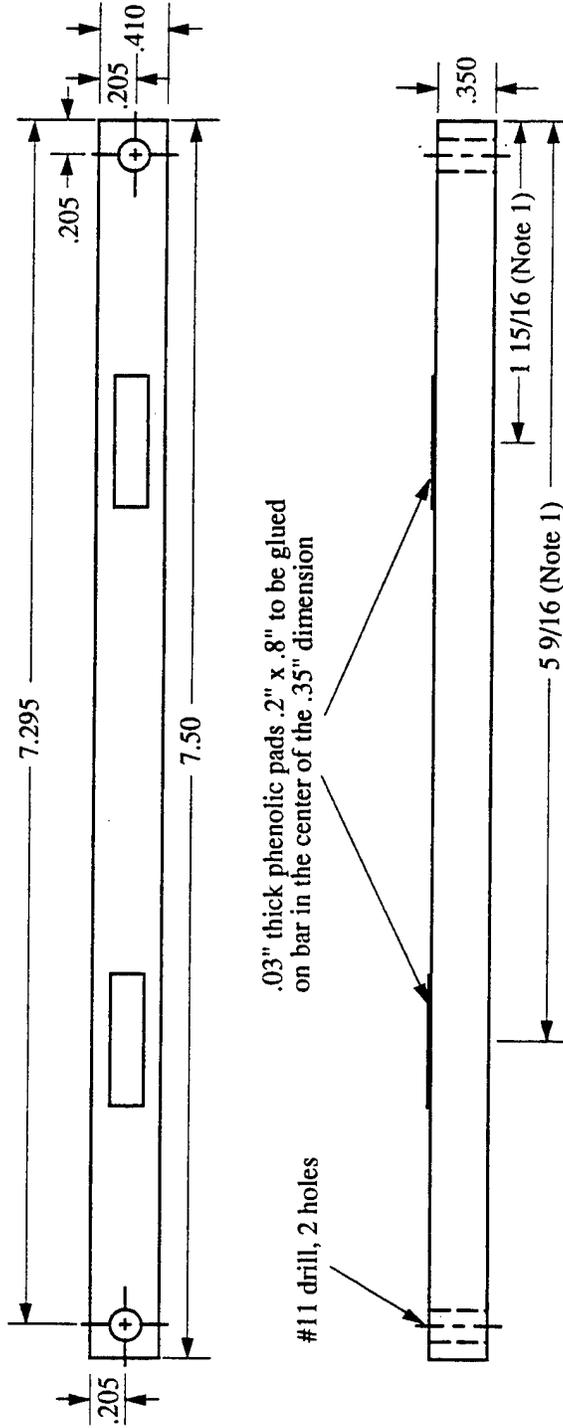
.03" thick phenolic pads .2" x .8" to be glued on bar in the center of the .35" dimension

#21 drill, tap for 10-32 threads, 2 holes

Notes: 1. These dimensions are to the centers of the phenolic pads.

1		beam	Al alloy	as mach.	N/A	none	QIDEC					
							Quality and Integrity Design Engineering Center					
Quan. Req.	Part No.	Description	Material/ Specification	Finish	Stock Size	Part Mark	DATE	DRAWN	CHECKED	APPROVED	DWG. TITLE	DWG. NO.
	- 1						Jan. 24, 1995	C. B. Elliott	A. Georgeson	C. B. Elliott	Riveted Joint Stiffener System - Constraint Beam away from Jt. - Unthread	FAAMSC95036
							TOLERANCES:		DECIMALS X.XXX ±0.005"		X.XX ±0.03"	
									unless otherwise stated		NEXT ASSY. FAAMSC95036	

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	.175 to .205 three times, .350 to .410, 7.325 to 7.295	CBE	CG	Feb. 7, 1995



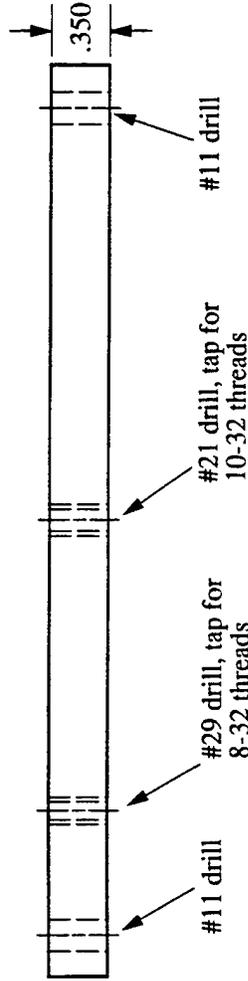
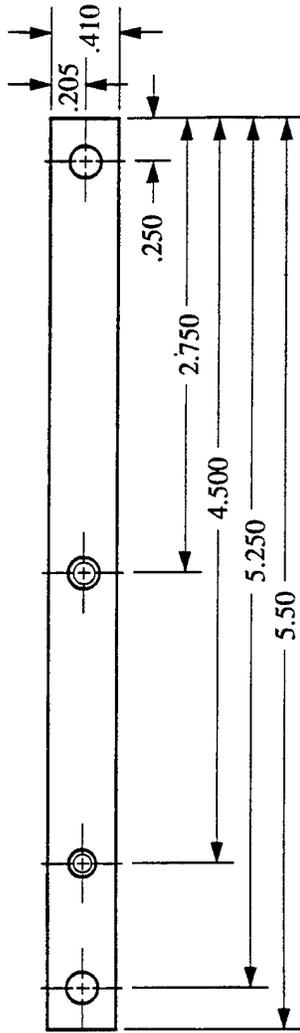
#11 drill, 2 holes

.03" thick phenolic pads .2" x .8" to be glued on bar in the center of the .35" dimension

Notes: 1. These dimensions are to the centers of the phenolic pads.

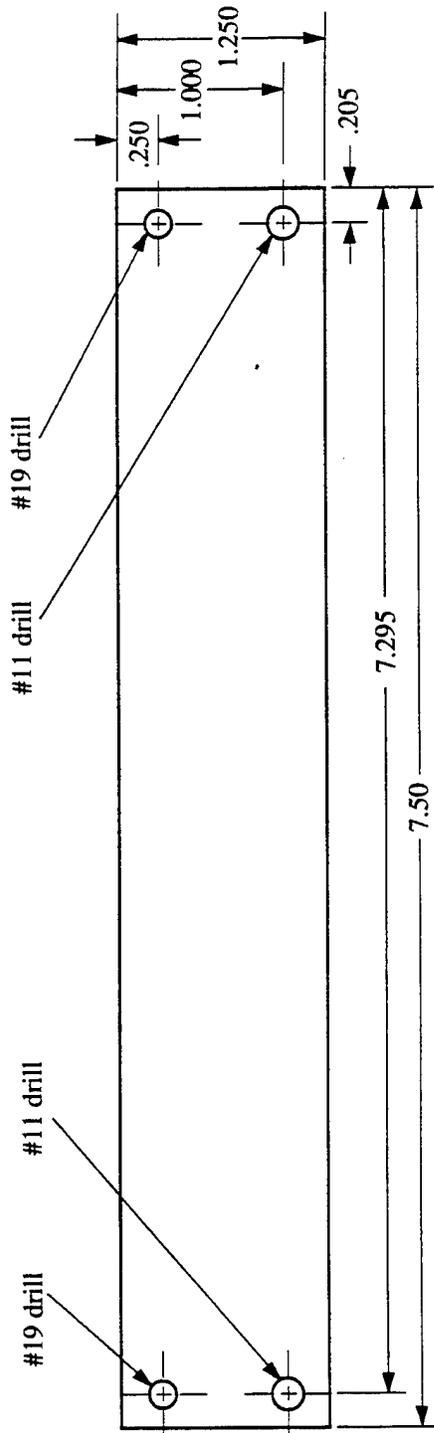
1		- 1	beam	Al alloy	as mach.	N/A	none	QIDEC	
								Quality and Integrity Design Engineering Center	
Quan. Req.	Part No.	Description	Material/ Specification	Finish	Stock Size	Part Mark	SCALE 1:1	DATE Jan. 24, 1995	DRAWN C. B. Elliott
							DWG. SIZE A	CHECKED A. Georgeson	TOLERANCES: DECIMALS X.XXX ±0.005" X.XX ±0.03" unless otherwise stated
							DWG. TITLE Riveted Joint Stiffner System - Constraint Beam away from Jt. - Unthread	APPROVED C. B. Elliott	SHEET NEXT ASSY. 1 of 1 FAAMSC95036
									DWG. NO. FAAMSC95039

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	.175 to .205, .350 to .410	CBE	AG	Feb. 7, 1995



4		- 1	beam	Al alloy	as mach.	N/A	none	Part Mark	Description	Material/ Specification	Finish	Stock Size	Part Mark	- 1	Part No.	- 1	Quant. Req.
<b>QIDEC</b> Quality and Integrity Design Engineering Center																	
DATE		Jan. 24, 1995		DRAWN		C. B. Elliott		TOLERANCES:		DECIMALS X.XXX ±0.005"		X.XX ±0.03"		unless otherwise stated		SHEET	
SCALE		1:1		CHECKED		A. Georgeson		DWG. SIZE		A		APPROVED		C. B. Elliott		1 of 1	
DWG. TITLE		Riveted Joint Stiffner		System - Side Beam		DWG. NO.		FAAMSC95036		NEXT ASSY.		FAAMSC95036		DWG. NO.		FAAMSC95040	

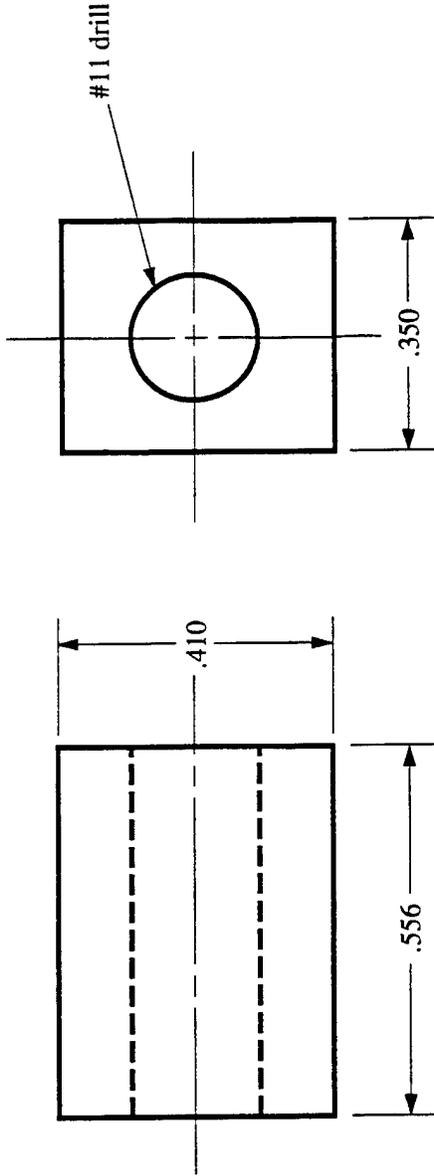
Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE	AG	
1	.175 to .205, 7.325 to 7.295	CBE	AG	Feb 67 1995



<b>QIDEC</b> Quality and Integrity Design Engineering Center		DRAWN	C. B. Elliott	TOLERANCES:	DECIMALS X.XXX ±0.005"
		CHECKED	A. Georgeson		X.XX ±0.03"
DATE	Jan. 24, 1995	APPROVED	C. B. Elliott	unless otherwise stated	
SCALE	1:1	DWG. TITLE	Riveted Joint Stiffener	SHEET	NEXT ASSY. FAAMSC95036
DWG. SIZE	A	System - Shape Plate		1 of 1	DWG. NO. FAAMSC95041

2	- 1	sheet metal	Al alloy	as rec.	.057"	none
Quan. Req.	Part No.	Description	Material/ Specification	Finish	Stock Size	Part Mark

Rev.	Description	Name	Check	Date
N/C	Initial Release	CBE		



QIDEC Quality and Integrity Design Engineering Center		DRAWN C. B., Elliott		TOLERANCES: DECIMALS X.XXX ±0.005" X.XX ±0.03" unless otherwise stated	
DATE	Feb. 6, 1995	CHECKED	<i>[Signature]</i>	SHEET	NEXT ASSY.
SCALE	4:1	APPROVED	<i>[Signature]</i>	1 of 1	FAAMSC95036
DWG. SIZE	A	DWG. TITLE Riveted Joint Stiffener System - Spacer		DWG. NO. FAAMSC95042	
4	- 1	spacer	Al alloy	as mach.	N/A
Quan. Req.	Part No.	Description	Material/ Specification	Finish	Stock Size
					Part Mark

## APPENDIX D—PLAN FOR FABRICATION, FATIGUE TESTING, AND EXAMINATION OF RIVETED JOINT SPECIMENS (AUGUST 4, 1995)

### 1. Overview

1.1. The Quality and Integrity Design Engineering Center (QIDEC) of the University of Utah is in the process of fabricating 12 riveted joint specimens to be used in fatigue testing. QIDEC is either contracting or performing all fabrication procedures as described in paragraph 2. QIDEC will perform all testing and examination of failed specimens in accordance with paragraph 3. This is a follow-on group of specimens. Testing of the initial specimens discussed in the March 20, 1995, plan resulted in significantly shorter lives for the CE specimens than for the FV specimens. It was considered that the rivet depth may have been too great for the CE specimens and that further testing was needed. This is the resulting experimentation program.

1.2. The 12 specimens will be fabricated from 0.063 gage 2024-T3 clad sheet aluminum alloy with riveting procedures as follow:

four C-squeeze joints with 7050 FV rivets,  
four C-squeeze joints with 2017 CE rivets, and  
four rivet gun with bucking bar joints with 2017 CE rivets.

1.3. In the paragraphs that follow, reference is continually made to the attached table D1 which shows the specimen preparation and testing protocol to be followed. Referenced drawings were provided with the March 20, 1995, plan

### 2. Joint Fabrication Procedures

2.1. QIDEC obtained 66 6.25-in-wide coupons of 2024-T3 clad sheet aluminum. CE and FV rivets remained from the initial test program.

2.1.1. The coupons were numbered consecutively as unpacked and the thickness of one end of each coupon was measured twice with a digital micrometer and recorded. This procedure was stopped when 28 coupons (24 for use and 4 spares) with the same thickness of 0.062 in were identified. Random number generation was used to match coupons and within each experimental block to determine rivet type, riveting procedure, the order of machining and the order of riveting. The results of this sorting procedure are shown in table D1.

2.1.2. Sufficient rivets of each type with identical measurements were available for all joints to be manufactured. These dimensions were 0.1875 in. diameter by 0.3475 in. length (including head) for the FV rivets and 0.188 in. diameter by 0.346 in. length (including head) for the CE rivets. These are the same dimensions as for the initial program.

2.2. Each coupon was machined to size in accordance with QIDEC drawings FAASPC94042 and FAASPC94043.

2.3. The 28 coupons were allodined in a single batch in accordance with MIL-C-5541 class 1A by the same company that did the allodining for the initial program.

2.4. The 28 coupons are being coated in a single batch with BMS 10-11 type 1 class A adhesive primer by the same company that did the priming for the initial program.

2.5. Each set of coupons will be clamped together in a machining fixture and not separated until all of the machining in the rivet area is completed.

2.6. All sets will be machined and marked in accordance with QIDEC drawings FAASPC94044, and FAASPC94045 using one milling machine. The order of machining will be as shown in table D1. Initially, pilot holes will be drilled with a centering drill to ensure alignment. Number 10 double margin drills ground to 0.192 in. diameter will be used to drill all rivet holes. Countersink stops will be used to drill all countersinks in order to ensure consistent countersink depth. These stops were set based on inspection of holes drilled and countersunk during the initial 36 specimen test program. The FV countersink will be set as it was in the initial program. The CE countersink will be set to reduce rivet depth. The settings will not be changed during the specimen machining program. The specific drill and countersink used to machine each hole will be identified and documented.

2.7. Rivet hole diameters will be measured using a Meyer C-10 gage set and recorded for all rivet holes in each coupon where the rivet heads are to be located.

2.8. Joints will be riveted in accordance with QIDEC drawings FAASPC94046, FAASPC94047, and FAASPC94049. The order of riveting will be as shown in table D1. All specimens within a block will be riveted during the same riveting session. Procedures for each block and between blocks will be as consistent as possible and documented for each rivet, specimen, and block. Specifically, the same person will perform the same tasks using the same equipment and procedures during riveting of all specimens to the extent feasible and appropriate. The diameter of each bucktail will be determined to be within specifications by use of go/no-go gages. Environmental (temperature and relative humidity) and lighting conditions will be as consistent as possible. Temperature and relative humidity will be recorded at the time that the riveting of each specimen is begun.

2.9. Rivet head clearances will be computed and recorded for all rivets in each specimen based upon measurements using a machinists microscope.

### 3. Joint Fatigue Testing And Examination Procedures

3.1. All testing will be conducted on the same calibrated MTS servo hydraulic tension testing load system. Prior to testing the load system will be aligned by direct alignment of the grips. Additionally, alignment will be verified prior to or following testing of each block of specimens (see table D1) using an alignment specimen produced by QIDEC.

3.2. Testing will be conducted sequentially within four test blocks. The order of testing, based on random number generation within each block, will be as shown in table D1. Fatigue loading will be with a 10 Hz sine wave driving signal, a maximum nominal tensile stress of 17 ksi, and a stress ratio of  $R = 0.1$ . The system shown in the drawings provided with the March 20, 1995, plan will be used during the fatigue testing to counter the tendency of the riveted joints to bend in the riveted area when under tensile loading. Temperature and relative humidity will be recorded at the time that the testing of each specimen is begun. Total cycles to fracture (test system interlock activation) will be determined for each specimen.

3.3. Following testing, failed joints will be stored in a desiccant chamber when they are not being examined.

3.4. The joints will be examined optically with little or no magnification and using an optical microscope to identify probable locations and causes of crack nucleation. Then, metallographic and fractographic investigations will be performed to confirm the causes of crack nucleation and identify the mechanisms involved in the process. Photographs will be taken as appropriate.

Table D1  
Riveted Joint Work Order Plan

Block No.	Joint No.	Coupon Numbers <sup>1</sup>	Rivet Type and Procedure <sup>2</sup>	Machining			Riveting			Test		
				Order <sup>2</sup>								
1	1	16	CE C-squeeze	3			3			1		
	2	19	CE Rivet gun	2			2			3		
	3	26	FV C-squeeze	1			1			2		
2	4	18	CE Rivet gun	6			4			5		
	5	10	CE C-squeeze	5			6			6		
	6	7	FV C-squeeze	4			5			4		
3	7	9	FV C-squeeze	7			8			9		
	8	21	CE Rivet gun	9			7			7		
	9	24	CE C-squeeze	8			9			8		
4	10	25	FV C-squeeze	11			12			11		
	11	3	CE Rivet gun	10			10			10		
	12	1	CE C-squeeze	12			11			12		

1. Chosen randomly from all coupons with 0.062 in thickness. Spares are 8 with 2 and 5 with 27.

2. Chosen randomly within each block.

APPENDIX E—DATA FOR FABRICATION, FATIGUE TESTING, AND  
EXAMINATION OF RIVETED JOINT SPECIMENS

Specimen 1    FV rivet, rivet gun with bucking bar    Riveting temp. 75 deg. F/ RH 40%  
216,450 cycles to failure    Test temp. 74 deg. F/ RH 48%

① Hole diam. .192" Head clear. .007" Remarks: Bucktail .284"	⑥ Hole diam. .192" Head clear. .002" Remarks:	⑪ Hole diam. .192" Head clear. .003" Remarks:
② Hole diam. .192" Head clear. .007" Remarks:	⑦ Hole diam. .192" Head clear. .004" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .001" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .001" Remarks:
⑤ Hole diam. .192" Head clear. .006" Remarks:	⑩ Hole diam. .192" Head clear. .003" Remarks:	⑮ Hole diam. .192" Head clear. .004" Remarks:

Remarks: No fatigue cracking observed at rivet holes 14 and 15.

Specimen 2    FV rivet, C-squeeze    Riveting temp. 75 deg. F/ RH 39%  
193,520 cycles to failure    Test temp. 74 deg. F/ RH 48%

① Hole diam. .192" Head clear. .004" Remarks:	⑥ Hole diam. .191" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .004" Remarks:	⑦ Hole diam. .192" Head clear. .004" Remarks:	⑫ Hole diam. .192" Head clear. .006" Remarks:
③ Hole diam. .192" Head clear. .003" Remarks:	⑧ Hole diam. .192" Head clear. .003" Remarks:	⑬ Hole diam. .191" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .003" Remarks:	⑨ Hole diam. .192" Head clear. .001" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .191" Head clear. .004" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .001" Remarks:

Remarks:

Specimen 3    CE rivet, C-squeeze    Riveting temp. 75 deg. F/ RH 39%  
40,270 cycles to failure    Test temp. 74 deg. F/ RH 47%

① Hole diam. .192" Head clear. .000" Remarks:	⑥ Hole diam. .192" Head clear. -.001" Remarks:	⑪ Hole diam. .192" Head clear. -.001" Remarks:
② Hole diam. .192" Head clear. -.001" Remarks:	⑦ Hole diam. .191" Head clear. -.001" Remarks:	⑫ Hole diam. .192" Head clear. .005" Remarks:
③ Hole diam. .192" Head clear. .000" Remarks:	⑧ Hole diam. .192" Head clear. .001" Remarks:	⑬ Hole diam. .191" Head clear. -.001" Remarks:
④ Hole diam. .192" Head clear. .000" Remarks:	⑨ Hole diam. .192" Head clear. .000" Remarks:	⑭ Hole diam. .191" Head clear. -.001" Remarks:
⑤ Hole diam. .192" Head clear. -.001" Remarks:	⑩ Hole diam. .191" Head clear. -.001" Remarks:	⑮ Hole diam. .192" Head clear. -.001" Remarks:

Remarks: Specimen did not fracture. No crack observed at rivet hole 11.

Specimen 4 CE rivet, rivet gun with bucking bar Riveting temp. 75 deg. F/ RH 40%  
 43,130 cycles to failure Test temp. 75 deg. F/ RH 47%

① Hole diam. .192" Head clear. .001" Remarks:	⑥ Hole diam. .192" Head clear. .000" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .001" Remarks:	⑦ Hole diam. .192" Head clear. .001" Remarks:	⑫ Hole diam. .192" Head clear. .000" Remarks: riveted
③ Hole diam. .192" Head clear. .001" Remarks:	⑧ Hole diam. .192" Head clear. .000" Remarks:	⑬ Hole diam. .191" Head clear. .001" Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .001" Remarks:	⑭ Hole diam. .192" Head clear. .000" Remarks:
⑤ Hole diam. .192" Head clear. -.002" Remarks:	⑩ Hole diam. .192" Head clear. -.002" or more Remarks:	⑮ Hole diam. .192" Head clear. -.002" or more Remarks:

Remarks:

Specimen 5 FV rivet, rivet gun with bucking bar Riveting temp. 75 deg. F/ RH 42%  
 178,020 cycles to failure Test temp. 74 deg. F/ RH 32%

① Hole diam. .192" Head clear. .003" Remarks:	⑥ Hole diam. .192" Head clear. .001" Remarks:	⑪ Hole diam. .192" Head clear. .003" Remarks:
② Hole diam. .192" Head clear. .002" Remarks:	⑦ Hole diam. .192" Head clear. .002" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .003" Remarks:	⑬ Hole diam. .192" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .003" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .192" Head clear. .004" Remarks:	⑩ Hole diam. .192" Head clear. .004" Remarks:	⑮ Hole diam. .192" Head clear. .002" Remarks:

Remarks:

Specimen 6 CE rivet, C-squeeze Riveting temp. 74 deg. F/ RH 43%  
 58,030 cycles to failure Test temp. 74 deg. F/ RH 38%

① Hole diam. .192" Head clear. .000" Remarks:	⑥ Hole diam. .192" Head clear. .000" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .006" Remarks: riveted	⑦ Hole diam. .191" Head clear. .001" Remarks:	⑫ Hole diam. .192" Head clear. .000" Remarks:
③ Hole diam. .192" Head clear. .000" Remarks:	⑧ Hole diam. .192" Head clear. .001" Remarks:	⑬ Hole diam. .192" Head clear. .000" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .192" Head clear. .000" Remarks:	⑩ Hole diam. .192" Head clear. .000" Remarks:	⑮ Hole diam. .192" Head clear. .000" Remarks:

Remarks:

Specimen 7 CE rivet, rivet gun with bucking bar Riveting temp. 75 deg. F/ RH 42%  
 36,880 cycles to failure Test temp. 74 deg. F/ RH 34%

① Hole diam. .192" Head clear. .000" Remarks:	⑥ Hole diam. .192" Head clear. .000" Remarks:	⑪ Hole diam. .192" Head clear. -.005" Remarks:
② Hole diam. .191" Head clear. -.001" Remarks:	⑦ Hole diam. .192" Head clear. .000" Remarks:	⑫ Hole diam. .191" Head clear. .000" Remarks: riveted
③ Hole diam. .192" Head clear. .001" Remarks:	⑧ Hole diam. .191" Head clear. -.001" Remarks:	⑬ Hole diam. .192" Head clear. -.002" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .001" Remarks:	⑭ Hole diam. .192" Head clear. -.001" Remarks:
⑤ Hole diam. .192" Head clear. -.001" Remarks:	⑩ Hole diam. .192" Head clear. -.004" Remarks:	⑮ Hole diam. .192" Head clear. -.002" Remarks:

Remarks:

Specimen 8 FV rivet, C-squeeze Riveting temp. 74 deg. F/ RH 43%  
 142,660 cycles to failure Test temp. 74 deg. F/ RH 35%

① Hole diam. .192" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .002" Remarks:	⑪ Hole diam. .192" Head clear. .003" Remarks:
② Hole diam. .191" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .005" Remarks:	⑫ Hole diam. .191" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .004" Remarks:	⑬ Hole diam. .192" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .191" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .001" Remarks:
⑤ Hole diam. .192" Head clear. .001" Remarks:	⑩ Hole diam. .192" Head clear. .001" Remarks:	⑮ Hole diam. .192" Head clear. .001" Remarks:

Remarks:

Specimen 9 CE rivet, rivet gun with bucking bar Riveting temp. 74 deg. F/ RH 40%  
 32,200 cycles to failure Test temp. 74 deg. F/ RH 43%

① Hole diam. .192" Head clear. .000" Remarks:	⑥ Hole diam. .192" Head clear. -.001" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .000" Remarks:	⑦ Hole diam. .192" Head clear. .000" Remarks:	⑫ Hole diam. .192" Head clear. -.001" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .000" Remarks:	⑬ Hole diam. .192" Head clear. -.003" Remarks:
④ Hole diam. .191" Head clear. -.001" Remarks:	⑨ Hole diam. .192" Head clear. .000" Remarks:	⑭ Hole diam. .192" Head clear. .000" Remarks:
⑤ Hole diam. .192" Head clear. -.002" Remarks:	⑩ Hole diam. .192" Head clear. -.002" Remarks:	⑮ Hole diam. .192" Head clear. -.001" Remarks:

Remarks:

Specimen 10 CE rivet, C-squeeze  
54,080 cycles to failure

Riveting temp. 75 deg. F/RH 39%  
Test temp. 74 deg. F/RH 46%

① Hole diam. .192" Head clear. .002" Remarks:	⑥ Hole diam. .192" Head clear. .002" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .001" Remarks:	⑦ Hole diam. .192" Head clear. .002" Remarks:	⑫ Hole diam. .192" Head clear. .001" Remarks:
③ Hole diam. .192" Head clear. .004" Remarks:	⑧ Hole diam. .192" Head clear. -.001" Remarks:	⑬ Hole diam. .191" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .004" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .191" Head clear. .002" Remarks:
⑤ Hole diam. .192" Head clear. .000" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .002" Remarks:

Remarks:

Specimen 11 FV rivet, C-squeeze  
146,860 cycles to failure

Riveting temp. 74 deg. F/RH 42%  
Test temp. 74 deg. F/RH 39%

① Hole diam. .192" Head clear. .005" Remarks:	⑥ Hole diam. .192" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .002" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .192" Head clear. .007" Remarks:
③ Hole diam. .192" Head clear. .003" Remarks:	⑧ Hole diam. .192" Head clear. .001" Remarks:	⑬ Hole diam. .192" Head clear. .004" Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .001" Remarks:
⑤ Hole diam. .192" Head clear. .003" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .001" Remarks:

Remarks:

Specimen 12 FV rivet, rivet gun with bucking bar  
363,780 cycles to failure

Riveting temp. 74 deg. F/RH 42%  
Test temp. 74 deg. F/RH 45%

① Hole diam. .191" Head clear. .005" Remarks:	⑥ Hole diam. .191" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .004" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .191" Head clear. .004" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .004" Remarks:	⑧ Hole diam. .192" Head clear. .004" Remarks:	⑬ Hole diam. .192" Head clear. .004" Remarks: riveted
④ Hole diam. .192" Head clear. .003" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .003" Remarks:
⑤ Hole diam. .192" Head clear. .003" Remarks:	⑩ Hole diam. .192" Head clear. .003" Remarks:	⑮ Hole diam. .191" Head clear. .002" Remarks:

Remarks: Not machined with specimens 9-11 as planned.

Specimen 13 FV rivet, C-squeeze  
167,720 cycles to failure

Riveting temp. 77 deg. F/RH 40%  
Test temp. 74 deg. F/RH 43%

① Hole diam. .192" Head clear. .003" Remarks:	⑥ Hole diam. .192" Head clear. .001" Remarks:	⑪ Hole diam. .192" Head clear. .004" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .002" Remarks:	⑫ Hole diam. .192" Head clear. .004" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .004" Remarks:	⑬ Hole diam. .192" Head clear. .004" Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .191" Head clear. .001" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .191" Head clear. .003" Remarks:

Remarks: Specimen did not fracture. No crack observed at rivet hole 15.

Specimen 14 CE rivet, C-squeeze  
37,380 cycles to failure

Riveting temp. 77 deg. F/RH 41%  
Test temp. 74 deg. F/RH 42%

① Hole diam. .192" Head clear. .001" Remarks: riveted	⑥ Hole diam. .192" Head clear. .004" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks: riveted
② Hole diam. .192" Head clear. -.001" Remarks: riveted	⑦ Hole diam. .192" Head clear. .000" Remarks: riveted	⑫ Hole diam. .192" Head clear. -.002" Remarks: riveted
③ Hole diam. .192" Head clear. .001" Remarks: riveted	⑧ Hole diam. .192" Head clear. .000" Remarks: riveted	⑬ Hole diam. .192" Head clear. .001" Remarks: riveted
④ Hole diam. .191" Head clear. .000" Remarks: riveted	⑨ Hole diam. .192" Head clear. .000" Remarks: riveted	⑭ Hole diam. .192" Head clear. .000" Remarks:
⑤ Hole diam. .192" Head clear. .000" Remarks:	⑩ Hole diam. .192" Head clear. -.002" Remarks: riveted	⑮ Hole diam. .192" Head clear. .000" Remarks: riveted

Remarks: Data are suspect because the riveting was done the day after the joint was initially riveted.

Specimen 15 CE rivet, rivet gun with bucking bar  
41,400 cycles to failure

Riveting temp. 78 deg. F/RH 42%  
Test temp. 74 deg. F/RH 50%

① Hole diam. .192" Head clear. .000" Remarks:	⑥ Hole diam. .192" Head clear. .000" Remarks:	⑪ Hole diam. .192" Head clear. .000" Remarks:
② Hole diam. .191" Head clear. .000" Remarks:	⑦ Hole diam. .192" Head clear. -.001" Remarks:	⑫ Hole diam. .192" Head clear. .000" Remarks:
③ Hole diam. .192" Head clear. .000" Remarks:	⑧ Hole diam. .192" Head clear. .000" Remarks:	⑬ Hole diam. .192" Head clear. .000" Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .001" Remarks:	⑭ Hole diam. .192" Head clear. .001" Remarks:
⑤ Hole diam. .192" Head clear. .000" Remarks:	⑩ Hole diam. .192" Head clear. .000" Remarks:	⑮ Hole diam. .192" Head clear. -.001" Remarks:

Remarks:

Specimen 16 FV rivet, rivet gun with bucking bar Riveting temp. 78 deg. F/ RH 39%  
 248,860 cycles to failure Test temp. 74 deg. F/ RH 49%

① Hole diam. .192" Head clear. .003" Remarks:	⑥ Hole diam. .192" Head clear. .002" Remarks:	⑪ Hole diam. .192" Head clear. .004" Remarks:
② Hole diam. .192" Head clear. .001" Remarks:	⑦ Hole diam. .192" Head clear. .001" Remarks:	⑫ Hole diam. .192" Head clear. .001" Remarks:
③ Hole diam. .192" Head clear. .001" Remarks:	⑧ Hole diam. .192" Head clear. .001" Remarks:	⑬ Hole diam. .192" Head clear. .001" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .001" Remarks:	⑭ Hole diam. .192" Head clear. .001" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks:

Specimen 17 FV rivet, rivet gun with bucking bar Riveting temp. 78 deg. F/ RH 37%  
 248,090 cycles to failure Test temp. 74 deg. F/ RH 46%

① Hole diam. .192" Head clear. .002" Remarks:	⑥ Hole diam. .192" Head clear. .003" Remarks:	⑪ Hole diam. .191" Head clear. .002" Remarks:
② Hole diam. .192" Head clear. .002" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .001" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .001" Remarks:

Remarks:

Specimen 18 CE rivet, rivet gun with bucking bar Riveting temp. 78 deg. F/ RH 39%  
 33,580 cycles to failure Test temp. 74 deg. F/ RH 42%

① Hole diam. .191" Head clear. .000" Remarks:	⑥ Hole diam. .192" Head clear. .001" Remarks:	⑪ Hole diam. .192" Head clear. unknown Remarks:
② Hole diam. .192" Head clear. .000" Remarks:	⑦ Hole diam. .192" Head clear. .000" Remarks:	⑫ Hole diam. .192" Head clear. unknown Remarks:
③ Hole diam. .192" Head clear. .000" Remarks:	⑧ Hole diam. .192" Head clear. .000" Remarks:	⑬ Hole diam. .192" Head clear. unknown Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .000" Remarks:	⑭ Hole diam. .192" Head clear. unknown Remarks:
⑤ Hole diam. .192" Head clear. -.001" Remarks:	⑩ Hole diam. .192" Head clear. -.001" Remarks:	⑮ Hole diam. .192" Head clear. unknown Remarks:

Remarks:

Specimen 1R CE rivet, C-squeeze  
168,980 cycles to failure

Riveting temp. 74 deg. F/RH 52%  
Test temp. 75 deg. F/RH 44%

① Hole diam. .191" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .004" Remarks:	⑪ Hole diam. .191" Head clear. .002" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .191" Head clear. .004" Remarks:
③ Hole diam. .191" Head clear. .003" Remarks:	⑧ Hole diam. .191" Head clear. .003" Remarks:	⑬ Hole diam. .192" Head clear. .004" Remarks:
④ Hole diam. .192" Head clear. .005" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .191" Head clear. .003" Remarks:
⑤ Hole diam. .191" Head clear. .004" Remarks:	⑩ Hole diam. .192" Head clear. .003" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks:

Specimen 2R CE rivet, rivet gun with bucking bar  
87,680 cycles to failure

Riveting temp. 74 deg. F/RH 52%  
Test temp. 75 deg. F/RH 55%

① Hole diam. .191" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .004" Remarks:	⑪ Hole diam. .191" Head clear. .004" Remarks:
② Hole diam. .191" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .002" Remarks:	⑫ Hole diam. .191" Head clear. .002" Remarks:
③ Hole diam. .191" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .003" Remarks:	⑬ Hole diam. .192" Head clear. .004" Remarks:
④ Hole diam. .191" Head clear. .005" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .004" Remarks:
⑤ Hole diam. .192" Head clear. .003" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks:

Specimen 3R FV rivet, C-squeeze  
120,050 cycles to failure

Riveting temp. 74 deg. F/RH 51%  
Test temp. 75 deg. F/RH 41%

① Hole diam. .192" Head clear. .003" Remarks:	⑥ Hole diam. .192" Head clear. .005" Remarks:	⑪ Hole diam. .192" Head clear. .002" Remarks:
② Hole diam. .192" Head clear. .002" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .192" Head clear. .004" Remarks:
③ Hole diam. .192" Head clear. .003" Remarks:	⑧ Hole diam. .192" Head clear. .003" Remarks:	⑬ Hole diam. .192" Head clear. .002" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .004" Remarks:
⑤ Hole diam. .192" Head clear. .001" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .000" Remarks:

Remarks: Grips failed at 38,090 cycles. During analysis the test was found to be invalid.

Specimen 4R CE rivet, rivet gun with bucking bar  
95,690 cycles to failure

Riveting temp. 74 deg. F/RH 57%  
Test temp. 75 deg. F/RH 51%

① Hole diam. .191" Head clear. .005" Remarks:	⑥ Hole diam. .192" Head clear. .004" Remarks:	⑪ Hole diam. .192" Head clear. .004" Remarks:
② Hole diam. .191" Head clear. .004" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .191" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .003" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .005" Remarks:
④ Hole diam. .192" Head clear. .003" Remarks:	⑨ Hole diam. .192" Head clear. .004" Remarks:	⑭ Hole diam. .192" Head clear. .003" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .003" Remarks:	⑮ Hole diam. .192" Head clear. .002" Remarks:

Remarks:

Specimen 5R CE rivet, C-squeeze  
139,250 cycles to failure

Riveting temp. 74 deg. F/RH 57%  
Test temp. 74 deg. F/RH 53%

① Hole diam. .191" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .004" Remarks:	⑪ Hole diam. .192" Head clear. .006" Remarks:
② Hole diam. .191" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .004" Remarks:	⑫ Hole diam. .192" Head clear. .004" Remarks:
③ Hole diam. .192" Head clear. .006" Remarks:	⑧ Hole diam. .192" Head clear. .003" Remarks:	⑬ Hole diam. .192" Head clear. .003" Remarks:
④ Hole diam. .191" Head clear. .004" Remarks:	⑨ Hole diam. .192" Head clear. .004" Remarks:	⑭ Hole diam. .192" Head clear. .004" Remarks:
⑤ Hole diam. .191" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks:

Specimen 6R FV rivet, C-squeeze  
157,340 cycles to failure

Riveting temp. 74 deg. F/RH 57%  
Test temp. 75 deg. F/RH 56%

① Hole diam. .192" Head clear. .002" Remarks:	⑥ Hole diam. .192" Head clear. .001" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .001" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .000" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .191" Head clear. .003" Remarks:
④ Hole diam. .191" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .001" Remarks:	⑭ Hole diam. .192" Head clear. .001" Remarks:
⑤ Hole diam. .191" Head clear. .001" Remarks:	⑩ Hole diam. .192" Head clear. .001" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks:

Specimen 7R FV rivet, C-squeeze  
174,460 cycles to failure

Riveting temp. 74 deg. F/ RH 58%  
Test temp. 74 deg. F/ RH 43%

① Hole diam. .192" Head clear. .001" Remarks:	⑥ Hole diam. .192" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .002" Remarks:
② Hole diam. .192" Head clear. .002" Remarks:	⑦ Hole diam. .192" Head clear. .002" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .004" Remarks:
④ Hole diam. .192" Head clear. .002" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .003" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .001" Remarks:	⑮ Hole diam. .192" Head clear. .001" Remarks:

Remarks: No fatigue cracking observed at rivet holes 12 and 15.

Specimen 8R CE rivet, rivet gun with bucking bar  
107,710 cycles to failure

Riveting temp. 75 deg. F/ RH 58%  
Test temp. 75 deg. F/ RH 50%

① Hole diam. .192" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .004" Remarks:
② Hole diam. .192" Head clear. .004" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .004" Remarks:	⑧ Hole diam. .192" Head clear. .003" Remarks:	⑬ Hole diam. .192" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .004" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .003" Remarks:
⑤ Hole diam. .192" Head clear. .003" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks:

Specimen 9R CE rivet, C-squeeze  
127,980 cycles to failure

Riveting temp. 75 deg. F/ RH 58%  
Test temp. 75 deg. F/ RH 50%

① Hole diam. .192" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .007" Remarks:
② Hole diam. .192" Head clear. .007" Remarks:	⑦ Hole diam. .192" Head clear. .014" Remarks:	⑫ Hole diam. .192" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .002" Remarks:	⑧ Hole diam. .192" Head clear. .005" Remarks:	⑬ Hole diam. .192" Head clear. .005" Remarks:
④ Hole diam. .191" Head clear. .005" Remarks:	⑨ Hole diam. .192" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .004" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .002" Remarks:

Remarks:

Specimen 10R FV rivet, C-squeeze  
156,640 cycles to failure

Riveting temp. 75 deg. F/RH 60%  
Test temp. 75 deg. F/RH 37%

① Hole diam. .192" Head clear. .001" Remarks:	⑥ Hole diam. .192" Head clear. .000" Remarks:	⑪ Hole diam. .192" Head clear. .001" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .005" Remarks:	⑫ Hole diam. .192" Head clear. .002" Remarks:
③ Hole diam. .192" Head clear. .001" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .001" Remarks:
④ Hole diam. .192" Head clear. .001" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .000" Remarks:	⑮ Hole diam. .192" Head clear. .002" Remarks:

Remarks: Specimen did not fracture on side of rivet hole 15.

Specimen 11R CE rivet, rivet gun with bucking bar  
101,890 cycles to failure

Riveting temp. 75 deg. F/RH 60%  
Test temp. 75 deg. F/RH 43%

① Hole diam. .192" Head clear. .004" Remarks:	⑥ Hole diam. .192" Head clear. .003" Remarks:	⑪ Hole diam. .192" Head clear. .002" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .002" Remarks:	⑫ Hole diam. .191" Head clear. .003" Remarks:
③ Hole diam. .192" Head clear. .003" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .003" Remarks:
④ Hole diam. .192" Head clear. .003" Remarks:	⑨ Hole diam. .191" Head clear. .003" Remarks:	⑭ Hole diam. .192" Head clear. .003" Remarks:
⑤ Hole diam. .192" Head clear. .002" Remarks:	⑩ Hole diam. .192" Head clear. .002" Remarks:	⑮ Hole diam. .192" Head clear. .002" Remarks:

Remarks:

Specimen 12R CE rivet, C-squeeze  
112,890 cycles to failure

Riveting temp. 75 deg. F/RH 59%  
Test temp. 74 deg. F/RH 38%

① Hole diam. .192" Head clear. .002" Remarks:	⑥ Hole diam. .192" Head clear. .002" Remarks:	⑪ Hole diam. .192" Head clear. .005" Remarks:
② Hole diam. .192" Head clear. .003" Remarks:	⑦ Hole diam. .192" Head clear. .003" Remarks:	⑫ Hole diam. .192" Head clear. .002" Remarks:
③ Hole diam. .192" Head clear. .004" Remarks:	⑧ Hole diam. .192" Head clear. .002" Remarks:	⑬ Hole diam. .192" Head clear. .005" Remarks:
④ Hole diam. .191" Head clear. .003" Remarks:	⑨ Hole diam. .192" Head clear. .002" Remarks:	⑭ Hole diam. .192" Head clear. .002" Remarks:
⑤ Hole diam. .192" Head clear. .003" Remarks:	⑩ Hole diam. .192" Head clear. .003" Remarks:	⑮ Hole diam. .192" Head clear. .003" Remarks:

Remarks: